MAINTENANCE PROGRAM

AIRBUS HELICOPTERS

EC 130 T2

DOCUMENT NUMBER	REVISION NUMBER	DATE
SCA/TEK/1-003	00	01 March 2020
PT. Smart Cakrawala Aviation		

PREPARED BY:

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TECHNICAL SUPPORT

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TECHNICAL MANAGER

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PT.SCA

MAINTENANCE

CMP

PROGRAM

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PROGRAM

ORIGINAL

MINISTRY OF TRANSPORTATION
DIRECTORATE GENERAL OF CIVIL AVIATION

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Our Ref : AU. 407/11/16/DKPPU-2020

Tangerang, 23 June 2020

To : Andreas Heryansyah

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Subject : **REVIEW FOR THE APPROVAL OF MAINTENANCE PROGRAM (MP) AIRBUS HELICOPTERS EC 130 T2 REV. 00 DATE 29 MAY 2020**

Dear Mr. Andreas Heryansyah,

I refer to the submission of the above mentioned document for review and approval on June 2020.

The document Airbus Helicopters EC 130 T2 Maintenance Program Rev. 00 date 29 May 2020 PT. Smart Cakrawala Aviation has been reviewed and found in compliance with the Civil Aviation Safety Regulation Part 135 and is **Approved**.

Sincerely Yours,

KEMENTERIAN PERHUBUNGAN DIREKTORAT JENDERAL PERHUBUNGAN UDARA

SOKHIB AL ROKHMAN
On behalf of the Director of Airworthiness and Aircraft Operation
Deputy Director for Airworthiness

cc. : Director of DAAO

MINISTRY OF TRANSPORTATION
DIRECTORATE GENERAL OF CIVIL AVIATION

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CONTROL PAGE

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LIST EFFECTIVE PAGES	LEP.1	00	29 May 2020
	LEP.2	00	29 May 2020
	LEP3	00	29 May 2020

This Maintenance Program (MP) Airbus Helicopters EC 130 T2 has been reviewed and found to meet in accordance with Airworthiness Limitations Section (ALS) Revision 009 date 09 September 2019, Master Servicing Manual (MSM) Revision 013 dated 09 September 2019 and Engine Turbomeca Arriel 2D Maintenance Manual Revision 19 date 30 December 2019, all applicable requirements set forth in the Civil Aviation Safety Regulations (CASR). This Maintenance Program Airbus Helicopters EC 130 T2 approved for used by PT. Smart Cakrawala Aviation with understanding that Directorate General of Civil Aviation (DGCA) may require further revisions to this Manual as regulatory requirement or airworthiness standards are amended.

Any change to these manuals shall be reported to the Director General of Civil Aviation (DGCA) for Approval.

Tangerang, 23 June 2020

On behalf of the Director of Airworthiness and Aircraft Operations

SOKHIB AL ROKHMAN
Deputy Director of Airworthiness

DISTRIBUTION LIST OF MAINTENANCE PROGRAM AIRBUS HELICOPTERS EC 130 T2

This Airbus Helicopters EC 130 T2 Maintenance Program shall be distributed to all personnel involved and will be responsible of Inspection Unit as the copy's controller.

DISTRIBUTION	COPIES NUMBER
Library	ORIGINAL
Chief Inspector	01
Technical Manager	02
DGCA	03
PK-SNX	10
Base Singkawang	11
Base Nabire	12

1. Printed Format Distribution List

   Any Printed-Format (Paper Copy) of this Maintenance Program is UNCONTROLLED except for Document as listed on Distribution List Table above.

2. Electronic Format Distribution List

   PT. Smart Cakrawala Aviation utilizes an electronic system for the management and control of this Maintenance Program. This document will be available and distributed throughout the organization in Portable Document Format (PDF).

RECORD OF REVISION

This record of revisions shall be retained in this Maintenance Program AIRBUS HELICOPTERS EC 130 T2. Revisions shall be inserted to replace the superseded pages in this document with the revision date, insertion date and name of person incorporating the revision annotated in the appropriate block below.

REVISION NUMBER	REVISION DATE	INSERTION DATE	INSERTED BY
00	01 March 2020	01 March 2020	ILHAM

REVISION HIGHLIGHT

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00	01 March 2020	All	All	Original

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PT. SMART CAKRAWALA AVIATION	DGCA
Andreas Heryansyah Technical Manager	Dafeo A.S DAAO Inspector

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1.1. PREFACE

PT. Smart Cakrawala Aviation is engaged in the carriage of air charter under the terms and conditions of CASR Part 135. The Maintenance Program for EC 130 T2 has been prepared in accordance with Maintenance Manual and Civil Aviation Safety Regulations, Parts 135.367 and is not contrary to any applicable Regulation, or the Company's Operations Specifications.

The Maintenance Program has been compiled for the use and guidance of all personnel responsible for performing maintenance and/or overhaul on aircraft, engine and appliances. Each manual is controlled and assigned to specific individual's aircraft type as necessary.

This manual is to be used in conjunction with other manuals, manufactures' maintenance and overhaul manuals and in accordance with applicable Aviation Regulations. The Maintenance Program will provide direction for use with aircraft, engine, and component maintenance and overhaul manuals. It also provides guidelines on how to fulfill requirements outlined in CASR's, AD's, SB's, etc. and the proper completion of the forms related to and distribution of the necessary reports in conjunction with the CASR's. If any material described in this Manual is in conflict with the CASR, the CASR will take priority. Manufacturer's manuals are also included and are considered part of the Company manual.

All maintenance employees are required to adhere to the instructions contained in this manual and follow the procedures outlined. In the event procedures in the manufacturers' publications differ from this manual, the manufacturer's manual prevails.

All personnel are encouraged to submit suggestions and recommendations to improve utilization and Maintenance quality of the manual.

1.2. INTRODUCTION

The purpose of this Maintenance Program is to provide guidance regarding the schedule, un-schedule maintenance, proper procedures and practices to be followed in conducting maintenance of EC 130 T2 under the Air Operator Certificate issued by Directorate General of Civil Aviation in compliance with Civil Aviation Safety Regulation (CASR) Part 135 and related Parts.

It will be the responsibility of Technical Manager and Chief Inspector to assure that all engineers, supervisors and inspectors are formally trained, kept current and familiar with the contents of this manual.

This Maintenance Program will be subjected to revision as necessary. It will be the responsibility of the person to whom this manual is issued to maintain it and insert all amendments and revisions. Such amendments and revisions will be issued by whom this manual is issued in the form of new revised pages.

1.3. USE OF MAINTENANCE PROGRAM MANUAL

This Maintenance Program Manual gives the necessary information required to help maintenance personnel to know the maintenance program, required inspection item and hard time components. It's designed to satisfy safety requirements, to avoid deterioration and to optimize aircraft availability with the reasonable costs in labor, material and facilities.

1.4. MANUAL CONTROL SYSTEM

1.4.1. Policy

Manuals are distributed on a required basis to the DGCA Office, PT. Smart Cakrawala Aviation Office and/or Maintenance Contract Agencies and should be used accordingly.

1. The manuals are not transferable. Holder shall retain the manuals originally issued regardless of change of station or location.
2. Text within a section (subject matter) should not be taken out of context. The reader should read the entire section for a complete understanding of the policies and procedures regarding a specific subject. If question arise, contact the Chief Inspector for clarification. A written response shall be made to clarify the matter in question.

1.4.2. Page Control System

1. Record of Revision
   Designed to quickly identify the current revision status of the manual.
2. List of Effective Pages
   Designed to provide a summary listing of all applicable pages and the revision date for the entire manual.

1.5. MANUAL REVISION AND PROCEDURE

1. Manual Revision

All amendments, revisions, and / or alterations to the Maintenance Program must be approved by DGCA. The changes shall be recorded through incorporation in a Record of Revision.

This Maintenance Program is amended as necessary to keep the information contained therein up to date in accordance with CASR's, Manufacture's Manual, AD's and SB's.

Changes shall not be introduced through written notation on the documents but through the removal of expired pages and insertion of revised pages. The bottom of each page shall indicate the issue and amendment status (dates and numbers).

2. Procedures

a. Each Maintenance Program will have control number and assignment entry on the manual cover page. Master list containing the manual number, location and revision status will be kept.
b. Chief Inspector will periodically review the Maintenance Program with all relevant manufacture's manual, this review will either confirm that manual still current and valid for the Air Operator Certificate use or will be identified needed change.
c. This manual and revision will be approved by Chief Inspector, and forward to DGCA for Approval. Upon acceptance and approved by DGCA, sufficient copies will be made and distributed the revision page to each manual holder.
d. A list effective pages will be issued with each revision so each manual can be checked and kept current.
e. Revision shall be numbered consecutively. The revised pages shall replace the earlier issued pages with the same part page number. Revision index shall be up dated; the revision status in the bottom left hand corner must have been adjusted. The removed pages must be deleted / destroyed. The changes on the revised pages with respect to the preceding ones shall be indicated with one vertical line.

f. In case a new issue to be published, the respective issue number shall be one higher than the preceding manual. The revision number of all pages shall become zero. The preceding issued manual shall now be deleted / destroyed.

g. Whenever revisions are made, either by the company or the manufacturer, Chief Inspector shall route them to the holders of the manuals. The responsibility for inserting revisions is the direct responsibility of the manual holder.

h. The portion of text which has been revised by the addition of, or a change in, information is shown by yellow highlighting of the text. Each revised page will only show revision highlight for text changed by revision. There will be no highlight if text was deleted from the page.

1.6. GENERAL INFORMATION

1. Company Address

PT. Smart Cakrawala Aviation is authorized by the Indonesia Directorate General of Civil Aviation (DGCA) under the Civil Aviation Safety Regulations (CASRs) as a Commercial Aircraft Operator.

The company office is in Jakarta, mailing address is as follows:

PT. SMART CAKRAWALA AVIATION
Head Office
Gedung Smartdeal lt.4
Jalan Cideng Timur No.16A
Jakarta Pusat 11310
Indonesia
Phone Number : +62 216305210
Fax : +62 216324873
Email : info@smartaviation.co.id

2. Aircraft General Specification:

a. Aircraft

Model	: EC 130 T2
Manufacturer	: Airbus Helicopter

b. Engine

Model	: Arriel 2D
Manufacturer	: Turbomeca

1.7. AFFECTED AIRCRAFT

This Maintenance Program is applicable to the following aircraft:

NO.	Make and Model	Serial Number	Reg. Mark
1	Airbus Helicopter EC 130 T2		PK-SNX

1.8. REFERENCES-

This Maintenance Program is the basic document, which provides and specifies all scheduled and unscheduled inspection program for the EC 130 T2 helicopter(s) and the related components in order to meet the minimum standard of airworthiness, which is required by the Civil Aviation Safety Regulations (CASRs) and manufacturers.

The Maintenance Program is prepared in accordance with the following Manufacturer's Technical Publications and Documents for maintenance:

a. Airbus Helicopter EC130T2 Airworthiness Limitations Section (ALS) Revision 009 dated September 09, 2019 and Master Servicing Manual (MSM) Revision 013 dated September 09, 2019.
b. Turbomeca Arriel 2D Maintenance Manual update No. 19 dated Dec. 30/2019.
c. Instruction for Continuous Airworthiness (ICAs) from STCs may applied to the aircrafts.
d. AD, SB, SL and other Information Concerning Airworthiness.
e. Indonesian Civil Aviation Safety Regulation.
f. Other documents concerning this Aircraft Maintenance Program.

1.9. GRACE PERIOD FOR NEW/REVISED TASK

For Task introduced in CAMP as a result of Airworthiness Limitation and Inspection Requirement, SB or SIL revision without any specific grace periods, the initial accomplishment of the task may be deferred to the nearest down time or aircraft inspection.

1.10. PERMITTED MARGINS TO MAINTENANCE INTERVAL

To introduce “flexibility” into maintenance planning in order to compensate for unpredictable situations (e.g. unforeseeable increase in the helicopter utilization rate), a value called “margin” is defined to be added to a limit value. This margin, added to the limit value to which it is applicable, results in the maximum limit value (limit value + margin value) to be taken into account for two consecutive inspections.

The margin can be used repetitively for each limit value interval. However, to maintain certain “flexibility” in maintenance planning, it is recommended to plan the maintenance operation using the limit value without taking the margin into account.

2.1. SCOPE

This chapter gives the mandatory replacement times and inspection intervals for components and structures that are life-limited. The section also gives the scheduled inspection requirements for structural and fatigue components that are considered a part of the certification process.

The Airworthiness Limitations is DAAO approved and specifies maintenance required by Parts 43.16 and 91.409 of the Civil Aviation and Safety Regulation. The following Airworthiness Limitations related to life or inspection of the airplane and its components have been established with respect to the Airworthiness Limitation Section (ALS) and Master Servicing Manual (MSM) of latest revision of Airbus Helicopters EC 130 T2 technical publications, Arriel 2D Maintenance Manual. These data are based on helicopters utilization, operation and maintenance in a category of service for which the airplane was originally designed.

2.2. LIFE LIMITED COPONENTS – AIRFRAME

2.2.1. SERVICE LIFE LIMITS (SLL)

Component which are essential for operating safety which are subject to undetectable damage due to the loads they withstand, are covered by a Service Life Limit (SLL). These components, as listed in table below must be removed from service when the specified limit is reached.

NO	DESCRIPTION	PART NUMBER	LIMITATION (SSL)	REMARK
ATA 62 – ROTOR(S)				
62-11 MAIN ROTOR BLADES				
1	Main Rotor Blade	355A11-0030-04	20000 FH	Margin: 0
62-21 MAIN ROTOR HEAD				
2	Starflex Star	350A31-1918-00	3000 FH	Margin: 0
3	Spherical Thrust Bearing	57910700 (704A33633211) LB4-1231-1 (704A33633208)	4600 FH	Margin: 0
4	Lower Sleeve Flange	350A31-1850-02	5750 FH	Margin: 0
5	Upper Sleeve Flange	350A31-1850-03	5750 FH	Margin: 0
6	Main Rotor Shaft	350A37-1290-04 350A37-1290-05	58000 TC	Margin: 0
7	Blade Horn	350A31-1877-03	20000 FH	Margin: 0
8	Spherical Thrust Bearing Bolt	350A31-1854-21	3000 FH	Margin: 0
9	MRH attachment bolt	350A37-1244-20 350A37-1245-20	3000 FH	Margin: 0
10	Unequipped blade pin	350A31-1771-21	5000 FH	Pin assembly MP/N 350A31- 1770-01. This MP/N must NOT mixed with MP/N 350A31-3020-20. Margin: 0
11	Unequipped blade pin	350A31-3020-20	10000 FH	Pin assemblies MP/N 350A31-1770-02, 350A31- 1772-03

NO	DESCRIPTION	PART NUMBER	LIMITATION (SSL)	REMARK
				and 350A31-1773-02. This MP/N must NOT be mixed with MP/N 350A31- 1771-21. Margin: 0
62-32 SWASH PLATE ASSEMBLY				
12	Scissors Drive Adapter	355A37-1235-27 355A37-1235-29	20000 FH	Margin: 0
13	Scissors Link Upper	350A37-1126-02 350A27-1126-03	20000 FH	When scrapping the scissors link upper, discard the bolts MP/N 350A37-1146-21 Margin: 0
14	Scissors Link Lower	350A37-1127-02	20000 FH	When scrapping the scissors link lower, discard the bolts MP/N 350A37-1146-21 Margin: 0
15	Non-rotating swashplate	350A37-1002-07	20000 FH	Margin: 0
16	Rotating Swashplate	350A37-1003-08 350A37-1018-00	20000 FH	Margin: 0
17	Mast Lower Housing	350A37-1292-20	20000 FH	Mast lower housing assembly MP/N 350A37-1292-00 Margin: 0
18	Mast Upper Housing	350A37-1291-20	20000 FH	Mast upper housing assembly MP/N 350A37- 1291-00 Margin: 0
ATA 63 – ROTOR DRIVE(S)				
63-11 MGB / ENGINE COUPLING				
19	MGB-Engine Link Shaft	350A35-1109-20	201000 TC	This part must be installed on helicopters carrying out external load carrying operations. Margin: 0
20	MGB-Engine Link Flange	350A35-1088-20	121000 TC	Fitted on assembly MP/N 350A35-1081-03. This part must be installed on helicopters carrying out external load carrying operations. Margin: 0
63-21 MGB				

NO	DESCRIPTION	PART NUMBER	LIMITATION (SSL)	REMARK
21	Fixed Ring Gear	350A32-1051-21 350A32-1086-21 350A32-1086-22 350A32-1086-23 350A32-1118-20 350A32-1119-20	20000 FH	Margin: 0
22	Planet Gear, Unequipped	350A32-1082-20 350A32-1082-21 350A32-1114-20 350A32-1114-21	20000 FH	Installed on planet gear MP/N 350A32-1082-03, 350A32-1082-04 and 350A32-1114- 00. Margin: 0
23	Planet Gear Cage	350A32-1089-21	44600 TC	Margin: 0
24	Bevel Wheel	350A32-3165-20 350A32-3166-20	20000 FH	Margin: 0
25	Input bevel pinion	350A32-3186-20 350A32-3186-25	20000 FH 215300 TC	Margin: 0
26	Main Housing Assembly	350A32-3156-02	76500 TC	Unequipped housing MP/N 350A32-3156-22 Margin: 0
27	MGB Bottom Housing	350A32-3119-05	63500 TC	Margin: 0
28	Sun Gear	350A32-1075-20 350A32-1075-21 350A32-1075-22	20000 FH	Margin: 0
29	Free Wheel Shaft	0.292.90.023.0	120000 TC	Fitted to the free wheel assembly MP/N 0.292.90.020.0 (P/N 704A43622004). Margin: 0
30	Free Wheel Flange	0.292.90.025.0	90000 TC	Fitted to the free wheel assembly: MP/N 0.292.90.020.0 (P/N 704A43622004). Margin: 0
31	Main Housing Assembly	350A32-3191-00	76500 TC	Unequipped housing MP/N 350A32-3191-20 Margin: 0
63-30 MGB MOUNT AND ATTACHMENT				
32	MGB suspension cross bar	350A38-1041-00 350A38-1041-20	160000 TC	One of these parts must be installed on helicopters carrying out external load carrying operations. Margin: 0

NO	DESCRIPTION	PART NUMBER	LIMITATION (SSL)	REMARK
63-32 MGB SUSPENSION BAR				
33	Front Bar	704A33-633-074 704A33-633-262	20000 FH	Margin: 0
34	Rear Bar	704A33-633-075	20000 FH	Margin: 0
35	Front Attachment Fitting	350A21-1104-03	20000 FH	Margin: 0
36	Rear Attachment Fitting	350A21-1105-03	20000 FH	Margin: 0
ATA 64 – TAIL ROTOR				
64-21 TAIL ROTOR HUB				
37	Tail Rotor Hub Assembly	350A33-3120-01	6900 FH	Margin: 0
38	Tail Rotor Blade Assembly	350A33-3002-02 350A33-3002-03 350A33-3002-04 350A33-3002-05 350A33-3004-00 350A33-3004-01	10000 FH	Margin: 0
ATA 65 – TAIL ROTOR DRIVE				
65-11 TAIL ROTOR DRIVE SHAFT				
39	Front Shaft Section Assembly	350A34-0151-00	20000 FH	Margin: 0
40	Center Shaft Section Tube Assembly	350A34-3020-00	2300 FH	Center shaft section assembly MP/N 350A34- 0214-00 Margin: 0
41	Center Shaft Section Tube Assembly	350A34-3020-01	8400 FH	Center shaft section assembly MP/N 350A34- 0214-01 Margin: 0
42	Front shaft assy	350A34-5010-00	40000FH	Margin: 0
43	Rear shaft assy	350A34-5020-00	40000FH	Margin: 0
44	Sliding flange	350A34-5035-20	40000FH	Margin: 0
45	Bearing support	350A34-5030-00	40000FH	Margin: 0
46	Flexible coupling assy	350A34-1000-01	40000FH	Margin: 0
65-21 TAIL GEARBOX				
47	Input Flange	350A33-5102-20	20000 FH	Margin: 0
48	Bevel Wheel	350A33-5111-20 350A33-5111-21	20000 FH	Margin: 0
49	Bevel Pinion	350A33-5118-20 350A33-5118-21	12400 FH	Margin: 0
50	Housing Assembly	350A33-5140-01	20000 FH	Margin: 0
ATA 67 – ROTORS FLIGHT CONTROL				
67-30 SERVO CONTROL				
51	Servo control Main	SA7450-100 (704A44831170) SA7650-100 (704A44831171) SC8042 (704A44831102)	20000 FH	Margin: 0

NO	DESCRIPTION	PART NUMBER	LIMITATION (SSL)	REMARK
		SC8043 (704A44831103)		

2.2.2. COMPONENTS AIRWORTHINESS INSPECTION (CHECK)

Component subject to an airworthiness inspection must be inspected periodically to confirm, that there is no deterioration, or that the deterioration found is within the removal. These components, as listed in table below must be removed from service when the specified limit is reached.

