PT. SMART CAKRAWALA AVIATION

WORK ORDER

Form: SCA/MTC/030

Subject : **Inspection 1600Hrs&Add.Task**	No.	WO/043-SNP/V/2023
	Date	25 May 2023
	A/C Reg.	PK-SNP C208B-5495
Reference : MP C208B ISSUED 01	Prepared By	TS
	Checked By	CI
	Approved By	TM

To : Engineer In Charge

Description :

1. Perform Inspection 1600 Hrs & Add.Task
2. Make an entry in Maintenance Log.
3. Return the Completed Work Order and Form to PPC.

#If any finding, please close the routine card, and transferred to inspection card.

Additional Work :

Compliance Statement	Sign & Date Company Lic. No.: (Engineer In Charge)	Signature (Technical Manager)

AIRCRAFT CHECK WORK SUMMARY
(Form: SCA/MTC/051)

DATE OF ISSUED	JO/WO #	TYPE OF MAINTENANCE	DATE OF ACCOMPLISHED
25 May 2023	WO/043-SNP/V/2023	Inspection 1600 Hrs & Add.Task	

A/C Type	Mfg. Serial Number	A/C Registration
C208B	C208B-5495	PK-SNP

AIRCRAFT DATA

Subject	Pos #	Serial Number (SN)	TTSN/TCSN
Engine	#1	PCE-VA0723	
	#2	-	
Propeller/Rotor	#1	181158	
	#2	-	
Landing Gear	NLG		
	LH MLG		
	RH MLG		

PACKAGE COVERED

No	Subject	Qty	Remark
1	Non-Routine Card	5	
2	Inspection Card	1	
3	Work Order	1	
4	Summary Inspection List	1	
5	Material and Tool List	-	
6	Escalation form	-	
7	CRS (SMI / Unscheduled Maintenance)	1	

INSPECTION CARD (IC) LIST (Finding during maintenance)

No	Taskcard Ref	Subject	Status		Name/ Sign & Stamp
			Open	Close	
IC-001					
IC-002					
IC-003					
IC-004					
IC-005					
IC-006					

IC-007					
IC-008					
IC-009					
IC-010					
IC-011					
IC-012					
IC-013					
IC-014					
IC-015					

Prepared by :	Checked by :	Verified by :	Approved by :
Technical Support	Chief Maintenance	Chief Inspector	Technical Manager
			
.....	.....	.....	.....
Hani	Dodit	Yanuar	Istiono

SUMMARY INSPECTION ITEMS
(Form: SCA/MTC/050)

WO Ref: WO/043-SNP/V/2023

NO.	TASK CARD NO.	DESCRIPTION	DATE	EST MHR	NAME	STAMP
1	APPENDIX B08	PT6A-140 ENGINE GROUND RUN PERFORMANCE				
2	APPENDIX D08	ENGINE PT6A-140 100 HOUR INSPECTION/ MINOR INSPECTION				
3	APPENDIX C03	200HRS/12MTS INSPECTION				
4	APPENDIX C05	400HRS/12MTS INSPECTION				
5	APPENDIX C04	400HRS/24MTS INSPECTION				
6	APPENDIX C07	800HRS/12MTS INSPECTION				
7	APPENDIX C06	800HRS/24MTS INSPECTION				
8	APPENDIX C09	1600HRS/24MTS INSPECTION				
9	APPENDIX C08	1600HRS.60MTS INSPECTION				
10	NRC-01	REPLACEMENT VACUUM CENTRAL AIR FILTER				
11	NRC-02	REPLACEMENT VACUUM RELIEF VALVE FILTER				
12	NRC-03	REDUNDANT CONNECTION CHECK G1000 NXI				
13	NRC-04	ELECTRICAL BONDING RESITANCE CHECK G1000 NXI				
14	NRC-05	ELT BATTERY REPLACEMENT				
15	NRC-06	ELT TEST				
16	SCA/MTC/023	EMERGEN9CY EQUIPMENT CHECK				

PT. SMART CAKRAWALA AVIATION

CERTIFICATE RETURN TO SERVICE

SCHEDULED MAINTENANCE INSPECTION
(CRS-SMI)

A/C TYPE	: CESSNA 208B	TTSN	:
A/C REG	: PK-SNP	TCSN	:
MSN	: C208B-5495	DATE	:

TYPE OF INSPECTION : INSPECTION 1600 HRS&ADD.TASK
DUE AT : 4791.50 HOURS
REFF : MP C208B ISSUED 01

EXCEPTION

AUTHORIZED PERSON

I hereby certify that this aircraft has been maintained accordance with CASR and Maintenance Program. Aircraft safe and airworthy for flight

NAME	CAT	AMEL/OTR NO	SIGN&STAMP	DATE
	AIRFRAME & POWER PLANT			
	EIRA			

THE NEXT DUE TYPE OF INSPECTION :

DUE AT :

Form: SCA/MTC/049

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/043-SNP/v/2023		**REPLACEMENT COMPONENT**	**PK-SNP**
5. CARD #	6. ATA SPEC	7. TRADE	8. STA
#01	**37**		
9. ZONE	10. PANEL		

11. DESCRIPTION

PERFORM VACUUM SYSTEM CENTRAL AIR FILTER REPLACEMENT
P/N: D9-18-1 / C294502-0201

REFERENCE	[X] AMM Ch. 37-10-00-960	[ ] EMM Ch	[ ] OTHER
RII (*)	[ ] Y	[X] N	MHR :

12. RESULT				MECH	ENG	INSP (*)
Performed at A/C TT : ..... A/C TC /LDG :						
FINDING	[ ] Y	[ ] N	ACT MHR :	DATE/TIME (DD/MM/YY)		
INSPECTION CARD (IC) #						

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS
FILTER ELEMENT	D9-18-1 / C294502-0201	1		

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/043/SNP/V/2023		**REPLACEMENT COMPONENT**	**PK-SNP**
5. CARD #	6. ATA SPEC	7.TRADE	8. STA
#02	**37**		
9. ZONE	10. PANEL		

11. DESCRIPTION

PERFORM VACUUM RELIEF VALVE FILTER REPLACEMENT
P/N: B3-5-1 / C482001-0202

REFERENCE	[X] AMM Ch. 37-10-00-961	[ ] EMM Ch	[ ] OTHER
RII (*)	[ ] Y	[X] N	MHR :

12. RESULT				MECH	ENG	INSP (*)
Performed at A/C TT : ..... A/C TC /LDG :						
FINDING	[ ] Y	[ ] N	ACT MHR :	DATE/TIME (DD/MM/YY)		
INSPECTION CARD (IC) #						

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS
FILTER ELEMENT	B3-5-1 / C482001-0202	1		

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS

VACUUM DISTRIBUTION - INSPECTION/CHECK

1. General

A. This section has the inspections and checks necessary to keep the vacuum distribution system in a serviceable condition.

TASK 37-10-00-960

2. Vacuum System Central Air Filter Discard

CAUTION: Do not operate the vacuum system with the filter removed or a vacuum line disconnected. Dust and other foreign objects can enter the system and damage the vacuum operated instruments.

A. General

(1) This task gives the instructions to discard the vacuum system central air filter.

B. Special Tools

(1) None

C. Access

(1) None

D. Discard the Vacuum System Central Air Filter.

(1) Remove the vacuum system central air filter. Refer to Chapter 12, Vacuum System Central Air Filter - Servicing.

(a) Discard the filter.

(2) Install a new vacuum system central air filter. Refer to Chapter 12, Vacuum System Central Air Filter - Servicing.

E. Restore Access

(1) None

END OF TASK

TASK 37-10-00-961

3. Vacuum Relief Valve Filter Discard

CAUTION: Do not operate the vacuum system with the filter removed or a vacuum line disconnected. Dust and other foreign objects can enter the system and damage the vacuum operated instruments.

A. General

(1) This task gives the instructions to discard the vacuum relief valve filter.

B. Special Tools

(1) None

C. Access

(1) None

D. Discard the Vacuum Relief Valve Filter.

(1) Get access to the relief valve behind the attitude gyro.

(2) Carefully stretch the foam element filter over the top of the retaining bezel.

(3) Remove the filter from the relief valve and discard it.

(4) Stretch a new relief valve filter over the top of the retaining bezel.

(5) Make sure that the filter is secure on the relief valve.

E. Restore Access

(1) None

END OF TASK

VACUUM SYSTEM CENTRAL AIR FILTER - SERVICING

1. General

A. The vacuum system central air filter keeps dust and dirt from entering the vacuum operated instruments.

CAUTION