NO	DESCRIPTION	PART NUMBER	LIMITATION (CHECK)	REMARK
ATA 53 – FUSELAGE				
53-31 AFT FUSELAGE				
1	Cone skins junctions POST MOD 074581 Detailed inspection of junction of cone with composite spacer and junction of cone metallic skins. DI (AMM 53-31-01, 6-3)	-	2400 FH	Margin: 240 FH
ATA 55 – STABILIZER				
55-11 HORIZONTAL STABILIZER				
2	Tail boom / Fenestron junction frame POST MOD 074775 & PRE MOD 074581 Visual check for absence of cracks. GVI (AMM 55-11-00, 6-4)	-	600 FH	Margin: 60 FH
ATA 62 – ROTOR(S)				
62-11 MAIN ROTOR BLADES				
3	Main Rotor Blade Check of the skin. Check for Cracks. GVI (AMM 62-11-00, 6-3)	355A11-0030-04	150 FH	Margin: 15 FH
4	Main Rotor Blade Monitoring of the stainless steel shielding adhesion using tapping test. DI (AMM 62-11-00, 6-1)	355A11-0030-04	600 FH	Margin: 60 FH
62-21 MAIN ROTOR HEAD				
5	MRH When removing the upper or lower sleeve flanges, check the bushes at the	355A31-0002-01	-	-

NO	DESCRIPTION	PART NUMBER	LIMITATION (CHECK)	REMARK
	arm ends. When removing the rotor hub, check the bushings in the center of the star. DI (AMM 62-21-00, 6-16)			
6	Spherical bearing Check of the elastomer part. GVI (AMM 05-40-00, 6-7)	57910700 (704A33633211) LB4-1231-1 (704A33633208)	10 FH	Margin: 0
7	Starflex star Check. GVI (AMM 05-40-00, 6-7)	350A31-1918-00	50 FH	Margin: 10 FH
8	Frequency adapter Check of the elastomer part. Must not be mixed together. GVI (AMM 05-40-00, 6-7)	365A31-1019-25 (-) E-4165F01 (704A33640088) E-4165F11 (704A33640100)	10 FH	Margin: 0
ATA 63 – ROTOR DRIVE(S)				
63-21 MGB				
9	Epicyclic reduction gear - Non electrical magnetic plug Visual check. GVI (AMM 60-00-00, 6-2)	-	25 FH	Margin: 0
10	Epicyclic reduction gear - Electrical chip detector Visual check. GVI (AMM 60-00-00, 6-2)	-	50 FH	Margin: 0
11	MGB - Non electrical magnetic plug Visual check. GVI (AMM 60-00-00, 6-2)	-	25 FH	Margin: 0
12	MGB - Electrical chip detector Visual check. GVI (AMM 60-00-00, 6-2)	-	50 FH	Margin: 0
63-30 MGB MOUNT AND ATTACHMENT				
13	MGB suspension cross bar Remove and inspect. DI (AMM 63-31-00, 6-4)	350A38-1040-00 350A38-1040-20 350A38-1041-00 350A38-1041-20	3000 FH // 40000 TC	300 FH 0 TC
63-32 MGB SUSPENSION BAR				
14	Front and rear bars Check the condition of the MGB suspension bars during a MGB removal or whenever an operation requires suspension bar removal. DI (AMM 63-32-00, 6-1)	704A33-633-074 704A33-633-075 704A33-633-262	-	-

NO	DESCRIPTION	PART NUMBER	LIMITATION (CHECK)	REMARK
15	Laminated stop Check the condition of the laminated stops during a MGB removal or whenever an operation requires MGB suspension cross bar removal. DI (AMM 63-31-00, 6-1)	579069G (704A33633153) 579070H (704A33633152)	-	-
ATA 64 – TAIL ROTOR				
64-21 TAIL ROTOR HUB				
16	Torsion tie bar Tropical and damp atmosphere. Salt-laden atmosphere. Detailed check whatever the time spent in specific climatic conditions. Must not be mixed together. DI (AMM 64-21-00, 6-27)	350A33-3180-00 350A33-3180-01	1000 FH	Margin: 100 FH
17	Torsion tie bar Detailed check. Must not be mixed together. DI (AMM 64-21-00, 6-27)	350A33-3180-00 350A33-3180-01	1200 FH	Margin: 120 FH
ATA 65 – TAIL ROTOR DRIVE				
65-11 TAIL ROTOR DRIVE SHAFT				
18	Bearing Check without removal. GVI (AMM 65-11-00, 6-15)	593404 (704A33651181)	150 FH	Margin: 15 FH
19	Bearing Check and greasing bearing individually after disconnection bearing block. DI LUB (AMM 65-11-00, 3-1, 6-5)	593404 (704A33651181)	600 FH	Margin: 60 FH
20	Center shaft section sleeves Check. GVI (AMM 65-11-00, 6-12)	704A33-698-027	100 FH	Margin: 10 FH
65-21 TAIL GEARBOX				
21	TGB - Electrical chip detector Visual Check. GVI (AMM 60-00-00, 6-2)		50 FH	Margin: 0 FH

2.2.3. TIME BETWEEN OVERHAUL (TBO)

A TBO is assigned to a complete assembly. The interval corresponds to the operating time permitted before an overhaul is performed in a specialized workshop. The overhaul allows the assembly to be returned to service for a new period.

The interval corresponds to a period during which any increase in damage cannot be detected by routine maintenance (example: internal corrosion, fretting leading to the loosening of bolted assemblies, etc.)

NO	DESCRIPTION	PART NUMBER	LIMITATION (TBO)	REMARK
ATA 63 – ROTOR DRIVE(S)				
63-21 MGB				
1	Bevel reduction gear	350A32-0352-00	3000 FH / 288 M	Margin: 300 FH 180 D
2	Epicyclic reduction gear	350A32-0120-00	3000 FH / 288 M	Margin: 300 FH 180 D
3	Oil pump	350A32-0400-00	4000 FH	Margin: 300 FH
4	Free wheel assembly Free wheel ARRIEL 2D	0.292.90.020.0 (704A43622004)	4000 FH	Margin: 0
ATA 65 – TAIL ROTOR DRIVE				
65-21 TAIL GEARBOX				
5	TGB Assy	350A33-0502-00	3000 FH / 288 M	Margin: 300 FH 180 D
ATA 67 – ROTORS FLIGHT CONTROL				
67-30 SERVO CONTROL				
6	Main Servo Control	SA7450 (704A44831170) SA7650 (704A44831171) SC8042 (704A44831102) SC8043 (704A44831103)	3000 FH / 240 M TSM or TSO	Margin: 300 FH 180 D
ATA 80 – ENGINE STARTING				
80-00 STARTING				
7	Starter Generator (AUXILEC)	515-030 (704A46101018) 524-031 (704A46101011)	2400 FH	Margin: 100 FH
8	Starter Generator	150SG122Q-4 (704A46101027) 200SGL130Q-4 (704A46101026)	1200 FH	Margin: 120 FH

2.2.4. OPERATING TIME LIMIT (OTL)

Components whose possible failure would have lesser consequences on flight safety than components with a Service Life Limit, are assigned an Operating Time Limit. These components, as listed in table, must be removed from service when the specified limit is reached.

NO	DESCRIPTION	PART NUMBER	LIMITATION (OTL)	REMARK
ATA 21 – AIR CONDITIONING				
21-51 OPT – AIR CONDITIONING				
1	Elastic Belt	6PK637-SK801 (704A33690009)	3500 FH	Margin: 0
ATA 25 – EQUIPMENT / FURNISHINGS				
25-66 ELT				
2	Battery pack (Kannad Integra AP-H (ER))	S1854501-01 (704A45737078)	ED / 1 OPH	1 OPH = 1 hour of operation except for testing or operation for an unknown duration. Margin = 0
25-67 EMERGENCY FLOATATION GEAR				
3	Cylinder, Emergency Floatation Gear	200740-2 (704A42693012)	180 M (TSM)	Return the equipment to an approved repairer for overhaul and replacement of the bare cylinder. Margin = 0
4	Floats	217174-0 217195-0	180 M (TSM)	If an extension is required, return the assembly in question to the manufacturer for it to be examined. Margin = 0
5	Cover Assembly	217246-0 217247-0	180 M (TSM)	If an extension is required, return the assembly in question to the manufacturer for it to be examined. Margin = 180 D
ATA 26 – FIRE PROTECTION				
26-22 CABIN EXTINGUISHER				
6	Cabin Fire Extinguisher	12085-01 (S262A10T1001) H1-10AIR (704A32810008)	120 M (TSM)	Margin = 0

NO	DESCRIPTION	PART NUMBER	LIMITATION (OTL)	REMARK
ATA 62 – ROTOR(S)				
62-30 MAIN ROTOR MAST				
7	Bearing, Rotor mast	Y45RU10998S3M74 (704A33653110)	3000 FH	Margin = 300 FH
8	Bearing, Swashplate	Y51BB10843S2M74 (704A33651158)	6600 FH	Replace bolts MP/N 350A37- 1210-20 and 22129BC080066L on each removal of the bearing. Margin = 0
ATA 63 – ROTOR DRIVE(S)				
63-11 MGB / ENGINE COUPLING				
9	Hydraulic pump – Poly "V" belt	POLY-V 597K4	1800 FH / 72M	Margin = 180 FH / 180 D
10	Hydraulic pump drive bearing	594976 (704A33651269) F-594448 (704A33651243)	3600 FH / 72 M	Margin = 300 FH / 180 D
63-21 MGB				
11	Oil Filter	FA01315A (7050A3632296)	600 FH / 24M	Margin = 60 FH / 73 D
12	Free wheel	9.560.90.123.0	4000 FH	Free wheel ARRIEL 2D. Fitted to the free wheel assembly MP/N 0.292.90.020.0 (P/N 704A43622004) Margin = 0
13	Front ball bearing of free wheel	9.606.65.100.1	4000 FH	
14	Rear ball bearing of free wheel	9.606.68.100.1	4000 FH	
15	Rear ball bearing of the free wheel assembly	9.606.49.060.6	4000 FH	
ATA 65 – TAIL ROTOR DRIVE				
65-11 TAIL ROTOR DRIVE SHAFT				
16	Tail rotor drive bearing	593404 (704A33651181)	3600 FH / 72 M	Margin: 0

2.2.5. ON – CONDITION (OC)

A Component subject to an On-Condition maintenance must be inspected periodically to confirm that there is no damage or the damage found is within the removal or maintenance criteria specified in the documentation.

The component is kept in service until the next inspection. The component must be removed from service when it reaches the criterion for removal, or made serviceable again as per the instructions given in the documentation.

NO	DESCRIPTION	PART NUMBER	LIMITATION (OC)	REMARK
ATA 21 – AIR CONDITIONING				
21-51 OPT – AIR CONDITIONING				
1	Air conditioning - Compressor	200F54 (M215A10T0001)	OC	-
ATA 24 – ELECTRICAL POWER				
24-33 BATTERY				
2	Battery	150CH-1 (7050A4243027) 151CH-1 (7050A4243040)	OC	-
ATA 25 – EQUIPMENT AND FURNISHINGS				
25-00 EQUIPMENT AND FURNISHING				
3	Belt & Harness - Strap (Pilot, copilot, passenger and operator seat belt and harness straps)	ALL MP/N	OC	-
ATA 28 – FUEL				
28-20 FUEL DISTRIBUTION				
4	Fuel Pump	P99C16-623 (704A44510057)	OC	-
ATA 34 – NAVIGATION				
34-00 NAVIGATION				
5	Stand-by Battery	EE0033A (704A46140000)	OC	-
ATA 62 – ROTOR(S)				
62-21 MAIN ROTOR HEAD				
6	Spring-type vibration damper	350A31-0033-06	OC	-
7	MRH Assy	ALL MP/N	OC	-
62-30 MAIN ROTOR MAST				
8	Main Rotor Mast	355A37-0006-01	OC	-
ATA 63 – ROTOR DRIVE(S)				
63-30 MGB MOUNT AND ATTACHMENT				
9	MGB suspension cross bar	350A38-1040-00 350A38-1040-20 350A38-1041-00 350A38-1041-20	OC	-
10	Suspension assembly	350A38-0201-00	OC	-
ATA 64 – TAIL ROTOR				
64-21 TAIL ROTOR HUB				
11	TRH	350A33-3100-01	OC	-
ATA 65 – TAIL ROTOR DRIVE				
64-11 TAIL ROTOR DRIVE SHAFT				
12	Tail rotor drive assembly	350A34-0202-00 350A34-0202-01	OC	-

2.3. LIFE LIMITED COMPONENTS – ENGINE TURBOMECA ARRIEL 2D

2.3.1. AUTHORIZED IN-SERVICE LIFE LIMITS

The following table lists authorized in-service life limits of power plant components that are expressed in number of cycles. They represent the number of cycles that a given part may complete before it must be removed from service. Only parts whose rupture would place the aircraft at risk are concerned.

The number of cycles consumed between the start and subsequent shutdown of an engine is equal to the sum of complete cycles and partial cycles.

A complete cycle is a sequence of engine operations that includes: Start – Significant power increase – Shutdown. A partial cycle is an engine operation sequence corresponding to a significant in-flight turbine speed variation (N1 or N2), without shutting down the engine.

MODULE	DESCRIPTION	ITEM P/N	LIMITS (CYCLES)	COUNTERS
Module 02	Axial Compressor Wheel	2292153370	22,000	Gas Generator (CSN1)
Module 03	Centrifugal Impeller	0292260110	22,000	
	Injection Wheel	2292260750	8,000	
	HP Turbine Disc	2292260060	17,000	
Module 04	Power Turbine Disc	2292810790	22,000	Power Turbine (CSN2)

2.3.2. TIME BETWEEN OVERHAUL (TBO)

The TBO of engine, module, equipment or accessory is the maximum authorized time the material may be used under normal use conditions before it must be overhauled. Normal use conditions are conditions that conform to criteria defined by the manufacturer and the DGCA. TBO is expressed in flight hours or in calendar TBO (return of material once whichever max. threshold is reached first).

ENGINE AND MODULES SUBJECT TO TBO

ITEM	MODULE	PERIODICITY	REFERENCE COUNTER
72-61-00-01-001	M01 - Accessory Gearbox	5000	TSO
72-32-00-01-001	M02 - Axial Compressor	5000	TSO
72-43-00-01-001	M03 - Gas Generator	5000	TSO
72-54-00-01-001	M04 - Power Turbine	5000	TSO
72-15-00-01-001	M05 - Reduction Gear	5000	TSO

ENGINE EQUIPMENT AND ACCESSORIES SUBJECT TO ON CONDITION AND TBO

NO	EQUIPMENT & ACCESSORIES	PARTS CATALOG REFERENCE	MAINTENANCE METHOD	OC/TBO
1	Engine Electronic Control Unit (EECU)	71-00-00-01-003	On condition (R)	
2	Control and monitoring harness	71-51-00-01-010	On condition (R)	
3	Control harness	71-51-00-01-020	On condition (R)	
4	Igniter	72-43-00-01-460	On condition (E)	
5	Injector	72-43-00-01-550	On condition (E)	
6	Alternator	72-61-00-01-650	On condition (E)	
7	Torque meter sensor	72-61-00-01-880	On condition (E)	
8	Exhaust pipe	72-00-00-01-200	On condition (R)	
9	Adjusted valve assembly	73-14-00-01-010	On condition (R)	
10	Pump and Metering Unit assembly	73-23-00-01-010	TBO: Repairable	4.000 HRS / 10 Years
11	Fuel pressure and temperature transmitter	73-23-14-02-650	On condition (E)	
12	Fuel pressure transmitter	73-23-14-02-740	On condition (E)	
13	Fuel filter visual blockage indicator	73-23-14-02-780	On condition (E)	
14	Ignition Box	74-11-10-01-010	On condition (E)	
15	Bleed valve	75-31-00-01-050	On condition (R)	
16	P3 pressure transmitter	75-41-00-01-010	On condition (E)	
17	N1 speed sensor	77-11-00-01-010	On condition (E)	
18	N2 speed sensor	77-12-00-01-010	On condition (E)	
19	N2 speed sensor	77-12-00-01-020	On condition (E)	
20	N2 speed sensor	77-12-00-01-030	On condition (E)	
21	Pyrometric harness	77-21-00-01-010	On condition (E)	
22	Pyrometric harness	77-21-00-01-020	On condition (E)	
23	Engine Data Recorder	77-33-10-01-010	On condition (E)	
24	Oil pump	79-24-00-01-010	On condition (R)	
25	Valve assembly	79-25-00-01-010	On condition (R)	
26	Oil pressure transmitter	79-32-00-01-010	On condition (E)	
27	Oil pressure and temperature transmitter	79-33-00-01-010	On condition (E)	
28	Mechanical magnetic plugs	79-36-00-01-010	On condition (E)	
29	Mechanical magnetic plugs	79-36-00-01-030	On condition (E)	
30	Electrical magnetic plug	79-38-00-01-030	On condition (E)	
31	Free wheel assy equipped	83-12-00-01-001	TBO: Repairable	5.000 HRS

2.3.3. USE-LIMITED PARTS

Use limits, expressed in hours or cycles, have been defined to optimize the service life of certain components by keeping them in service for as long as possible without compromising flight safety (both prior to and following repairs). These use limits allow parts to have their service lives extended rather than be replaced during overhaul.

MODULE	PARTS DESCRIPTION	PART NUMBER	USE LIMIT
M03	HP Turbine Blade[2]	229226A2M0 229226A2N0 229226A560	10,000 Cyc (CSN1) / 261% (Creep Damage)[1]
M04	Power Turbine Blade	229281A440	6,000 Hrs (TSN)/ 10,000 Cyc (CSN2)[1]
	Power Turbine Nut	0292810450	6,000 Hrs (TSN)
M05	Sleeve Assy	0292717600	6,000 Hrs (TSN)
	Spline Nut	0292710510	6,000 Hrs (TSN)

NOTE:

1) Whichever reached first.
2) When the first limit is reached, remove M03, send it back to Turbomeca approved repair center for gas generator turbine blades replacement or replace the gas generator turbine (deep maintenance).

3.1. LINE MAINTENANCE PROGRAM

1. Check before the first flight of the day (BFF)

The purpose of this check is to confirm the airworthiness of the aircraft once it has been positioned on the takeoff area after elimination of possible failures that have been reported by the pilot in the aircraft log book and that are liable to affect the safety level of the aircraft. The interval between this check and the first flight of the day should be as short as possible.

2. Check after the last flight of the day (ALF)

The check after the last flight of the day is intended to confirm the serviceability of the aircraft for the flights scheduled for the next flying day.
This check must be performed after the last flight of the day without exceeding a 10 FH interval between two checks.

3. P-check

The P-check is intended to check the operational availability of the helicopter between two checks.

This check must be performed at the latest at 10 FH without exceeding 7 days.

There are three possible cases in which this check is to be triggered:

- The helicopter flies 10 FH within less than 7 days. The P-check must be performed at the latest at 10 FH.
- The helicopter flies less than 10 FH within 7 days. The P-check must be performed at the latest at 7 days.
- The helicopter is grounded during 7 days or more. The P-check must be performed before resuming flights.

4. Specific Scheduled Inspections

Some schedule inspections were possible to be performed at line of operation (non – hangarage). Prior approval by DGCA, it must be evaluated by Chief Inspector base on the difficulties, tools and equipment and environment of the area. List of Inspections which can be performed in Line Maintenance (non-hangarage) available in Company Maintenance Manual Chapter 3.9.4.

3.2. SCHEDULED INSPECTIONS – AIRFRAME

The helicopter and its component parts, accessories, and appliances shall be maintained in an airworthy condition in accordance with the maximum time limits, which have Direct Inclusion, hereinafter set forth for the accomplishment of the overhaul, periodic inspections, and its components parts, accessories, and appliances.

3.2.1. Scheduled maintenance of the **basic** equipment:

TYPE OF INSPECTION	CHAPTER/SECTION/SUBJECT	INTERVAL	MARGIN	FREQUENCY
P-check	05-20-02	10 FH // 7 D	0	Periodic
S Inspection	05-21-00	150FH// 12M	15FH // 36D	Periodic
S Inspection	05-21-01	150 FH	15FH	Periodic
S Inspection	05-21-02	12 M	36D	Periodic

TYPE OF INSPECTION	CHAPTER/SECTION/SUBJECT	INTERVAL	MARGIN	FREQUENCY
T or A Inspection	05-22-00	600FH//24M	60FH // 73D	Periodic
T Inspection	05-22-01	600FH	60FH	Periodic
A Inspection	05-22-02	24 M	73D	Periodic
2T or 2A Inspection	05-23-00	1200 FH // 48M	120FH // 146D	Periodic
2T Inspection	05-23-01	1200FH	120FH	Periodic
2A Inspection	05-23-02	48M	146D	Periodic
-	05-25-00	Specific	Ch: 3.3.1	Periodic
-	05-26-00	Specific	Ch: 3.3.1	Perform Once

3.2.2. Scheduled maintenance of the **optional** equipment:

TYPE OF INSPECTION	CHAPTER/SECTION/SUBJECT	INTERVAL	MARGIN	FREQUENCY
Daily checks	05-30-00	BFF // ALF	0	Periodic
P-check	05-30-02	10 FH // 7 D	0	Periodic
S Inspection	05-31-00	150FH//12M	15FH // 36D	Periodic
S Inspection	05-31-01	150 FH	15FH	Periodic
S Inspection	05-31-02	12 M	36D	Periodic
T or A Inspection	05-32-00	600FH//24M	60FH // 73D	Periodic
-	05-35-00	Specific	Ch: 3.3.2	Periodic
-	05-36-00	Specific	Ch: 3.3.2	Perform Once

3.3. SPECIFIC INTERVALS INSPECTIONS

The limits of this inspections must be complied with:

- **Periodically:** the maintenance operation must be performed at the latest when the indicated limit is reached.

  NOTE: For operational reasons, the maintenance operation can be performed before the limit is reached, however, the maximum interval between two maintenance operations must be complied with.

- **Once only:** The maintenance operation must be performed once only, when the indicated limit is reached.

  It must not be performed before the limit is reached.

  These operations can be performed either:

  - After installation of a component removed from the same helicopter,
  - After the introduction to service of a new, overhauled or repaired component,

- After installation of a component originating from another helicopter,
- Each time after the component is installed.

The start of the maintenance operation is specified on a routine basis in each task concerned.

3.3.1. Specific Periodic Inspection of **basic** and **optional** equipment:

NO	DESCRIPTION	REFERENCE	LIMITATION	MARGIN
ATA 21 – AIR CONDITIONING				
21-21 VENTILATION / DEMISTING				
1	**Air intakes – Radiator** Cleaning. Optional – In the event of intensive operation in salt-laden atmosphere, sand-laden and/or dust-laden atmosphere, tropical and damp atmosphere, all the necessary measures should be taken to ensure optimum operation of the aircraft, if necessary, by shortening the suggested intervals. CLN	AMM 21-00-00, 7-1	600 FH // 12 M	60 FH 36D
2	**Demisting or Heating** Check. VC|DI	AMM 21-21-02, 6-1 AMM 21-21-03, 6-1	144 M	180 D
***21-51 OPT - AIR CONDITIONING**				
3	**Freon air conditioning system** Check for correct operation if not used during 1 M. FT	AMM 21-51-02, 2-1	1M	3D
4	**Cockpit and cabin evaporator harnesses - Condenser** Cleaning. Optional – In the event of intensive operation in salt-laden atmosphere, sand-laden and/or dust-laden atmosphere, tropical and damp atmosphere, all the necessary measures should be taken to ensure optimum operation of the aircraft, if necessary, by shortening the suggested intervals. CLN	AMM 21-00-00, 7-1	600FH // 12M	60FH 36D
5	**Probe** TH90. Functional test. FT	AMM 21-51-02, 5-5	3000FH	0
ATA 24 – ELECTRICAL POWER				
24-00 ELECTRICAL POWER				
6	**Bonding** Check. DI	AMM 24-00-00, 6-2	144 M	180 D
7	**Battery** 150CH-1 (7050A4243027)	AMM 24-33-00, 6-1	6 M	18 D

NO	DESCRIPTION	REFERENCE	LIMITATION	MARGIN
	151CH-1 (7050A4243040) Hot climatic conditions Check. DI			
	ATA 25 – EQUIPMENT/FURNISHINGS			
25-11 COCKPIT SEAT				
8	**Energy absorption front seat** SICMA 159. Detailed check. DI	AMM 25-21-00, 6-7	1800 FH	180 FH
9	**Energy absorption front seat - Rails** SICMA 159. Check. DI	AMM 25-21-00, 6-2	2400FH	240FH
10	**Energy absorption front seat** SICMA 198. Detailed check. DI	AMM 25-21-01, 6-3	1800 FH	180 FH
11	**Energy absorption front seat - Rails** SICMA 198. Check. DI	AMM 25-21-00, 6-1	2400FH	240FH
25-21 EXECUTIVE FURNISHING				
12	**Rear seat** SICMA 284. Detailed check. DI	CMM 25-12-32	1800 FH	180 FH
13	**Rear seat - Fitting** SICMA 284. Detailed check. DI	AMM 25-22-01, 6-1	1800 FH	180 FH
25-66 EMERGENCY LOCATER TRANSMITTER				
14	**Emergency locator transmitter** KANNAD INTEGRA AP-H (ER). Self-test. FT	AMM 25-66-20, 5-1	1 M	3 D
***25-67 EMERGENCY FLOATATION GEAR**				
15	**Cylinder** 200740-2 (704A42693012) Proof-test. 1 OPC = 1 utilization (ditching or untimely percussion of the inflation cylinder). To be returned to the manufacturer or to an approved workshop. Interval starting from the date of manufacture (TSM), then from the date of the last proof test. NPT		60M // 1 OPC	0 0

NO	DESCRIPTION	REFERENCE	LIMITATION	MARGIN
16	**Emergency floatation gear** Check and functional test. DI|FT	AMM 25-67-00, 5-1 AMM 25-67-00, 6-1	600FH // 18M	60FH 54D
17	**Floats** Check. Interval starting from the date of manufacture. DI	CMM 25-69-20 INSPECTION/CHECK §4	18M	54D
***25-91 CARGO SWING INSTALLATION**				
18	**Cargo sling installation** Tropical and damp atmosphere Salt-laden atmosphere Cleaning. CLN	AMM 25-92-00, 3-2	15D	0
19	**Cargo sling installation** Normal climatic conditions Cleaning. In the event of intensive use of the cargo hook, it is recommended to reduce the interval by half. CLN	AMM 25-92-00, 3-2	1M	3D
20	**Cargo sling installation** Check and functional test. OPH = Operating hours logged with underslung loads. GVI|FT	AMM 25-92-00, 6-1	12M // 100 OPH	36D 10 OPH
21	**Cargo sling installation** Cleaning and greasing. In the event of intensive use of the cargo hook, it is recommended to reduce the interval by half. OPH = Operating hours logged with underslung loads. CLN|LUB	AMM 25-92-00, 3-3	500 OPH	50 OPH
22	**Cargo sling installation** Detailed check. OPH = Operating hours logged with underslung loads. DI	AMM 25-92-00, 6-4	60M // 1000 OPH	180D 100 OPH
***25-92 CARGO SLING**				
23	**Cargo sling** Check. DI|LUB	AMM 25-92-00, 6-3	144 M	180 D
ATA 26 – FIRE PROTECTION				
26-11 FIRE DETECTION				
24	**Engine fire detector** Checking the engine fire detector opening threshold. DI	AMM 26-11-00, 6-1	3500FH	300FH

NO	DESCRIPTION	REFERENCE	LIMITATION	MARGIN
26-22 CABIN EXTINGUISHER				
25	**Cabin fire extinguisher** H1-10AIR (704A32810008) Check. WGH	AMM 26-21-00, 6-1	6M	18D
ATA 28 – FUEL				
28-00 FUEL SYSTEM				
26	**Fuel tank** Tropical and damp atmosphere Check and maintenance. DI	MTC 20.08.06.401	3M	9D
28-11 STORAGE				
27	**Fuel tank** Check. DI	AMM 28-11-01, 6-1	144M	180D
28-41 INDICATING				
28	**Low fuel level warning** Functional test. FT	AMM 28-41-01, 5-2	300FH // 24M	3000FH 73D
ATA 29 – HYDRAULIC POWER				
29-00 HYDRAULIC POWER				
29	**Hydraulic system** **Salt-laden atmosphere** Filling / Draining of the hydraulic system. Servicing of the strainer and the filter. DRN	AMM 29-00-00, 3-6	300FH // 24M	3000FH 73D
ATA 32 – LANDING GEAR				
32-11 MAIN LANDING GEAR				
30	**Skid Type Landing Gear** Check. DI	AMM 32-11-00, 6-6	144M	180D
ATA 53 – FUSELAGE				
53-00 FUSELAGE				
31	**Cabin floor** Check. VC	AMM 53-00-00, 6-5	144M	180D
53-10 STRUCTURE				
32	**Fuel tank compartment** Check. DI	AMM 53-10-00, 6-3	144M	180D
53-31 AFT FUSELAGE				
33	**Tail boom-to-fuselage junction** Check. DI	AMM 53-31-00, 6-8	144M	180D
34	**Composite spacer** POST MOD 074581 Detailed inspection.	AMM 53-31-01, 6-2	2400FH	240FH

NO	DESCRIPTION	REFERENCE	LIMITATION	MARGIN
	DI			
35	**Intermediate structure/Tail boom coupling** PRE MOD 074581 Visual check and tightening torque check. GVI|TCK	AMM 53-31-00, 6-1	2500FH // 72M	250FH 180D
36	**Intermediate structure/Tail boom coupling** POST MOD 074581 Visual check and tightening torque check. GVI|TCK	AMM 53-31-00, 6-1	2400FH	240FH
53-41 FUSELAGE				
37	**Fenestron** Check. DI	AMM 53-41-00, 6-2	144M	180D
ATA 60 – STD. PRACTS. PROP/ROTOR				
60-00 GENERAL - MECHANICS SYSTEM				
38	**MRH - Main Rotor Mast - TRH - Engine/MGB coupling** Operation to be carried out after having logged 144 M (margin 180 D) after initial setting to service. DI	AMM 05-60-00, 6-1	72M	180D
39	**MGB assembly - TGB assembly** Check. Operation to be carried out after having logged 144 M (margin 180 D) after initial setting to service or since the last Overhaul. DI	AMM 05-60-00, 6-1	72M	180D
ATA 62 – ROTOR(S)				
62-11 MAIN ROTOR BLADES				
40	**Main rotor blade** Sand-laden and/or dust-laden atmosphere Check for erosion. In the event of intensive operation in heavily sand-laden areas, all the necessary measures should be taken to ensure optimum operation of the aircraft. GVI	AMM 62-11-00, 6-1	15FH	1FH
41	**Main rotor blade** Tropical and damp atmosphere Cleaning the blades. CLN	AMM 62-11-00, 3-1	30FH	3FH

NO	DESCRIPTION	REFERENCE	LIMITATION	MARGIN
42	**Main rotor blade** Salt-laden atmosphere Cleaning the blades. After hovering or low-altitude flight, this operation should be carried out during the P check. CLN	AMM 62-11-00, 3-1	30FH	3FH
62-21 MAIN ROTOR HEAD				
43	**Spring-type vibration damper** Check. DI	AMM 62-21-00, 6-11	3000FH // 72M	300FH 180D
62-30 MAIN ROTOR MAST				
44	**Rotor mast - Lower section** Detailed check. DI	AMM 62-31-00, 6-5	72M	180D
45	**Swashplates** Visual check of guide. GVI	AMM 62-32-00, 6-2	300FH	30FH
46	**Swashplates** Tactile check of balljoint. GVI	AMM 62-32-00, 6-1	300FH	30FH
ATA 63 – ROTOR DRIVE (S)				
63-30 MGB MOUNT AND ATTACHMENT				
47	**Laminate suspension blocks** Remove and inspect. DI	AMM 63-31-00, 6-1	3000FH // 72M	300FH 180D
48	**Laminate suspension block supports** Remove and inspect. DI	AMM 63-31-00, 6-2	3000FH //72M	300FH 180D
49	**Laminate suspension blocks - Attachment bolts** Remove and inspect. DI	AMM 63-31-00, 6-3	3000FH // 72M	300FH 180D
63-32 MGB SUSPENSION BAR				
50	**MGB suspension bar fittings** Check. DI	AMM 63-32-00, 6-6	144M	180D
67 – ROTORS FLIGHT CONTROL				
67-00 ROTOR FLIGHT CONTROLS				
51	**Rotor flight control channels** Check. DI	AMM 67-00-00, 6-2 AMM 67-00-00, 6-4	144M	180D
71 – POWER PLANT				
71-41 ENGINE SUPPORT BRACKER				
52	**Engine support** Check. DI	AMM 71-41-00, 6-1	144M	180D
77 – ENGINE INDICATING				
77-00 ENGINE PARAMETER INDICATORS				

NO	DESCRIPTION	REFERENCE	LIMITATION	MARGIN
53	**Engine parameter indicators** Functional test. FT	AMM 77-00-00, 5-1	144M	180D
79 – ENGINE OIL				
79-21 DISTRIBUTION				
54	**Oil Cooler** AMM 79-21-00, 3-1 Cleaning. CLN	AMM 79-21-00, 3-1	144M	180D
80 - ENGINE STARTING				
80-00 STARTING				
55	**Starter generator** 515-030 (704A46101018) 524-031 (704A46101011) **THALES.** Check of radial play. DI	AMM 80-00-02, 6-4	300FH	30FH

NOTE : * = Optional equipment.

3.3.2. Specific Perform Once of **basic** and **optional** equipment:

NO	DESCRIPTION	REFERENCE	LIMITATION	MARGIN
ATA 25 – EQUIPMENTS/FURNISHINGS				
***25-67 EMERGENCY FLOATATION GEAR**				
1	**Emergency floatation gear** Detailed check and proof test. Interval starting from the date of manufacture. NPT	CMM 25-69-20 INSPECTION/CHECK §5	**PO** 72M 108 M 144 M	 180D 180 D 180 D
ATA 28 – FUEL				
28-11 STORAGE				
2	**Fuel tank** Each time after the component is installed. Adjustment - Strap tension. RTQ	AMM 28-11-01, 5-1	**PO** 2FH	8FH
53-00 FUSELAGE				
3	**Tail boom/Rear structure connecting** Each time after the component is installed. Check. DI	AMM 53-31-00, 6-1	**PO** 30FH	3FH
ATA 62 – ROTOR(S)				
62-11 MAIN ROTOR BLADES				
4	**Main rotor blade** After the introduction to service of a new, overhauled or repaired component. Check.	AMM 62-11-00, 6-1	**PO** 30FH & 150FH & 300FH	 5FH

NO	DESCRIPTION	REFERENCE	LIMITATION	MARGIN
	GVI		& 450FH	15FH 15FH 15FH
62-21 MAIN ROTOR HEAD				
5	**Spherical thrust bearing, Frequency adapter/sleeve** Each time after the component is installed. Tightening torque check. TCK	AMM 62-21-00, 6-20	**PO** 2FH	8FH
6	**Assembly - MRH** Each time after the component is installed. Stop ring lubrication. LUB	AMM 62-21-00, 4-2	**PO** 2FH	8FH
62-30 MAIN ROTOR MAST				
7	**Mast/Hub coupling** Each time after the component is installed. Tightening torque check. TCK	AMM 62-21-00, 6-19	**PO** 2FH	8FH
8	**Pitch change link fitting bolt** Each time after the component is installed. Tightening torque check. TCK	AMM 62-33-00, 6-6	**PO** 2FH	8FH
9	**Main rotor mast** Each time after the component is installed. Tightening torque check: – upper casing/flared housing attachment bolts, – flared housing/MGB main casing attachment bolts, – swashplate guide-to-upper casing attachment bolts, – scissors drive coupling. TCK	AMM 62-31-00, 6-8 AMM 62-31-00, 6-9 AMM 62-31-00, 6-10 AMM 62-31-00, 6-11	**PO** 2FH	8FH
10	**Main rotor mast** Each time after the equipment is installed. Readjustment of the tightening torque: – of the scissors hinge attachments (link between rotating scissors / scissors drive – link between non-rotating scissors / swashplate guide	AMM 62-33-00, 6-7	**PO** 2FH	8FH

NO	DESCRIPTION	REFERENCE	LIMITATION	MARGIN
	– connection between scissors links), – of the delta-shaped ball joint attachments with the swashplates. RTQ			
	ATA 63 – ROTOR DRIVE (S)			
63-21 MGB				
11	**MGB** After the introduction to service of a new, overhauled or repaired component. Filling/Draining. DRN	AMM 12-10-00, 3-1	**PO** 30FH	3FH
63-30 MGB MOUNT AND ATTACHMENT				
12	**Suspension bars and cross bar** Each time after the component is installed. Tightening torque check. TCK	AMM 63-31-00, 6-7 AMM 63-32-00, 6-5	**PO** 2FH	8FH
	ATA 64 – TAIL ROTOR			
64-21 TAIL ROTOR HUB				
13	**Tail rotor** Each time after the component is installed. Tightening torque check. TCK	AMM 64-21-00, 6-32	**PO** 2FH	8FH
	ATA 65 – TAIL ROTOR DRIVE			
65-11 TAIL ROTOR DRIVE SHAFT				
14	**Bearing** Each time after the component is installed. Lubrication. LUB	AMM 65-11-00, 3-1	**PO** 2FH	8FH
15	**Front shaft assy** POST MOD 079809 Each time after the component is installed. Check. FT	AMM 65-11-01, 6-11	**PO** 2FH	8FH
16	**Bearing block assembly** POST MOD 079809 Each time after the component is installed. Check. DI	AMM 65-11-01, 6-12	**PO** 2FH	8FH
65-21 TAIL GEARBOX				
17	**TGB** Each time after the component is installed. Check - Attachment bolts. DI	AMM 65-21-00, 6-1	**PO** 2FH	8FH

NO	DESCRIPTION	REFERENCE	LIMITATION	MARGIN
18	**TGB** After the introduction to service of a new, overhauled or repaired component. Filling/draining. DRN	AMM 12-10-00, 3-1	**PO** 30FH	3FH
	67 – ROTORS FLIGHT CONTROL			
67-30 SERVOCONTROL				
19	**Main servocontrol** Each time after the component is installed. Readjust the tightening torque of servocontrol attaching bolts. RTQ	AMM 67-31-00, 6-4	**PO** 2FH	8FH
	80 – ENGINE STARTING			
80-00 STARTING				
20	**Starter generator** 150SG122Q-4 (704A46101027) 200SGL130Q-4 (704A46101026) SKURKA. After the introduction to service of a new or overhauled component. Checking generator and brushes. DI	AMM 80-00-01, 6-1 AMM 80-00-01, 6-3 AMM 80-00-01, 6-5	**PO** 300FH	30FH

NOTE : * = Optional equipment.

3.4. SCHEDULED INSPECTION – ENGINE

Schedule Inspections are performed at regular intervals with an allocated tolerance. Maintenance operations must be performed before the maximum frequency (frequency + tolerance) is reached. Scheduled engine inspections consist of the following interval:

1) Inspection Before the first flight of the day, Tolerance: Not available.
2) Turn-Around Inspection (between flight), Tolerance: Not available.
3) Inspection after 15 flight hours or 7 days, Tolerance: Not available.
4) Inspection at 25 flight hours, Tolerance: 2.5 FH.
5) Inspection at 300 flight hours, Tolerance: 30 FH.
6) Inspection at 800 flight hours, Tolerance: 80 FH.
7) Inspection at 4,000 flight hours, Tolerance: Not available.

NO	ENGINE INSPECTION DESCRIPTION	REFERENCE	LIMITATION / INTERVAL	TOLERANCE
1	Remove the blanks and make sure that there are no foreign objects: examine near the air intakes and the exhaust zone.	-	FFD	N/A
2	Inspection of oil level in tank and top up if required.	-	FFD	N/A

NO	ENGINE INSPECTION DESCRIPTION	REFERENCE	LIMITATION / INTERVAL	TOLERANCE
3	Make sure that nothing is flowing through the main drain of the engine deck.	-	BETWEEN FLIGHT	N/A
4	In the engine log book, record the number of C1 and C2 cycles. Note: If the automatic counting is not available	-	BETWEEN FLIGHT	N/A
5	Inspection of oil level in tank and top up if required.	-	BETWEEN FLIGHT	N/A
6	Record the corrective maintenance operations performed in the engine logbook, if any.	-	15 FH // 7 D	N/A
7	Make sure that the automatic cycle counting is correct.	EMM: 05-10-02-200-801	15 FH // 7 D	N/A
8	In the engine log book, record the total number of C1 and C2 cycles consumed as counted by the EECU.	EMM: 05-10-02-200-801	15 FH // 7 D	N/A
9	Visually examine the engine and the engine floor for leakage.	-	15 FH // 7 D	N/A
10	Visually examine the engine attachments and fire protections for signs of impact or deterioration Note : Fire protections of shut off valve and HMU	-	15 FH // 7 D	N/A
11	Record the values from the creep damage counter in the engine logbook.	EMM: 05-15-00-200-801	15 FH // 7 D	N/A
12	Record in the engine logbook the accumulation of flying hours.	EMM: 05-15-00-200-801	15 FH // 7 D	N/A
13	Check that the HP gas generator rotates freely (no abnormal noises) and visually check that the engine is in good condition. Remarks: Manually or during a dry crank cycle	-	15 FH // 7 D	N/A
14	Manually check that the power turbine rotates freely (no abnormal noises).	-	15 FH // 7 D	N/A
15	Inspection of oil level in tank and top up if required. Remarks: Within 15 minutes following engine shut-down.	-	15 FH // 7 D	N/A
16	Make sure that there are no foreign objects: examine near the air intakes and the exhaust zone. Install the blanks.	-	15 FH // 7 D	N/A

NO	ENGINE INSPECTION DESCRIPTION	REFERENCE	LIMITATION / INTERVAL	TOLERANCE
17	Engine health inspection. Note: Check also P0/T0 (Engine/Aircraft) consistency.	-	25 FH	2.5 FH
18	Module 02 - Axial compressor - Erosion check. Operating condition: In a sandy air	EMM: 72-00-32-200-801	300 FH	30 FH
19	Boroscope inspection of combustion chamber including nozzle guide vane.	EMM: 72-00-43-200-802	800 FH	80 FH
20	Functional check in the auxiliary mode. EBCAU test. Application conditions: Ground run	-	800 FH	80 FH
21	Module 03 - Turbine-casing drain valve assembly - Test. Application conditions: Ground run	EMM: 71-71-00-700-801	800 FH	80 FH
22	Module 03 - Checking of fluids discharged during injection wheel bleeding and checking of fluids discharged in tank fuel return pipe. Application conditions: Ground run	EMM: 73-14-00-700-802	800 FH	80 FH
23	Module 02 - Axial compressor - Erosion check. Operating condition: Not used in a sandy air	EMM: 72-00-32-200-801	800 FH	80 FH
24	Mechanical magnetic plug - Magnetism test.	EMM: 79-36-00-700-801	800 FH	80 FH
25	Test of the magnetism of the electrical magnetic plug.	EMM: 79-38-00-700-801	800 FH	80 FH
26	Test of the electrical magnetic plug.	EMM:79-38-00-750-801	800 FH	80 FH
27	Module 03 - Permeability check of the injection wheel.	EMM:72-00-43-200-801	800 FH	80 FH
28	Replacement of the fuel filtering element.	EMM:73-23-14-900-802	800 FH	80 FH
29	Bleed valve filter - Check and inspection. Component standard: Pre TU 94	EMM:75-31-00-200-802	800 FH	80 FH
30	Bleed valve filtering kit – Filter element replacement. Component standard: Post TU 94	EMM:75-61-00-950-801	800 FH	80 FH
31	Check of mechanical magnetic plug.	EMM: 79-36-00-900-801	800 FH	80 FH
32	Functional test of oil pressure transmitter.	EMM: 79-32-00-750-801	800 FH	80 FH
33	Functional test of oil pressure and temperature transmitter.	EMM: 79-33-00-750-801	800 FH	80 FH
34	Replacement of the oil filtering element.	EMM: 79-21-00-900-801	800 FH	80 FH

NO	ENGINE INSPECTION DESCRIPTION	REFERENCE	LIMITATION / INTERVAL	TOLERANCE
35	Module 03 - Preformed packing of rear bearing ducts - Replacement. Component standard: Pre TU 181	EMM: 72-43-00-900-806	800 FH	80 FH
36	Module 03 - Inspection of the ignition system. Application conditions: Engine running	EMM: 72-43-00-200-806	800 FH	80 FH
37	Module 03 - Inspection of the igniter plugs. Application conditions: Engine running	EMM: 72-43-00-200-803	800 FH	80 FH
38	Exhaust pipe - Check and inspection.	EMM: 72-70-00-200-801	800 FH	80 FH
39	Module 01 - Engine front support - Check and inspection.	EMM: 72-61-00-200-805	800 FH	80 FH
40	Inspection of pyrometric harness.	EMM: 77-21-00-210-801	800 FH	80 FH
41	Boroscope inspection of HP turbine blades and HP turbine nozzle guide vane.	EMM:72-00-43-200-803	800 FH	80 FH
42	Oil system - Draining. Operating condition: In a sandy air or with the use of 3 cSt oil.	EMM: 79-00-00-610-801	800 FH // 12 M	80 FH // 1 M
43	Oil system - Draining. Operating condition: With the use of 5 cSt oil.	EMM: 79-00-00-610-801	800 FH // 24 M	80 FH // 2 M
44	Remove and visually inspect power turbine pressurization pipe with magnifying glass x6: If pipe is cracked: replace it, And If presence of through-going crack: discard power turbine wheel	EMM: 75-29-00-900-805	Every 2,000 from 4,000	N/A
45	Inspect constant ΔP valve (including membrane replacement and functional test of auxiliary valve). Note: Remove HMU and send it back to a Turbomeca approved repair center.	EMM: 73-23-00-900-801	4000 FH	N/A
46	Check the start injector purge valve of Adjusted Valve Assembly.	EMM: 73-14-00-700-801	4000 FH	N/A

NO	ENGINE INSPECTION DESCRIPTION	REFERENCE	LIMITATION / INTERVAL	TOLERANCE
47	Remove and send Module 03 back to Safran Helicopter Engines approved repair center or apply Maintenance Technical Instruction to send back HP turbine wheel only for visual inspection of HP turbine wheel fir-tree roots for clogging. If one of the fir-tree root is completely clogged, discard HP turbine disc.	EMM: 72-00-43-900-801	6,000 FH	N/A
48	Remove engine or Module 03 and send it back to Safran Helicopter Engines approved repair center for checking that the HP turbine containment shield is free from crack using dye penetrant inspection. If presence of crack, discard the HP turbine containment shield.	EMM: 72-00-43-900-801	6,000 FH	N/A
49	Remove engine or Module 03 and send it back to Safran Helicopter Engines approved repair center for checking that the power turbine containment shield is free from crack using dye penetrant inspection. If presence of crack, discard the power turbine containment shield.	EMM: 72-00-43-900-801	6,000 FH	N/A
50	Functional test of oil pressure transmitter.	EMM: 79-32-00-750-801	6,000 FH	N/A
51	Functional test of oil pressure and temperature transmitter.	EMM: 79-33-00-750-801	6,000 FH	N/A
52	Remove and send the Module 03 to a certified Maintenance Center for calendar inspection.	EMM: 71-02-16-280-801	15 Years	1.5 Years

3.5. UNSCHEDULED MAINTENANCE CHECKS

Unscheduled maintenance which is listed in Airbus Helicopters EC 130 T2 Ch 05-50-00 and Engine Arriel 2D Maintenance Manual Ch 05-50-00 will be carried out when any of the events occur.

NO.	UNSCHEDULED MAINTENANCE CHECK	REFERENCE
1.	Detection of Chips illumination of “MGB P” and / or “MGB TEMP” warning lights – TGB / MGB	AMM 05-50-00, 6-1
2.	Rotor Overspeed.	AMM 05-50-00, 6-3
3.	Hard Landing	AMM 05-50-00, 6-5
4.	Impact on a Main Rotor Blade	AMM 05-50-00, 6-6

5.	Impact on Tail Rotor Blade	AMM 05-50-00, 6-7
6.	Non-rotating Blades in Gusts	AMM 05-50-00, 6-8
7.	Flight in strong turbulence	AMM 05-50-00, 6-9
8.	Aircraft struck by lightning	AMM 05-50-00, 6-10
9.	Sudden application of the main rotor brake	AMM 05-50-00, 6-11
10.	Jerks on the Freewheel	AMM 05-50-00, 6-12
11.	Ditching	AMM 05-50-00, 6-13
12.	Oil leaks from the gearboxes	AMM 05-50-00, 6-14
13.	Clogging of the hydraulic system (P/N BFS 156x)	AMM 05-50-00, 6-15a
14.	Fuel contamination	AMM 05-50-00, 6-16
15.	Engine-health check – No Torque Margin	AMM 05-50-00, 6-17
16.	Contaminated Engine Oil Circuit	AMM 05-50-00, 6-19
17.	Abnormal behavior of helicopter on the ground, with rotor spinning.	AMM 05-50-00, 6-20
18.	Diagnosis of defects through vibration analysis	AMM 05-50-00, 6-21a
19.	Diagnosis of defects through vibration analysis using STEADY Control adjustment.	AMM 05-50-00, 6-21b
20.	Immersion	AMM 05-50-00, 6-24
21.	T4.5 Overshooting	EMM 71-00-01-940-801
22.	N1 Overspeed	EMM 71-00-01-940-801
23.	N2 Overspeed	EMM 71-00-01-940-801
24.	Overtorque	EMM 71-00-01-940-801
25.	Extinction of a fire or accidental release of an extinguisher on a hot turboshaft engine	EMM 71-02-01-280-801
26.	Accidental release of an extinguisher on a cold turboshaft engine, without any extinguishing product entering the air path	EMM 71-02-01-280-802
27.	Foreign object damage	EMM 71-02-03-280-801
28.	Lightning strike	EMM 71-02-04-280-801
29.	Hard landing	EMM 71-02-06-280-801
30.	Surge	EMM 71-00-06-813-801
31.	Abnormal evolution of the oil pressure	EMM 71-00-06-814-808
32.	Rotor Damage	EMM 71-02-05-280-801
33.	Break of engine linking tube / MGB	EMM 71-02-05-280-802
34.	Fire	EMM 71-02-01-280-803
35.	Water dumping	EMM 71-02-01-280-804
36.	Corrosive or saline atmosphere	EMM 71-01-00-940-801
37.	Contaminated atmosphere	EMM 71-01-00-940-801
38.	Accidental dropping of turboshaft engine	EMM 71-02-280-801
39.	Oil system drainage	EMM 79-00-00-610-801
40.	Oil system rinsing	EMM 79-00-00-610-802

3.6. CASR REQUIREMENTS

1. Weight and Balance

Re-Weighing of the aircraft should be accomplished when:

1) Every 5 (Five) years.
2) After Paint stripping and repainting.
3) After any modification which significantly affecting empty weight.
4) If there is any complaint from the pilot flying the aircraft regarding the aircraft stability.
5) Empty weight of the aircraft has been changed more than 0.5% of the maximum take-off weight or empty center of gravity (C.G. location) has been change more than 0.5% of Mean Aerodynamic Chord (M.A.C.).
6) Specially required by the Directorate General of Civil Aviation (DGCA).

Accomplishment of the Weight and Balance must be supervised by a person appointed and authorized by PT. SMART CAKRAWALA AVIATION. Weight and balance, aircraft weight and c.g determination (Form No SCA/MTC/025) should be issued.

2. Magnetic Compass Calibration

Compass swing (calibration) is to be accomplished at the following times or any of the following reasons:

1) Whenever required by any maintenance procedure in Technical Publication after component replacement,
2) Whenever the accuracy of a Compass is suspected,
3) After any modification is carried out which may affect the compass system,
4) After lightning strike,
5) Fixed parking in one direction more than 12 months

Accomplishment of the Compass Swing must be supervised by a person appointed and authorized by PT. SMART CAKRAWALA AVIATION. After accomplishment of the Compass Swing, the maintenance release in the Aircraft Flight & Maintenance Log should be signed and the Compass Swing report should be issued.

(see form: SCA/MTC/026).

3. ATC Transponder Inspection

The inspection should be completed every 24 calendar months (No grace period).

This Inspection must be performed and released by an authorized person. After accomplishment of this Inspection and its discrepancies rectification, the maintenance release in the Aircraft Flight & Maintenance Log and the appropriate form should be signed.

4. Altimeter and Pitot Static Test

The inspection should be completed every 24 calendar months (No grace period).

This Inspection must be performed and release by an authorized person. After accomplishment of this Inspection and its discrepancies rectification, the maintenance release in the Aircraft Flight & Maintenance Log and the appropriate form should be signed.

5. Emergency Locator Transmitter

The inspection should be completed every 12 calendar months (No grace period).

This Inspection must be performed and release by an Authorized person. After accomplishment of this Inspection and its discrepancies rectification, the maintenance release in the Aircraft Flight & Maintenance Log and the appropriate form should be signed.

3.7. REQUIRED INSPECTION ITEM (RII)

If as a result of the application of this program, any part of either the main or any associated system is dismantled, adjusted, repaired or renewed, that part of the system(s) which has been disturbed shall be designated as a REQUIRED INSPECTION ITEM, (RII). Throughout this program, as far as possible all inspections which would require an RII have been identified but as a further guide work which disturbs the following systems would require RII. The RII must be carried out by a qualified person who holds RII authorization in accordance with the Smart Aviation CMM and must be other than the person who performed the work.

3.8. LIST OF REQUIRED INSPECTION ITEM

ATA	REQUIRED INSPECTION ITEM
21	**AIR CONDITIONING**
	a. Air conditioning Installation.
28	**FUEL SYSTEM**
	a. Installation and adjustment of fuel system components. b. Calibration of fuel quantity indicating system. c. Replacement/reinstallation or repair of fuel tank.
28	**FUEL SYSTEM**
	Installation of hydraulic pumps.
32	**LANDING GEAR**
	Installation of skid.
34	**NAVIGATION**
	Replacement, reinstallation or repair of components in the pitot/ static system. It also includes lines pneumatic airspeed indicators, pneumatic altimeter, compass swing (when adjustment required).
53	**FUSELAGE**
	Major, repair, major alteration, replacement / reinstallation of primary structure components.
62	**MAIN ROTOR SYSTEM**
	a. Installation of Main Rotor hub. b. Installation of Main Rotor Blade. c. Installation and rigging of Main Rotor control system component.

63 MAIN ROTOR DRIVE SYSTEM

a. Installation of Main Drive Shaft.
b. Installation of Main Rotor Transmission.
c. Installation of Main Rotor Mast.

64 TAIL ROTOR SYSTEM

a. Installation of Tail Rotor Hub and Blade assembly.
b. Installation and rigging of Tail Rotor control system components.

65 TAIL ROTOE DRIVE SYSTEM

a. Installation Tail Rotor gearbox.
b. Installation of Tail Rotor drive shafts.
c. Installation of Hanger Bearings assemblies.

67 ROTOR FLIGHT CONTROL

Major repair, rigging/adjustment, replacement/reinstallation of flight controls including related components e.g. cables, pulleys, linkages and hinges.

71 POWER PLANT

Installation of engine assembly.

72 ENGINE

Replacement of engine modules.

73 ENGINE FUEL AND FUEL CONTROL

Replacement or installation of the fuel control unit, engine driven fuel pump. governor and fuel lines.

76 ENGINE CONTROL

Rigging/adjustment, replacement/reinstallation of power plant controls.

Reg. Mark : PK - Station :

Date : FML No :

Do a special check for the following optional equipment:						
ITEM CODE NO.	**TASK**	**INITIAL**		**ITEM CODE NO.**	**TASK**	**INITIAL**
Emergency Floatation Gear						
BFF001	Fittings on bars: Condition, attachment, safety pins installed.			BFF003	Inflating pipe: Condition, attachment.	
BFF002	Protective cover: Condition, correct lacing.			BFF004	Cylinder assembly: Condition, attachment, correct pressure.	
KANNAD INTEGRA AP-H Emergency Locator Transmitter						
BFF005	Make sure that the remote control "ON-ARMED-TEST/RESET" switch is set to "ARMED".			BFF006	Make sure that the ELT "ARM-ON-OFF" switch is set to "ARM".	
Cargo-sling installation – Do the following checks before the first slinging operation of the day						
BFF007	Make sure that the load indicator is operational.			BFF011	Visually check that the elastomer stops attached to the gimbal joint are in good condition (no missing parts or significant tears).	
BFF008	Visually check that the hook is in good condition (no corrosion, impact marks, etc.).			BFF012	Visually check that the hydraulic pipe and its connections are in good condition.	
BFF009	Move the hook assembly in all directions and make sure that the electrical harnesses and the hydraulic hose do not prevent the hook assembly from moving freely.			BFF013	Check for leaks on all the visible parts of the hydraulic release system in the cockpit and outside, under the aircraft.	
BFF010	Check that the hook, load sensor and gimbal joint rotate freely.			BFF014	Check that the electrical harnesses and their connections are in good condition.	
BFF015	Check for correct operation of the primary release control: a) Install a load (2,5 kg (5.6 lb.) minimum) on the hook. b) Press the "SLING" push button on the control unit. **CAUTION** **DO NOT ACTUATE THE "CARGO REL" BUTTON FOR MORE THAN 20 SECONDS TO PREVENT OVERHEATING OF AND PERMANENT DAMAGE TO THE RELEASE SOLENOID LOCATED INSIDE THE HOOK.** c) Press the "CARGO REL" button on the cyclic stick. d) Make sure that the load is released. **WARNING** **IF THE HOOK DOES NOT OPEN, DO NOT USE THE HOOK UNTIL THE PROBLEM HAS BEEN SOLVED.** e) Close the hook by hand after its opening.					

<table>
<tr><th>ITEM CODE NO.</th><th>TASK</th><th>INITIAL</th><th></th><th>ITEM CODE NO.</th><th>TASK</th><th>INITIAL</th></tr>
<tr><td>BFF016</td><td colspan="5">Check for correct operation of the secondary release control
a) Install a load (2,5 kg (5.6 lb) minimum) on the hook.
b) Push on the lever attached to the collective pitch lever in the cockpit.
c) Make sure that the control operates with no hard points.
d) Make sure that the load is released.
WARNING
IF THE HOOK DOES NOT OPEN, DO NOT USE THE HOOK UNTIL THE PROBLEM HAS BEEN SOLVED.
e) Close the hook by hand after its opening.
f) Make sure that the hook locking indicator located on the side of the hook returns to the "fully closed" position: In the "fully closed" position, the indicator must be aligned with the marks engraved on the hook flange.</td><td></td></tr>
<tr><td>BFF017</td><td colspan="5">Make sure that the hydraulic fluid level in the hydraulic emergency release control reservoir is correct:
a) Move the collective pitch lever to the low position.
b) Make sure that the hydraulic fluid level is still between the "MIN" and "MAX" marks on the reservoir.</td><td></td></tr>
<tr><td>BFF018</td><td colspan="5">Make sure that the hydraulic emergency release control operates correctly as indicated in 25-50-97 CMM, "Hydraulic system operation check" paragraph.
a) If a problem is detected when checking the position of the push rod, bleed the hydraulic system as per 25-50-97 CMM, "Hydraulic system bleeding procedure" paragraph.</td><td></td></tr>
<tr><td>BFF019</td><td colspan="5">Check that the storage system operates correctly by making sure that the gimbal joint returns to its original position and bears against the front stops.</td><td></td></tr>
<tr><td colspan="7">APPAREO Imaging and Flight Data Monitoring Device</td></tr>
<tr><td>BFF020</td><td colspan="5">Check the LED status condition:
<table>
<tr><th>Condition</th><th>Status</th><th>Configuration</th></tr>
<tr><td>1</td><td>Red</td><td>Fault</td></tr>
<tr><td>2</td><td>Blue</td><td>Starting</td></tr>
<tr><td>3</td><td>Green</td><td>Operational</td></tr>
<tr><td>4</td><td>Yellow</td><td>SD card not inserted
or SD card not formatted to FAT32 format
or GPS locking signal not received</td></tr>
<tr><td>5</td><td>LED not lit</td><td>Does not operate.</td></tr>
</table></td><td></td></tr>
<tr><td>BFF021</td><td>If faults are found, apply the procedure described in (AMM 25-10-40,1-1).</td><td></td><td></td><td></td><td></td><td></td></tr>
<tr><td colspan="7">Engine Area</td></tr>
<tr><td>BFF022</td><td>Remove the blanks and make sure that there are no foreign objects: examine near the air intakes and the exhaust zone.</td><td></td><td></td><td>BFF023</td><td>Inspection of oil level in tank and top up if required</td><td></td></tr>
<tr><td colspan="7">*** End of BFF Inspection Items ***</td></tr>
</table>

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - _____ Station : _____

Date : _____ FML No : _____

ITEM CODE NO.	TASK	INITIAL			
Engine Area – Between flight (Turn-Around Inspection)		**FLIGHT 2**	**FLIGHT 3**	**FLIGHT 4**	**FLIGHT 5**
ALF001	Make sure that nothing is flowing through the main drain of the engine deck.				
ALF002	In the engine log book, record the number of C1 and C2 cycles. **Note: If the automatic counting is not available**				
ALF003	Inspection of oil level in tank and top up if required.				

Do a special check for the following optional equipment (ALF):

ITEM CODE NO.	TASK	INITIAL		ITEM CODE NO.	TASK	INITIAL
Emergency Floatation Gear						
ALF004	Fittings on bars: Condition, attachment, safety pins installed.			ALF006	Inflating pipe: Condition, attachment.	
ALF005	Protective cover: Condition, correct lacing.			ALF007	Cylinder assembly: Condition, attachment, correct pressure.	
NOTE Clean the equipment with fresh water after each flight in a sand-laden atmosphere.						
KANNAD INTEGRA AP-H Emergency Locator Transmitter						
ALF008	Make sure that the remote control "ON-ARMED-TEST/RESET" switch is set to "ARMED".			ALF009	Make sure that the ELT "ARM-ON-OFF" switch is set to "ARM".	

*** End of Between Flight and ALF Inspection Items ***

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
05/21/00 /000/000 /020	05-21	P-check VC **AMM 05-40-00, 6-7**	
05/21/00 /000/000 /030	05-21	P-check - Optional equipments Note: Maintenance operations can be carried out by an aircrew member. VC **AMM 05-40-00, 6-8**	
53/00/00 /000/000 /050	53-00	Structure Salt-laden atmosphere Rinsing and Drying. CLN **AMM 12-20-00, 3-3**	
*** End of 10FH // 7D Inspection Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
21/51/00 /000/000 /015	21-51	Freon air conditioning system – **Optional equipment** Visual inspection of the installation. GVI **AMM 21-51-02, 6-1**	
21/51/00 /000/000 /020	21-51	P2 emergency shut-off valve – **Optional equipment** POST MOD OP4353 Visual check. VC **AMM 21-51-02, 6-4**	
24/00/00 /000/000 /000	24-00	Antivibrator Visual check VC **AMM 18-30-00, 6-1**	
52/11/00 /000/000 /030	52-11	Crew door Functional test. Visual check. FT|DI **AMM 52-11-01, 6-1**	
52/12/00 /000/000 /010	52-12	Sliding door Functional test. Visual check. FT|DI **AMM 52-12-01, 6-1**	
53/00/00 /000/000 /025	53-00	Rear fuselage PRE MOD 074581 Visual check without removal. GVI **AMM 53-00-00, 6-2**	
53/21/00 /000/000 /000	53-21	Pick-up web of canopy frame Check. GVI **AMM 53-21-00, 6-1**	

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
56/11/00 /000/000 /010	56-11	Windows and windshield Bonding area of the transparent panels. Check. GVI **AMM 56-11-00, 6-1**	
62/11/00 /000/000 /100	62-11	Blade pin Tropical and damp atmosphere Salt-laden atmosphere Greasing. LUB **AMM 62-11-00, 3-3**	
62/21/00 /000/000 /230	62-21	Starflex star - Swivel bearing Visual check and play check. GVI **AMM 62-21-00, 6-2**	
62/21/00 /000/000 /235	62-21	Starflex star Checking the B/C/D/F areas. GVI **AMM 62-21-00, 6-1**	
62/30/00 /000/000 /030	62-30	Swashplates Greasing the bearing. LUB **AMM 62-32-00, 3-1**	
62/30/00 /000/000 /040	62-30	Scissors bushings and attachment bolts Sand-laden and/or dust-laden atmosphere Greasing. LUB **AMM 62-33-00, 3-1**	
63/11/00 /000/000 /065	63-11	Flexible coupling Visual check. VC **AMM 63-11-00, 6-18**	

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
63/11/00 /000/000 /070	63-11	Hydraulic pump - Drive-belt and bearing Check. VC **AMM 63-11-00, 6-2** **AMM 63-11-00, 6-15**		
63/11/00 /000/000 /300	63-11	Hydraulic pump – Bearing PRE MOD 079568 Greasing. LUB **AMM 63-11-00, 3-1**		
63/21/00 /000/000 /035	63-21	Spectrometric Oil Analysis Program (SOAP) Oil monitoring using SOAP is optional. SDI **AMM 60-00-00, 6-1**		
63/21/00 /000/000 /270	63-21	Epicyclic reduction gear – Electrical chip detector Check that the electrical system is operating correctly. FT **AMM 60-00-00, 6-2**	**RII:**	
63/21/00 /000/000 /280	63-21	MGB – Electrical chip detector Check that the electrical system is operating correctly. FT **AMM 60-00-00, 6-2**	**RII:**	
63/30/00 /000/000 /020	63-30	Laminated pads Visual check without removal. GVI **AMM 63-31-00, 6-5**		
63/30/00 /000/000 /030	63-30	Suspension cross member Visual check without removal. GVI **AMM 63-31-00, 6-6**		

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
64/10/00 /000/000 /005	64-10	Blade - Air duct Check - Clearance. GVI **AMM 64-21-00, 6-2**	
64/10/00 /000/000 /020	64-10	Tail rotor blade assy Visual check without removal. GVI **AMM 64-21-00, 6-21**	
64/21/00 /000/000 /060	64-21	Fairing assy Visual check. GVI **AMM 64-21-00, 6-4**	
64/21/00 /000/000 /070	64-21	Central plate Visual check without removal. GVI **AMM 64-21-00, 6-22**	
64/21/00 /000/000 /080	64-21	Control plate assy Visual check without removal. GVI **AMM 64-21-00, 6-23**	
64/21/00 /000/000 /090	64-21	Tail rotor hub assy Visual check without removal. GVI **AMM 64-21-00, 6-24**	
64/21/00 /000/000 /100	64-21	Outer bearing block Visual check without removal. GVI **AMM 64-21-00, 6-25**	

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
4/21/00/000/000/110	64-21	Inner bearing block Visual check without removal. GVI **AMM 64-21-00, 6-26**	
64/21/00/000/000/120	64-21	Torsion tie bar Visual check without removal. GVI **AMM 64-21-00, 6-10**	
64/21/00/000/000/130	64-21	Torsion tie bar-to-blade attach bolt Visual check without removal. GVI **AMM 64-21-00, 6-28**	
64/21/00/000/000/140	64-21	Torsion tie bar-to-rotor hub attach bolt Visual check without removal. GVI **AMM 64-21-00, 6-29**	
64/21/00/000/000/150	64-21	Upper chinese ring Visual check without removal. GVI **AMM 64-21-00, 6-30**	
64/21/00/000/000/160	64-21	Lower chinese ring Visual check without removal. GVI **AMM 64-21-00, 6-31**	
65/11/00/000/000/092	65-11	Bearing blocks no 1 and 2 to 5 PRE MOD 079809 Visual check. GVI **AMM 65-11-00, 6-17**	

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
65/11/00 /000/000 /212	65-11	Flexible coupling PRE MOD 079809 Visual check. GVI **AMM 65-11-00, 6-14**		
65/11/00 /000/000 /302	65-11	Blanking plate POST MOD 079061 & PRE MOD 079809 Visual check. GVI **AMM 65-11-00, 6-21**		
65/11/00 /000/000 /322	65-11	Rubber sleeve POST MOD 079059 & PRE MOD 079809 Visual check. GVI **AMM 65-11-00, 6-12**		
65/11/01 /000/000 /010	65-11	Tail rotor drive assembly POST MOD 079809 Visual check. GVI **AMM 65-11-01, 6-1**		
65/21/00 /000/000 /070	65-21	Spectrometric Oil Analysis Program (SOAP) Oil monitoring using SOAP is optional. SDI **AMM 60-00-00, 6-1**		
65/21/00 /000/000 /130	65-21	TGB – Electrical chip detector Check that the electrical system is operating correctly. FT **AMM 60-00-00, 6-2**	RII:	
76/12/00 /601/000 /005	76-12	Twist grip assembly Perform functional check. FT **AMM 76-12-04, 6-1**	RII:	

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
79/21/00 /000/000 /001	79-21	Hopper Check. GVI **AMM 79-21-00, 6-1**	
*** End of 150FH // 12M Inspection Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

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ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
53/31/01 /000/000 /020	53-31	Rear fuselage POST MOD 074581 Visual check. GVI **AMM 53-31-01, 6-1**		
53/31/01 /000/000 /027	53-31	Rear fuselage POST MOD 074581 Visual check. GVI **AMM 53-00-00, 6-2**		
62/30/00 /000/000 /005	62-30	Rotating swashplate - 4 contacts bearing Y51BB10843S2M74 (704A33651158) Inspection of free rotation. Operation to be performed from 5100 FH to 6600 FH of the OTL. DI **AMM 62-32-00, 6-1**	**RII:**	
71/61/00 /000/000 /015	71-61	Sand filter installation **– Optional equipment** Check presence indication of sand filter on VEMD. DI **AMM 71-61-10, 5-1**		
*** End of 150FH Inspection Items ***				

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

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Reg. Mark : PK - Date :

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ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
24/33/00 /000/000 /030	24-33	Battery 150CH-1 (7050A4243027) 151CH-1 (7050A4243040) Check. DI **AMM 24-33-00, 6-2**	
25/00/00 /000/000 /010	25-00	Belt and harness – Strap Check and functional test. Operation to be carried out each 12 M (margin 36 D) or after each installation of the seat or of the harness. GVI|FT **AMM 25-20-00, 5-1** **AMM 25-20-00, 6-1**	
25/10/40 /000/000 /000	25-10	APPAREO camera Functional test. FT **AMM 25-10-40, 5-2**	
25/66/20 /000/000 /010	25-66	Emergency locator transmitter KANNAD INTEGRA AP-H (ER). Installation check. GVI **AMM 25-66-20, 6-1**	
25/67/00 /000/000 /050	25-67	Emergency floatation gear **– Optional equipment** Functional test frangible disk. Check to perform during the first assembly and to initiate from this one. FT **AMM 25-67-00, 5-2**	
25/67/00 /000/000 /060	25-67	Emergency floatation gear **– Optional equipment** Weighing or pressure check of the cylinders. Check to perform during the first assembly and to initiate from this one. DI|WGH **CMM 25-69-11** **INSPECTION/CHECK** **§ 2A//2B//2C**	

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
26/22/00 /000/000 /025	26-22	Cabin fire extinguisher 12085-01 (S262A10T1001) Weighing. Interval starting from the date of manufacture. WGH **AMM 26-21-00, 6-2**	
34/00/00 /000/000 /020	34-00	Stand-by horizon - Stand-by battery **– Optional equipment** Check and charge the battery. FT **AMM 24-33-00, 5-2**	
*** End of 12M Inspection Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
21/51/00 /000/000 /021	21-51	P2 emergency shut-off valve – **Optional equipment** POST MOD OP4353 Functional test. FT **AMM 21-51-02, 6-5**	
21/51/00 /000/000 /050	21-51	Freon air conditioning system – **Optional equipment** Detailed inspection of the installation. DI|CLN|FT **AMM 21-51-02, 6-3**	
24/00/00 /000/000 /020	24-00	Electrical power supply system Check. GVI **AMM 24-00-00, 6-1**	
24/33/00 /000/000 /010	24-33	Battery temperature warning light Functional test. FT **AMM 24-33-00, 5-1**	
25/61/00 /000/000 /000	25-61	Wire strike protection system – **Optional equipment** Check. GVI **AMM 25-61-00, 6-1**	
26/11/00 /000/000 /020	26-11	Engine fire detector Condition and continuity check. FT **AMM 26-11-00, 6-2**	
28/00/00 /000/000 /005	28-00	Fuel system Check - Cleaning of the strainer. DI|CLN **AMM 28-00-01, 6-1**	

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
28/11/00 /000/000 /005	28-11	Fuel tank Adjustment strap tension. RTQ **AMM 28-11-01, 5-1**	
28/21/00 /000/000 /015	28-21	Fuel shut-off valve Functional test. FT **AMM 28-21-01, 5-2**	
29/11/00 /000/000 /010	29-11	Main hydraulic power system Functional test - Indicator - SERVO - LIMIT - HYDR. FT **AMM 29-11-00, 5-1**	
30/42/00 /000/000 /000	30-42	Check of the internal mechanism and the motor support. Check of axial and vertical play. Lubrication of the windshield wiper arm drive spindle. GVI|LUB **AMM 30-42-01, 3-1**	
31/00/00 /000/000 /000	31-00	Indicating and recording system Check. GVI **AMM 31-00-00, 6-1**	
32/11/00 /000/000 /000	32-11	Landing gear Check. DI **AMM 32-11-00, 6-1**	
33/00/00 /000/000 /010	33-00	External light Check of the lighting system. DI **AMM 33-00-00, 6-1**	

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
34/00/00 /000/000 /010	34-00	Navigation system Check. GVI **AMM 34-00-00, 6-1**	
52/11/00 /000/000 /040	52-11	Crew door Functional test - Jettison. FT **AMM 52-11-01, 5-1**	
52/11/00 /000/000 /050	52-11	Crew door Visual check without removal. GVI **AMM 52-11-01, 6-2**	
52/12/00 /000/000 /020	52-12	Sliding door Detailed inspection. DI **AMM 52-12-01, 6-2**	
52/31/00 /000/000 /020	52-31	Lateral cargo door Visual check without removal. Functional test - Indicating system. GVI|FT **AMM 52-31-01, 6-1**	
52/32/00 /000/000 /010	52-32	Rear cargo compartment door Visual check without removal. GVI **AMM 52-32-01, 6-1**	
53/00/00 /000/000 /005	53-00	Fuselage Check. DI **AMM 53-00-00, 6-1**	

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
53/10/00 /000/000 /050	53-10	Electrical connectors 130 & 131 DELTA Restoration. RS **MTC 20-80-20-444**		
53/41/00 /000/000 /010	53-41	Fenestron POST MOD 074581 Detailed inspection. DI **AMM 53-41-00, 6-3**	**RII:**	
55/11/00 /000/000 /000	55-11	Horizontal stabilizer Detailed check. DI **AMM 55-11-00, 6-1**		
56/11/00 /000/000 /020	56-11	Windshield, glazed panel, door window Check for cracks. GVI **AMM 56-11-00, 6-1**		
62/11/00 /000/000 /050	62-11	Main rotor blade Check. Check every 30 FH (margin 3 FH) for 120 FH, if a defect (inside the limits) is found: blade skin separation or blade leading edge stainless steel strip separation, cracks in the stainless steel leading edge. If no development is noted during this period, return to the normal inspection cycle. DI **AMM 62-11-00, 6-1**	**RII:**	

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
62/21/00 /000/000 /285	62-21	MRH Check. DI **AMM 62-21-00, 6-3** **AMM 62-21-00, 6-4** **AMM 62-21-00, 6-5** **AMM 62-21-00, 6-6** **AMM 62-21-00, 6-7** **AMM 62-21-00, 6-8** **AMM 62-21-00, 6-9** **AMM 62-21-00, 6-10** **AMM 62-21-00, 6-12** **AMM 62-21-00, 6-16** **AMM 62-21-00, 6-17** **AMM 62-21-00, 6-21**	**RII:**	
62/21/00 /000/000 /500	62-21	Droop stop ring Detailed check. DI **AMM 62-21-00, 6-13**	**RII:**	
62/30/00 /000/000 /060	62-30	Rotor mast - Upper visible section Detailed check. DI **AMM 62-31-00, 6-1**	**RII:**	
62/30/00 /000/000 /070	62-30	Rotor mast upper housing Check and draining. DRN **AMM 62-31-00, 6-2**		
62/30/00 /000/000 /080	62-30	Swashplates Check and grease lubrication. DI|LUB **AMM 62-32-00, 3-1** **AMM 62-32-00, 6-1**	**RII:**	
62/30/00 /000/000 /090	62-30	Diapason scissors link Check. DI **AMM 62-33-00, 6-2**	**RII:**	

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
62/30/00 /000/000 /100	62-30	Scissors link Detailed check and end play of hinges. DI **AMM 62-33-00, 6-3**	**RII:**	
62/30/00 /000/000 /110	62-30	Scissors bushings attachment bolts Greasing. LUB **AMM 62-33-00, 3-1**		
62/30/00 /000/000 /190	62-30	Seal bead Check. DI **AMM 62-31-00, 6-4**		
63/11/00 /000/000 /009	63-11	Hydraulic pump - Bearing POST MOD 079568 Greasing. LUB **AMM 63-11-00, 3-1**		
63/11/00 /000/000 /014	63-11	Hydraulic pump - Bearing Check. DI **AMM 63-11-00, 6-19**	**RII:**	
63/11/00 /000/000 /055	63-11	Hydraulic pump - Belt Tension check. RIG **AMM 63-11-00, 6-20**	**RII:**	
63/21/00 /000/000 /105	63-21	MGB - Mineral oil Draining/Filling. DRN **AMM 12-10-00, 3-1**		

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
63/30/00 /000/000 /040	63-30	MGB suspension bar Visual check without removal. GVI **AMM 63-32-00, 6-3**		
63/30/00 /000/000 /050	63-30	MGB bar attachment bolt Visual check without removal. GVI **AMM 63-32-00, 6-4**		
63/51/00 /000/000 /000	63-51	Rotor brake micro switch Check. FT **AMM 63-51-00, 5-1**		
64/21/00 /000/000 /267	64-21	Torsion tie bar Tropical and damp atmosphere Detailed check with removal. DI **AMM 64-21-00, 6-27**	**RII:**	
65/11/00 /000/000 /042	65-11	Flexible coupling PRE MOD 079809 Tightening torque check. TCK **AMM 65-11-00, 6-7**	**RII:**	
65/11/00 /000/000 /057	65-11	Bearing block n°1 PRE MOD 079809 Tightening torque check. TCK **AMM 65-11-00, 6-18**	**RII:**	
65/11/00 /000/000 /102	65-11	Front shaft section PRE MOD 079809 Visual check. GVI **AMM 65-11-00, 6-11**		

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
65/11/00 /000/000 /112	65-11	Central shaft section PRE MOD 079809 Visual check. GVI **AMM 65-11-00, 6-19**	
65/11/00 /000/000 /232	65-11	Rear shaft section PRE MOD 079809 Visual check. GVI **AMM 65-11-00, 6-13**	
65/11/00 /000/000 /242	65-11	Sliding flange PRE MOD 079809 Visual check. GVI **AMM 65-11-00, 6-16**	
65/11/00 /000/000 /312	65-11	Blanking plate POST MOD 079061 & PRE MOD 079809 Detailed check. DI **AMM 65-11-00, 6-22**	
65/11/01 /000/000 /100	65-11	Sliding flange - splines POST MOD 079809 Greasing. LUB **AMM 65-11-01, 3-2**	
65/11/01 /000/000 /110	65-11	Bearing block assembly - bearing POST MOD 079809 Greasing. LUB **AMM 65-11-01, 3-1**	
65/11/01 /000/000 /120	65-11	Bearing block assembly POST MOD 079809 Tightening torque check. TCK **AMM 65-11-01, 6-10**	**RII:**

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
65/11/01 /000/000 /130	65-11	Flexible coupling POST MOD 079809 Tightening torque check. TCK **AMM 65-11-01, 6-9**	**RII:**	
65/11/01 /000/000 /140	65-11	Damper POST MOD 079809 Visual check. GVI **AMM 65-11-01, 6-2**		
65/21/00 /000/000 /050	65-21	TGB Check. DI **AMM 65-21-00, 6-1**		
65/21/00 /000/000 /090	65-21	TGB Draining/Filling. DRN **AMM 12-10-00, 3-1**		
67/11/00 /000/000 /010	67-11	Main rotor control Check. GVI **AMM 67-10-00, 6-1**		
67/21/00 /000/000 /010	67-21	Tail rotor control Check. GVI **AMM 67-21-00, 6-1**		
67/30/00 /000/000 /020	67-30	Main servocontrol Tightening torque check of servocontrol attaching bolts. TCK **AMM 67-31-00, 6-3**	**RII:**	

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
67/30/00 /000/000 /040	67-30	Main servocontrol Check for correct operation of load detector of the RH servocontrol. FT **AMM 67-31-00, 5-1**	**RII:**	
71/00/00 /000/000 /000	71-00	Power plant installation Check for correct condition and attachment of the items. GVI **AMM 71-00-00, 6-1**		
71/61/00 /000/000 /000	71-61	Sand filter installation **– Optional equipment** Check. DI **AMM 71-61-10, 6-1**		
76/12/00 /000/000 /025	76-12	Twist grip assembly Functional test. FT **AMM 76-12-04, 5-1**	**RII:**	
76/21/00 /000/000 /000	76-21	Emergency shut down - Fuel shut-off ball type control Lever on cabin ceiling. Test without removal. FT **AMM 76-21-00, 5-1**		
77/00/00 /000/000 /000	77-00	Engine emergency governing system "EBCAU" Check. FT|GR **AMM 77-00-00, 5-4** **FLM Section 8.3 73**	**RII:**	
79/00/00 /000/000 /010	79-00	Lubrication system Check. GVI **AMM 79-00-00, 6-1**		

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
88/00/00 /000/000 /010	88-00	Electrical harnesses Check. GVI **AMM 88-00-00, 6-1**	
*** End of 600FH // 24M Inspection Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
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RETURN TO SERVICE

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ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
62/30/00 /000/000 /012	62-30	Rotating swashplate - 4 contacts bearing Y51BB10843S2M74 (704A33651158) Inspection of free rotation. Operation to be performed from 0 FH to 5100 FH of the OTL. DI **AMM 62-32-00, 6-1**	**RII:**	
62/30/00 /000/000 /230	62-30	Scissors drive coupling - Sealing bead Check. DI **AMM 62-31-00, 6-6**	**RII:**	
63/21/00 /000/000 /290	63-21	MGB - Chips detector Check of the magnetization of the electrical or non-electrical magnetic plug. DI **AMM 60-00-00, 6-4**		
63/21/00 /000/000 /295	63-21	Epicyclic reduction gear – Electrical chip detector Check of the magnetization of the electrical or non-electrical magnetic plug. DI **AMM 60-00-00, 6-4**		
63/51/00 /000/000 /050	63-51	Rotor brake Inspection. GVI **AMM 63-51-00, 6-1**		
64/21/00 /000/000 /370	64-21	Tail rotor blade - Bush Replacement. DS **AMM 64-21-00, 8-2**	**RII:**	

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
65/21/00 /000/000 /150	65-21	TGB - Chips detector Check of the magnetization of the electrical or non-electrical magnetic plug. DI **AMM 60-00-00, 6-4**	
80/00/02 /000/000 /120	80-00	Starter generator 515-030 (704A46101018) 524-031 (704A46101011) THALES. Checking generator, brushes and splines. DI **AMM 80-00-02, 6-1** **AMM 80-00-02, 6-2** **AMM 80-00-02, 6-3**	
*** End of 600FH Inspection Items ***			

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ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
24/33/00 /000/000 /040	62-30	Battery 150CH-1 (7050A4243027) 151CH-1 (7050A4243040) Overhaul. DI **AMM 24-33-00, 6-3**	
24/33/00 /000/000 /050	62-30	Battery Battery compartment cleaning. CLN **AMM 24-33-00, 2-1**	
*** End of 24M Inspection Items ***			

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ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
29/11/00 /000/000 /050	29-11	Hydraulic pump - Poly "V" belt - Key Condition check. VC **AMM 29-12-01, 6-1**		
63/11/00 /000/000 /030	63-11	Engine-to-MGB coupling Detailed check. DI **AMM 63-11-00, 6-4**	**RII:**	
63/11/00 /000/000 /040	63-11	Engine-to-MGB coupling housing Detailed check. DI **AMM 63-11-00, 6-5**		
63/11/00 /000/000 /060	63-11	Universal joint ring Detailed check. DI **AMM 63-11-00, 6-6**		
63/11/00 /000/000 /066	63-11	Flector Check. DI|TCK **AMM 63-11-00, 6-1** **AMM 63-11-00, 6-16**	**RII:**	
63/11/00 /000/000 /080	63-11	Universal joint ring pin Detailed check. DI **AMM 63-11-00, 6-7**		

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
63/11/00 /000/000 /100	63-11	Drive shaft Detailed check. DI **AMM 63-11-00, 6-8**		
63/11/00 /000/000 /120	63-11	Engine coupling flange Detailed check. DI **AMM 63-11-00, 6-9**	**RII:**	
63/11/00 /000/000 /160	63-11	Hydraulic pump flexible flange Detailed check. DI **AMM 63-11-00, 6-12**		
63/11/00 /000/000 /180	63-11	Coupling sleeve Detailed check. DI **AMM 63-11-00, 6-13**	**RII:**	
63/21/00 /000/000 /130	63-21	MGB - Synthetic oil Oil change. DRN **AMM 12-10-00, 3-1**		
64/21/00 /000/000 /190	64-21	Tail rotor blade assy Detailed check. DI **AMM 64-21-00, 6-3**	**RII:**	
64/21/00 /000/000 /200	64-21	Fairing assy Detailed check. DI **AMM 64-21-00, 6-4**		

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
64/21/00 /000/000 /210	64-21	Central plate Detailed check. DI **AMM 64-21-00, 6-5**		
64/21/00 /000/000 /220	64-21	Control plate assy Detailed check. DI **AMM 64-21-00, 6-6**		
64/21/00 /000/000 /230	64-21	Tail rotor hub assy Detailed check. DI **AMM 64-21-00, 6-7**	**RII:**	
64/21/00 /000/000 /240	64-21	Outer bearing block Detailed check. DI **AMM 64-21-00, 6-8**	**RII:**	
64/21/00 /000/000 /250	64-21	Inner bearing block Detailed check. DI **AMM 64-21-00, 6-9**	**RII:**	
64/21/00 /000/000 /270	64-21	Torsion tie bar-to-blade attach bolt Detailed check. DI **AMM 64-21-00, 6-11**	**RII:**	
64/21/00 /000/000 /280	64-21	Torsion tie bar-to-rotor hub attach bolt Detailed check. DI **AMM 64-21-00, 6-12**	**RII:**	

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
64/21/00 /000/000 /290	64-21	Upper chinese ring Detailed check. DI **AMM 64-21-00, 6-13**	**RII:**	
64/21/00 /000/000 /300	64-21	Lower chinese ring Detailed check. DI **AMM 64-21-00, 6-14**	**RII:**	
64/21/00 /000/000 /310	64-21	Sliding bush Detailed check. DI **AMM 64-21-00, 6-15**	**RII:**	
64/21/00 /000/000 /320	64-21	Splined flange Detailed check. DI **AMM 64-21-00, 6-16**	**RII:**	
64/21/00 /000/000 /330	64-21	Antibulging sector Detailed check. DI **AMM 64-21-00, 6-17**	**RII:**	
64/21/00 /000/000 /340	64-21	Thrust nut Detailed check. DI **AMM 64-21-00, 6-18**	**RII:**	
64/21/00 /000/000 /350	64-21	Lockwasher Detailed check. DI **AMM 64-21-00, 6-19**		

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
64/21/00 /000/000 /360	64-21	Thrust screw Detailed check. DI **AMM 64-21-00, 6-20**	**RII:**	
65/11/00 /000/000 /252	65-11	Front shaft section PRE MOD 079809 Detailed check. DI **AMM 65-11-00, 6-1**	**RII:**	
65/11/00 /000/000 /262	65-11	Central shaft section PRE MOD 079809 Detailed check. DI **AMM 65-11-00, 6-2**	**RII:**	
65/11/00 /000/000 /272	65-11	Rear shaft section PRE MOD 079809 Detailed check. DI **AMM 65-11-00, 6-3**	**RII:**	
65/11/00 /000/000 /282	65-11	Sliding flange PRE MOD 079809 Detailed check. DI **AMM 65-11-00, 6-6**	**RII:**	
65/11/00 /000/000 /292	65-11	Flexible couplings PRE MOD 079809 Detailed check. DI **AMM 65-11-00, 6-4**	**RII:**	
65/11/01 /000/000 /200	65-11	Front shaft assy POST MOD 079809 Detailed inspection. DI **AMM 65-11-01, 6-3**	**RII:**	

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
65/11/01 /000/000 /210	65-11	Rear shaft assy POST MOD 079809 Detailed inspection. DI **AMM 65-11-01, 6-4**	**RII:**	
65/11/01 /000/000 /220	65-11	Sliding flange POST MOD 079809 Detailed check. DI **AMM 65-11-01, 6-5**	**RII:**	
65/11/01 /000/000 /230	65-11	Bearing block assembly POST MOD 079809 Detailed inspection. DI **AMM 65-11-01, 6-6**	**RII:**	
65/11/01 /000/000 /240	65-11	Flexible couplings POST MOD 079809 Detailed inspection. DI **AMM 65-11-01, 6-8**	**RII:**	
65/11/01 /000/000 /250	65-11	Damper POST MOD 079809 Detailed inspection. DI **AMM 65-11-01, 6-7**		
65/11/01 /000/000 /260	65-11	Balancing system POST MOD 079809 Detailed inspection. FT **AMM 65-11-01, 5-1**		
71/31/00 /000/000 /000	73-31	Rear Firewall POST MOD 074581 Detailed inspection. FT **AMM 71-31-00, 6-1**		
*** End of 1200FH // 48M Inspection Items ***				

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

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ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
25/11/00 /000/000 /000	25-11	Energy absorption front seat SICMA 159. Check of the structure. DI **AMM 25-21-00, 6-6**		
25/11/00 /000/000 /050	25-11	Energy absorption front seat SICMA 198. Check of the structure. DI **AMM 25-21-01, 6-2**		
25/21/00 /000/000 /080	25-21	Rear seat SICMA 284. Check of the structure. DI **CMM 25-12-32**		
76/12/00 /000/000 /005	76-12	Twist grip - Load spring Replacement of spring. DS **AMM 76-12-04, 4-2**	**RII:**	
77/00/00 /501/000 /010	77-00	FADEC unit POST MOD 074830 & POST MOD 074831 Test of the overspeed protection system. Engine ARRIEL 2D. FT **AMM 73-20-00, 6-2**	**RII:**	
*** End of 1200FH Inspection Items ***				

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ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
60/00/00 /000/000 /005	60-00	MRH - Main Rotor Mast - TRH - Tail rotor drive assembly - Engine/MGB coupling Tropical and damp atmosphere Salt-laden atmosphere Operation to be carried out after having logged 96 M (margin 180 D) after initial setting to service. DI **AMM 05-60-00, 6-1**	**RII:**	
60/00/05 /000/000 /030	60-00	MGB assembly - TGB assembly Salt-laden atmosphere Tropical and damp atmosphere Check. Operation to be carried out after having logged 96 M (margin 180 D) after initial setting to service or since the last Overhaul. DI **AMM 05-60-00, 6-1**	**RII:**	
*** End of 48M Inspection Items ***				

PERSONNEL PRTICIPATING IN THIS INSPECTION			
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ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
N/A	N/A	Record the corrective maintenance operations performed in the engine log book, if any.	
N/A	N/A	Make sure that the automatic cycle counting is correct. **EMM 05-10-02-200-801**	
N/A	N/A	In the engine log book, record the total number of C1 and C2 cycles consumed as counted by the EECU. **EMM 05-10-02-200-801**	
N/A	N/A	Visually examine the engine and the engine floor for leakage.	
N/A	N/A	Visually examine the engine attachments and fire protections for signs of impact or deterioration. **Refer to Helicopter Manual** **Note: Fire protections of shut off valve and HMU**	
N/A	N/A	Record the values from the creep damage counter in the engine logbook. **EMM 05-15-00-200-801**	
N/A	N/A	Record in the engine logbook the accumulation of flying hours. **EMM 05-10-02-200-801**	
N/A	N/A	Check that the HP gas generator rotates freely (no abnormal noises) and visually check that the engine is in good condition. Note: **Manually or during a dry crank cycle**	
N/A	N/A	Manually check that the power turbine rotates freely (no abnormal noises). **EMM 05-10-02-200-801**	

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
N/A	N/A	Inspection of oil level in tank and top up if required. **Refer to Helicopter Manual** Note: **Within 15 minutes following engine shut-down.**	
N/A	N/A	Make sure that there are no foreign objects: examine near the air intakes and the exhaust zone. Install the blanks.	
*** End of Arriel 2D 15FH // 7D Inspection Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
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RETURN TO SERVICE

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ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
N/A	N/A	Engine health inspection. **Refer to Flight Manual** **NOTE:** Check also P0T0 (Engine/Aircraft consistency). Power check result: N1 : _____% N2 : _____Rpm TOT : _____ ^{0}C Altitude : _____Feet TRQ : _____% OAT : _____ ^{0}C TOT Margin : _____% N1 Margin : _____%	
*** End of Arriel 2D 25FH Inspection Items ***			

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TSN / CSN : _____ WO No. : _____

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
N/A	N/A	Module 02 - Axial compressor - Erosion check. **EMM: 72-00-32-200-801**	
*** End of Arriel 2D 300FH Inspection Items ***			

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Signature : _____ Place/Date : _____

Reg. Mark : PK - _____ Date : _____

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TSN / CSN : _____ WO No. : _____

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
N/A	N/A	Oil system - Draining. **EMM: 79-00-00-610-801** **Operating condition: In a sandy air or with the use of 3 cSt oil.**	
*** End of Arriel 2D 800FH // 12M Inspection Items ***			

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Reg. Mark : PK - _____ Date : _____

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TSN / CSN : _____ WO No. : _____

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
N/A	N/A	Oil system - Draining. **EMM: 79-00-00-610-801** **Operating condition: With the use of 5 cSt oil.**	
*** End of Arriel 2D 800FH // 24M Inspection Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
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ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
N/A	N/A	Boroscope inspection of combustion chamber including nozzle guide vane. **EMM: 72-00-43-200-802**	**RII:**	
N/A	N/A	Functional check in the auxiliary mode (EBCAU test) **Refer to Flight Manual** **Note: Ground run**		
N/A	N/A	Module 03 - Turbine-casing drain valve assembly - Test. **EMM: 71-71-00-700-801** **Note: Ground run**		
N/A	N/A	Module 03 - Checking of fluids discharged during injection wheel bleeding and checking of fluids discharged in tank fuel return pipe. **EMM: 73-14-00-700-802** **Note: Ground run**		
N/A	N/A	Module 02 - Axial compressor - Erosion check. **EMM: 72-00-32-200-801**		
N/A	N/A	Mechanical magnetic plug - Magnetism test. **EMM: 79-36-00-700-801**		
N/A	N/A	Test of the magnetism of the electrical magnetic plug. **EMM: 79-38-00-700-801**		
N/A	N/A	Test of the electrical magnetic plug. **EMM: 79-38-00-750-801**		
N/A	N/A	Module 03 – Permeability check of the injection wheel. **EMM: 72-00-43-200-801**		
N/A	N/A	Replacement of the fuel filtering element. **EMM: 73-23-14-900-802**		
N/A	N/A	Bleed valve filter - Check and inspection. **EMM: 75-31-00-200-802**		

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
N/A	N/A	Bleed valve filtering kit - Filter element replacement. **EMM: 75-61-00-950-801** **Note: POST TU 94**	
N/A	N/A	Check of mechanical magnetic plug. **EMM: 79-36-00-900-801**	
N/A	N/A	Replacement of the oil filtering element. **EMM: 79-21-00-900-801**	
N/A	N/A	Module 03 - Preformed packing of rear bearing ducts - Replacement. **EMM: 72-43-00-900-806** **Note: PRE TU 181**	
N/A	N/A	Module 03 - Inspection of the ignition system. **EMM: 72-43-00-200-806** **Note: Engine running**	
N/A	N/A	Module 03 - Inspection of the igniter plugs. **EMM: 72-43-00-200-803** **Note: Engine running**	
N/A	N/A	Exhaust pipe – Check and inspection. **EMM: 72-70-00-200-801**	
N/A	N/A	Module 01 - Engine front support - Check and inspection. **EMM: 72-61-00-200-805**	
N/A	N/A	Inspection of pyrometric harness. **EMM: 77-21-00-210-801**	
N/A	N/A	Boroscope inspection of HP turbine blades and HP turbine nozzle guide vane. **EMM: 72-00-43-200-803** **RII:**	
*** End of Arriel 2D 800FH Inspection Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
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MSN : _____ Station : _____

TSN / CSN : _____ WO No. : _____

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
N/A	N/A	Remove and visually inspect power turbine pressurization pipe with magnifying glass x6: If pipe is cracked: replace it, And If presence of through-going crack: discard power turbine wheel **EMM: 75-29-00-900-805** **Note: Every 2,000 Hrs from 4,000 Hrs**	
*** End of Arriel 2D 2000FH Inspection Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

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Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
N/A	N/A	Inspect constant ΔP valve (including membrane replacement and functional test of auxiliary valve). **EMM: 73-23-00-900-801** **Note: Remove HMU and send it back to a Turbomeca approved repair center.**	
N/A	N/A	Check the start injector purge valve of Adjusted Valve Assembly. **EMM: 73-14-00-700-801**	
*** End of Arriel 2D 4000FH Inspection Items ***			

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ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
N/A	N/A	Remove and send Module 03 back to Safran Helicopter Engines approved repair center or apply Maintenance Technical Instruction to send back HP turbine wheel only for visual inspection of HP turbine wheel fir-tree roots for clogging. If one of the fir-tree root is completely clogged, discard HP turbine disc. **EMM: 72-00-43-900-801**	**RII:**	
N/A	N/A	Remove engine or Module 03 and send it back to Safran Helicopter Engines approved repair center for checking that the HP turbine containment shield is free from crack using dye penetrant inspection. If presence of crack, discard the HP turbine containment shield. **EMM: 72-00-43-900-801**	**RII:**	
N/A	N/A	Remove engine or module 03 and send it back to Safran Helicopter Engines approved repair center for checking that the power turbine containment shield is free from crack using dye penetrant inspection. If presence of crack, discard the Power Turbine containment shield. **EMM: 72-00-43-900-801**	**RII:**	
N/A	N/A	Functional test of oil pressure transmitter. **EMM: 79-32-00-750-801**		

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
N/A	N/A	Functional test of oil pressure and temperature transmitter. **EMM: 79-32-00-750-801**	
*** End of Arriel 2D 6000FH Inspection Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

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Name : _____ Stamp : _____

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Reg. Mark : PK - Date :

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TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
		Remove and send the Module 03 to a certified Maintenance Center for calendar inspection. **EMM: 71-02-16-280-801**	**RII:**	
*** End of Arriel 2D 15 Years Inspection Items ***				

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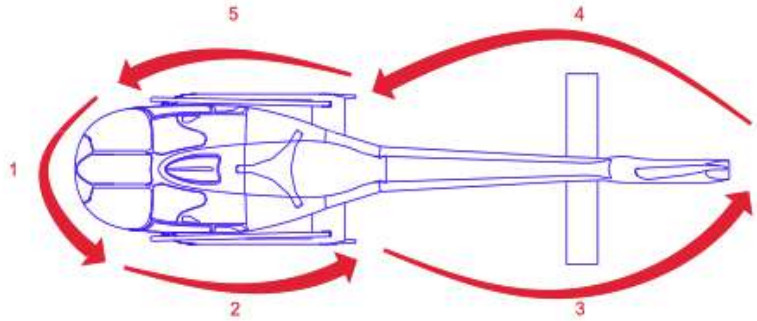
Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Station :

Date : FML No :

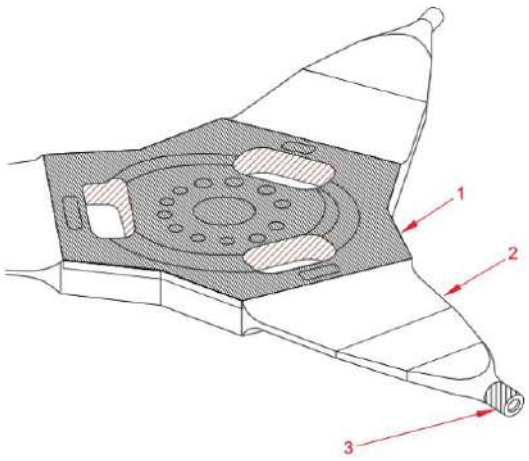


ITEM CODE NO.	TASK	INITIAL		ITEM CODE NO.	TASK	INITIAL
STATION 1						
PC001	Door jambs, canopy arches. Condition, no cracks.			PC003	Sideslip indicator. Condition.	
PC002	Pitot tube. Condition, cover removed or fitted as necessary.			PC004	MGB - Engine oil cooler air inlet. Condition, no obstruction or foreign bodies, blanking cover removed or fitted as necessary.	
STATION 2						
PC005	Front door jettison system Condition, no cracks (especially at the link).			PC019	Rear cargo door Closed and locked.	
PC006	Left cabin access doors Condition, attachment, locking, no abnormal play.			PC020	LH side MGB and engine cowlings Opening, condition of locking systems, no abnormal play.	
PC007	Landing gear Condition of crosstubes, skids, resistant plates, footstep attachment.			PC021	Upper cowlings Locked.	
PC008	Static pressure ports Condition, blanks removed or fitted as necessary.			PC022	NACA air inlet No obstructions (clean if necessary).	
PC009	OAT probes Condition, attachment.			PC023	MGB Condition, oil level, no leaks.	
PC0010	Antennas under bottom structure Condition.			PC024	Transmission deck Cleanliness.	
PC011	Landing and taxiing lights Condition.			PC025	MGB support bars Condition, attachment.	

ITEM CODE NO.	TASK	INITIAL		ITEM CODE NO.	TASK	INITIAL
PC012	Scoop Condition, cleanliness.			PC026	Hydraulic system (LH side) Condition, attachment points, pipes, filter clogging indicator retracted, no leaks.	
PC013	Lower cowlings Condition, secured.			PC027	Servos Attachment, no leaks or cracks.	
PC014	Left cargo door Opening, condition, attachment points, no abnormal play.			PC028	Gimbal ring assembly Fitting, safety pins in place and locked.	
PC015	Left cargo door Closed and locked.			PC029	Electrical harnesses Condition, attachment.	
PC016	Rear cargo door Opening, condition, attachment points, no abnormal play.			PC030	Fuel shut-off valve Condition, attachment, no interference.	
PC017	Rear cargo bay FADEC secured and harness condition.			PC031	MGB cowling (LH side) Closed and locked.	
PC018	KANNAD INTEGRA AP-H Emergency Locator Transmitter Condition, security, check "ARM" or "OFF" as necessary.			**NOTE** If the aircraft is grounded for a long period (more than 2 months), set the switch on the ELT to the "OFF" position.		
STATION 2 – ENGINE AND ENGINE BAY						
PC032	Engine air intake Attachment: condition, seal condition.			PC038	Fuel filter Clogging indicator retracted.	
PC033	Firewall Condition, no cracks.			PC039	Oil system No leaks.	
PC034	Engine and accessories General condition, cleanliness, leak-tightness, attachment, pipes, electrical harness.			PC040	Engine mounts Condition, attachment.	
PC035	Engine transmission deck Condition, cleanliness, no leaks.			PC041	Engine deck drain holes No obstructions or debris.	
PC036	Engine casing Attachment flange condition.			PC042	Exhaust pipe Condition, blanking cover removed or fitted, as necessary.	
PC037	Oil filter Clogging indicator retracted.					
STATION 3						
PC043	Tail boom Condition, condition of antennas.			PC048	TRH fairing Condition, damage or cracks, no rotations (paint marks).	
PC044	Drive shaft fairings Attachment, condition of heat shield.			PC049	Tail rotor duct Condition, check distance between blade tips and duct.	

ITEM CODE NO.	TASK	INITIAL		ITEM CODE NO.	TASK	INITIAL
PC045	Tail boom door Opening, attachment, bay condition, closed.			PC050	Tail rotor blades, visible part Condition, no impact, damage, scratches, erosion. Check play.	
PC046	Horizontal stabilizer, fin Condition, attachment points, condition of external lights and fin antennas.			PC051	Freewheel Make the freewheel turn by turning the tail rotor. The free turbine must be driven when the tail rotor is turned anti-clockwise. In the clockwise direction, the freeturbine does not drive.	
PC047	Keel, tail guard Condition, attachment.					
STATION 4						
PC052	TGB Condition, attachment, oil level.			PC055	Horizontal stabilizer, fin Condition, attachment points, external light condition.	
PC053	Guide vanes, duct Condition.			PC056	Tail boom Condition, condition of antennas.	
PC054	Tail rotor pitch change control rod Condition, attachment, no radial play in end- fitting.			PC057	Drive shaft fairings Attachment, condition of heat shield.	
STATION 5						
PF058	NACA air inlet No obstructions (clean if necessary).					
STATION 5 – ENGINE AND ENGINE BAY						
PC059	Engine air intake Attachment: condition, condition of seals. Blanking cover removed or fitted, as necessary.			PC065	Engine magnetic plugs - No chips on aft or forward plugs (without electrical indication).	
PC060	Firewall Condition, no cracks.			PC066	Engine mounts Condition, attachment.	
PC061	Engine and accessories General condition, cleanliness, leak-tightness, attachment, pipes, electrical harnesses.			PC067	Engine deck drain holes No obstructions or foreign bodies.	
PC062	Engine transmission deck Condition, cleanliness, no leaks.			PC068	Engine cowling Closed and locked.	
PC063	Engine casing Attachment flange condition.			PC069	Nozzle Blanking cover installed or removed as necessary.	
PC064	Oil system No leaks.					
STATION 5 – MAIN ROTOR SHAFT						

ITEM CODE NO.	TASK	INITIAL		ITEM CODE NO.	TASK	INITIAL
PC070	Swashplate bearing No grease runs, no change in paint color or paint flaking.			PC078	Visible part of the Starflex Star (1) arm (2): • no delamination, condition of the paint. If in doubt or damage is found (AMM 62-21-00,6-1). • Bushings (3) at the end of the arms of the STARFLEX (1) No gap between adhesive bead and bushing. If in doubt or damage is found (AMM 62-21-00,6-1).	
PC071	Scissors, swashplates, rod swivel bearings Condition, attachment, abnormal play (manual check).			PC079	Spherical thrust bearings No elastomer damage, delamination, splits, blisters, extrusions or cracks (other than minor and unchanging surface irregularities), elastomer protuberance at the laminated area/small reinforcing structure interface.	
PC072	Swashplate/pitch change rod end-fitting – interfaces No impact mark or paint flaking on swashplate attachment yokes.			PC080	Frequency adapters (PRE MOD 076232) If in doubt or damage is found (AMM 62-21-00,6-4). No elastomer damage, delamination, splits, blisters, extrusions or cracks (other than minor and unchanging surface irregularities). If in doubt or damage is found (AMM 62-21-00,6-7).	
PC073	Pitch change rods Condition, no radial play in end-fittings.			PC081	Ventilated frequency adapters (POST MOD 076232) No elastomer damage, delamination, splits, blisters, extrusions or cracks (other than minor and unchanging surface irregularities). If in doubt or damage is found (AMM 62-21-00,6-7). Check that ventilation holes are not obstructed (on both sides). On adapters fitted with drilled bush 365A31-1018-21 or 22, ensure lockwire is in place in holes on trailing edge side.	

ITEM CODE NO.	TASK	INITIAL		ITEM CODE NO.	TASK	INITIAL
PC074	Rotor shaft, all visible parts particularly under the – hub Paint condition, no cracks, scratches, blistering, corrosion nor tool marks.			PC082	Self-lubricating ball joints No debris or play. -	
PC075	1/4 turn non-electrical magnetic plug of flared housing. Remove 1/4 turn plug, if there are metal particles, apply (AMM 05-50-00,6-1), if there are no metal particles, do a second check, if there are metal particles, apply (AMM 05-50-00,6-1), if there are no metal particles, install the 1/4 turn plug.			PC083	Anti-vibrator Attachment.	
						
PC076	MAIN ROTOR HEAD Attachment, general condition.			PC084	MAIN ROTOR BLADES Attachment, general condition of the skin (lower surface, upper surface and trailing edge), trim tabs and polyurethane protective strips. Visually check for no delamination, scratches, cracks, impacts or distortions. No erosion holes on leading edge, no gaps or impacts. Condition of the electrical bonding braid. If in doubt or if a defect is found (AMM 62-11-00,6-1).	
PC077	STARFLEX Star (1): no delamination (splinters).			PC085	Right cargo door Opening, condition, attachment points, no abnormal play.	

ITEM CODE NO.	TASK	INITIAL		ITEM CODE NO.	TASK	INITIAL
STATION 5 – MAINTENANCE AREA						
PC086	Fuses All fuses set.			PC096	Engine oil tank Oil level, pipes condition, no leaks.	
PC087	Right cargo door Closed and locked.			PC097	Electrical harnesses Condition, attachment.	
PC088	GPU receptacle, access flap Closed or GPU connected, as applicable.			PC098	Gimbal ring assembly Fitting, safety pins in place and locked.	
PC089	RH MGB cowling Opening, condition of locking systems, no abnormal play.			PC099	RH side MGB cowling Closed and locked.	
PC090	Transmission deck Cleanliness.			PC100	Landing gear Condition of crosstubes, skids, wear resistant plates, footstep attachment.	
PC091	MGB support bars Condition, attachment, swivelling.			PC101	All lower central fairings Closed and locked.	
PC092	Oil cooler, fan and pipes Condition, no leaks, fan attachment, fan blade condition.			PC102	RH cabin access doors Condition, attachment, locking, no abnormal play.	
PC093	Servos Attachment, no leaks or cracks.			PC103	Front door jettison system Condition, no cracks (especially at the link).	
PC094	Hydraulic system (RH side) Attachment, pipes condition, no leaks, filter clogging indicator retracted.			PC104	Subdoor Condition, attachment.	
PC095	Hydraulic system reservoirs Levels, no leaks.					
STATION 5 – CABIN INTERIOR						
PC105	Cabin General cleanliness.			PC110	Fire extinguisher Attached - Checked.	
PC106	Seats Condition, attachment points.			PC111	Fuses All set.	
PC107	Belts/harnesses General condition, no wear, visible damage to the strap and reel in good working order.			PC112	Battery switch ON, check battery voltage.	

ITEM CODE NO.	TASK	INITIAL		ITEM CODE NO.	TASK	INITIAL
PC108	Belt attachments General condition, wear, loosening and locking.			PC113	VEMD Check flights of the day reports: (MAINT mode, FLIGHT REPORT page) VEMD flight times, Ng and Nf cycles: check written in white characters and above 0, Check advisory messages FAILURE or OVERLIMIT DETECTED, Record flights of the day data in aircraft and engine log-books.	
PC109	Door jettison system Checked - Plastic guard in place.			PC114	Battery switch OFF.	
OPTIONAL EQUIPMENT – ENGINE SAND FILTER						
PC1115	With the engine cowling closed, check the following: - the condition of the filter support cowling, - the external condition of the filter, - the condition and cleanliness of the separator tubes, - the condition of the ejector nozzles.			PC116	Open the engine cowling and check the following: - the condition of the particle separator unit, - the condition and cleanliness of the separator tubes, - the condition of the elbow union, - the condition of the pipes, - the condition of the air manifold, - the tightness of the air intake, - the internal cleanliness of the air manifold, - the condition and attachment of the electro-valve with hoses and P2 supply union.	
				PC117	Close the engine cowling.	
OPTIONAL EQUIPMENT – AIR CONDITIONING						
PC118	Open the MGB left cowling.			PC120	If a leak is detected at the MGB intake: replace the MGB intake seal (AMM 63-21-00,8-1).	
PC119	Check the condition of the air conditioning compressor drive belt (AMM 21-51-02,6-2) if an oil leak is detected at the MGB.			PC121	Close the MGB left cowling.	
OPTIONAL EQUIPMENT – INSTALLATION OF WINDSHIELD WIPERS						

ITEM CODE NO.	TASK	INITIAL		ITEM CODE NO.	TASK	INITIAL
PC122	Check the condition of the installation (no corrosion, impacts..).			PC124	Check the condition of the wiper and in particular, the good condition of the scraper.	
PC123	Check the condition of attachments.					
*** End of P Check Items ***						

RETURN TO SERVICE

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------ Reserved -------

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NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
1	8-20	Level the aircraft. Refer to AMM 8-20-00.	
2	RFM	Perform aircraft weighing. Follow procedures refer to Flight Manual Section 6.5.	
3	CMM	Record the weighing result and calculate C.G using form SCA/MTC/025.	
*** End of Appendix B01 Items ***			

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ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
25/91/00 /000/000 /030	25-91	**Cargo sling installation** Tropical and damp atmosphere Salt-laden atmosphere Cleaning. CLN **AMM 25-92-00, 3-2**	
*** End of Appendix C01 Items ***			

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ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
21/51/00 /000/000 /005	21-51	**Freon air conditioning system** Check for correct operation if not used during 1 M. FT **AMM 21-51-02, 2-1**	
25/66/20 /000/000 /020	25-66	**Emergency locator transmitter** KANNAD INTEGRA AP-H (ER). Self-test. FT **AMM 25-66-20, 5-1**	
25/91/00 /000/000 /040	25-91	**Cargo sling installation** Normal climatic conditions Cleaning. In the event of intensive use of the cargo hook, it is recommended to reduce the interval by half. CLN **AMM 25-92-00, 3-2**	
*** End of Appendix C02 Items ***			

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Signature : _____ Place/Date : _____

Reg. Mark : PK - _____ Date : _____

MSN : _____ Station : _____

TSN / CSN : _____ WO No. : _____

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
28/00/00 /000/000 /010	28-00	**Fuel tank** Tropical and damp atmosphere Check and maintenance. DI **MTC 20.08.06.401**	
*** End of Appendix C03 Items ***			

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MSN : _____ Station : _____

TSN / CSN : _____ WO No. : _____

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
24/33/00 /000/000 /020	24-33	**Battery** 150CH-1 (7050A4243027) 151CH-1 (7050A4243040) Hot climatic conditions Check. DI **AMM 24-33-00, 6-1**	
26/22/00 /000/000 /010	26-22	**Cabin fire extinguisher** H1-10AIR (704A32810008) Check. WGH **AMM 26-21-00, 6-1**	
*** End of Appendix C04 Items ***			

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Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
25/67/00 /000/000 /075	25-67	**Floats** Check. Interval starting from the date of manufacture. DI **CMM 25-69-20** **INSPECTION/CHECK** **§4**	
*** End of Appendix C05 Items ***			

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Reg. Mark : PK - _____ Date : _____

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TSN / CSN : _____ WO No. : _____

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
60/00/00 /000/000 /025	60-00	**MRH - Main Rotor Mast - TRH - Engine/MGB coupling** Operation to be carried out after having logged 144 M (margin 180 D) after initial setting to service. DI **AMM 05-60-00, 6-1**		
60/00/05 /000/000 /070	60-00	**MGB assembly - TGB assembly** Check. Operation to be carried out after having logged 144 M (margin 180 D) after initial setting to service or since the last Overhaul. DI **AMM 05-60-00, 6-1**		
62/30/00 /000/000 /025	62-30	**Rotor mast - Lower section** Detailed check. DI **AMM 62-31-00, 6-5**	**RII:**	
*** End of Appendix C06 Items ***				

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ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
21/21/00 /000/000 /020	21-21	**Demisting or Heating** Check. VC|DI **AMM 21-21-02, 6-1** **AMM 21-21-03, 6-1**	
24/00/00 /000/000 /050	24-00	**Bonding** Check. DI **AMM 24-00-00, 6-2**	
25/92/00 /000/000 /010	25-92	**Cargo sling** Check. DI|LUB **AMM 25-92-00, 6-3**	
28/11/00 /000/000 /060	28-11	**Fuel tank** Check. DI **AMM 28-11-01, 6-1**	
32/11/00 /000/000 /010	32-11	**Skid Type Landing Gear** Check. DI **AMM 32-11-00, 6-6**	
53/00/00 /000/000 /030	53-00	**Cabin floor** Check. VC **AMM 53-00-00, 6-5**	
53/10/00 /000/000 /000	53-10	**Fuel tank compartment** Check. DI **AMM 53-10-00, 6-3**	

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
53/31/00 /000/000 /010	53-31	**Tail boom-to-fuselage junction** Check. DI **AMM 53-31-00, 6-8**	
53/41/00 /000/000 /000	53-41	**Fenestron** Check. DI **AMM 53-41-00, 6-2**	
63/32/00 /000/000 /130	63-32	**MGB suspension bar fittings** Check. DI **AMM 63-32-00, 6-6**	
67/00/00 /000/000 /000	67-00	**Rotor flight control channels** Check. DI **AMM 67-00-00, 6-2** **AMM 67-00-00, 6-4**	
71/41/00 /000/000 /000	71-41	**Engine support** Check. DI **AMM 71-41-00, 6-1**	
77/00/00 /000/000 /030	77-00	**Engine parameter indicators** Functional test. FT **AMM 77-00-00, 5-1**	
79/21/00 /000/000 /000	79-21	**Oil Cooler** AMM 79-21-00, 3-1 Cleaning. CLN **AMM 79-21-00, 3-1**	
*** End of Appendix C07 Items ***			

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MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
62/11/00 /000/000 /065	62-11	**Main rotor blade** Sand-laden and/or dust-laden atmosphere Check for erosion. In the event of intensive operation in heavily sand-laden areas, all the necessary measures should be taken to ensure optimum operation of the aircraft. GVI **AMM 62-11-00, 6-1**	
*** End of Appendix C08 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
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Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
62/11/00 /000/000 /070	62-11	**Main rotor blade** Tropical and damp atmosphere Cleaning the blades. CLN **AMM 62-11-00, 3-1**	
62/11/00 /000/000 /080	62-11	**Main rotor blade** Salt-laden atmosphere Cleaning the blades. After hovering or low-altitude flight, this operation should be carried out during the P check. **AMM 62-11-00, 3-1**	
*** End of Appendix C09 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
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MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
62/30/00 /000/000 /120	62-30	**Swashplates** Visual check of guide. GVI **AMM 62-32-00, 6-2**	
62/30/00 /000/000 /130	62-30	**Swashplates** Visual check of guide. GVI **AMM 62-32-00, 6-1**	
80/00/02 /000/000 /220	80-00	**Starter generator** 515-030 (704A46101018) 524-031 (704A46101011) **THALES.** Check of radial play. DI **AMM 80-00-02, 6-4**	
*** End of Appendix C10 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
25/11/00 /000/000 /010	25-11	**Energy absorption front seat** SICMA 159. Detailed check. DI **AMM 25-21-00, 6-7**	
25/11/00 /000/000 /060	25-11	**Energy absorption front seat** SICMA 198. Detailed check. DI **AMM 25-21-00, 6-3**	
25/21/00 /000/000 /090	25-21	**Rear seat** SICMA 284. Detailed check. DI **CMM 25-12-32**	
25/21/00 /000/000 /100	25-21	**Rear seat - Fitting** SICMA 284. Detailed check. DI **AMM 25-22-01, 6-1**	
*** End of Appendix C11 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - _____ Date : _____

MSN : _____ Station : _____

TSN / CSN : _____ WO No. : _____

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
25/11/00 /000/000 /020	25-11	**Energy absorption front seat - Rails** SICMA 159. Check. DI **AMM 25-21-00, 6-2**	
25/11/00 /000/000 /070	25-11	**Energy absorption front seat - Rails** SICMA 198. Check. DI **AMM 25-21-00, 6-1**	
*** End of Appendix C12 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
21/51/00 /000/000 /080	21-51	**Probe** TH90. Functional test. FT **AMM 21-51-02, 5-5**	
*** End of Appendix C13 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
26/11/00 /000/000 /010	26-11	**Engine fire detector** Checking the engine fire detector opening threshold. DI **AMM 26-11-00, 6-1**	
*** End of Appendix C14 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - _____ Date : _____

MSN : _____ Station : _____

TSN / CSN : _____ WO No. : _____

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
21/21/00 /000/000 /000	21-21	**Air intakes – Radiator** Cleaning. Optional – In the event of intensive operation in salt-laden atmosphere, sand-laden and/or dust-laden atmosphere, tropical and damp atmosphere, all the necessary measures should be taken to ensure optimum operation of the aircraft, if necessary, by shortening the suggested intervals. CLN **AMM 21-00-00, 7-1**	
21/51/00 /000/000 /060	21-51	**Cockpit and cabin evaporator harnesses - Condenser** Cleaning. Optional – In the event of intensive operation in salt-laden atmosphere, sand-laden and/or dust-laden atmosphere, tropical and damp atmosphere, all the necessary measures should be taken to ensure optimum operation of the aircraft, if necessary, by shortening the suggested intervals. CLN **AMM 21-00-00, 7-1**	
*** End of Appendix C15 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
25/67/00 /000/000 /040	25-67	**Emergency floatation gear** Check and functional test. DI|FT **AMM 25-67-00, 5-1** **AMM 25-67-00, 6-1**	**RII:**	
*** End of Appendix C16 Items ***				

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - _____ Date : _____

MSN : _____ Station : _____

TSN / CSN : _____ WO No. : _____

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
53/31/00 /000/000 /040	53-31	**Intermediate structure/Tail boom coupling** PRE MOD 074581 Visual check and tightening torque check. GVI|TCK **AMM 53-31-00, 6-1**	**RII:**	
*** End of Appendix C17 Items ***				

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
28/41/01 /000/000 /000	28-41	**Low fuel level warning** Functional test. FT **AMM 28-41-01, 5-2**	**RII:**	
29/00/00 /000/000 /010	29-00	**Hydraulic system** **Salt-laden atmosphere** Filling / Draining of the hydraulic system. Servicing of the strainer and the filter. DRN **AMM 29-00-00, 3-6**		
*** End of Appendix C18 Items ***				

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - _____ Date : _____

MSN : _____ Station : _____

TSN / CSN : _____ WO No. : _____

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
62/21/00 /000/000 /205	62-21	**Spring-type vibration damper** Check. DI **AMM 62-21-00, 6-11**	**RII:**	
63/30/00 /000/000 /070	62-30	**Laminate suspension blocks** Remove and inspect. DI **AMM 63-31-00, 6-1**	**RII:**	
63/30/00 /000/000 /080	62-30	**Laminate suspension block supports** Remove and inspect. DI **AMM 63-31-00, 6-2**	**RII:**	
63/30/00 /000/000 /090	62-30	**Laminate suspension blocks – attachment bolts** Remove and inspect. DI **AMM 63-31-00, 6-3**	**RII:**	
*** End of Appendix C19 Items ***				

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - _____ Date : _____

MSN : _____ Station : _____

TSN / CSN : _____ WO No. : _____

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
25/67/00 /000/000 /016	25-67	**Cylinder** 200740-2 (704A42693012) Proof-test. 1 OPC = 1 utilization (ditching or untimely percussion of the inflation cylinder). To be returned to the manufacturer or to an approved workshop. Interval starting from the date of manufacture (TSM), then from the date of the last proof test. NPT	
*** End of Appendix C20 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - _____ Date : _____

MSN : _____ Station : _____

TSN / CSN : _____ WO No. : _____

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
25/91/00 /000/000 /085	25-91	**Cargo sling installation** Cleaning and greasing. In the event of intensive use of the cargo hook, it is recommended to reduce the interval by half. OPH = Operating hours logged with underslung loads. CLN|LUB **AMM 25-92-00, 3-3**	
*** End of Appendix C21 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - _____ Date : _____

MSN : _____ Station : _____

TSN / CSN : _____ WO No. : _____

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
25/91/00 /000/000 /090	25-91	**Cargo sling installation** Detailed check. OPH = Operating hours logged with underslung loads. DI **AMM 25-92-00, 6-4**	
*** End of Appendix C22 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - _____ Date : _____

MSN : _____ Station : _____

TSN / CSN : _____ WO No. : _____

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
25/91/00 /000/000 /060	25-91	**Cargo sling installation** Check and functional test. OPH = Operating hours logged with underslung loads. GVI|FT **AMM 25-92-00, 6-1**	**RII:**	
*** End of Appendix C23 Items ***				

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
62/21/00 /000/000 /145	62-21	**Spherical bearing** 57910700 (704A33633211) LB4-1231-1 (704A33633208) Check of the elastomer part. GVI **AMM05-40-00,6-7**	
62/21/00 /000/000 /185	62-21	**Frequency adapter** 365A31-1019-25 (-) E-4165F01 (704A33640088) E-4165F11 (704A33640100) Check of the elastomer part. Must not be mixed together. GVI **AMM05-40-00,6-7**	
*** End of Appendix D01 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
63/21/00 /000/000 /180	63-21	**Epicyclic reduction gear - Non electrical magnetic plug** Visual check GVI **AMM 60-00-00, 6-2**	
63/21/00 /000/000 /200	63-21	**MGB - Non electrical magnetic plug** Visual check. GVI **AMM 60-00-00, 6-2**	
*** End of Appendix D02 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
62/21/00 /000/000 /155	62-21	**Starflex star** 350A31-1918-00 (-) Check GVI **AMM 05-40-00, 6-7**	
63/21/00 /000/000 /190	63-21	**Epicyclic reduction gear - Electrical chip detector** Visual check. GVI **AMM 60-00-00, 6-2**	
63/21/00 /000/000 /210	63-21	**MGB - Electrical chip detector** Visual check. GVI **AMM 60-00-00, 6-2**	
65/21/00 /000/000 /110	65-21	**TGB - Electrical chip detector** Visual check. GVI **AMM 60-00-00, 6-2**	
*** End of Appendix D03 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
65/11/00 /000/000 /165	65-11	**Center shaft section sleeves** 704A33-698-027 (-) Check GVI **AMM 65-11-00, 6-12**	
*** End of Appendix D04 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
62/11/00 /000/000 /035	62-11	**Main rotor blade** 355A11-0030-04 (-) Check of the skin. Check for cracks. GVI **AMM 62-11-00, 6-3**	
65/11/00 /000/000 /025	65-11	**Bearing** 593404 (704A33651181) Check without removal. GVI **AMM 65-11-00, 6-15**	
*** End of Appendix D05 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK	ENGINEER'S SIGNATURE, STAMP & DATE
55/11/00 /000/000 /020	55-11	**Tail boom / Fenestron junction frame** POST MOD 074775 & PRE MOD 074581 Visual check for absence of cracks. GVI **AMM 55-11-00, 6-4**	
62/11/00 /000/000 /045	62-11	**Main rotor blade** 355A11-0030-04 (-) Monitoring of the stainless steel shielding adhesion using tapping test. NOTE: On new blade or after replacement of one or both stainless steel shieldings, the new stainless steel shieldings are corvered by a shorter interval, every 150 FH during the first 450 FH. DI **AMM 62-11-00, 6-1**	
65/11/00 /000/000 /030	65-11	**Bearing** 593404 (704A33651181) Check and greasing bearing individually after disconnection bearing block. DI|LUB **AMM 65-11-00, 3-1** **AMM 65-11-00, 6-5**	
*** End of Appendix D06 Items ***			

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
64/21/00 /000/000 /022	64-21	**Torsion tie bar** 350A33-3180-00 (-) 350A33-3180-01 (-) Tropical and damp atmosphere. Salt-laden atmosphere. Detailed check whatever the time spent in specific climatic conditions. Must not be mixed together. DI **AMM 64-21-00, 6-27**	**RII:**	
*** End of Appendix D07 Items ***				

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
53/31/00 /000/000 /020	53-31	**Cone skins junctions** POST MOD 074581 Detailed inspection of junction of cone with composite spacer and junction of cone metallic skins. DI **AMM 53-31-01, 6-3**	**RII:**	
*** End of Appendix D08 Items ***				

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

ITEM CODE NO.	CHAPTER	TASK		ENGINEER'S SIGNATURE, STAMP & DATE
63/30/00 /000/000 /065	63-30	**MGB suspension cross bar** 350A38-1040-00 (-) 350A38-1040-20 (-) 350A38-1041-00 (-) 350A38-1041-20 (-) Remove and inspect. DI **AMM 63-31-00, 6-4**	**RII:**	
*** End of Appendix D09 Items ***				

PERSONNEL PRTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____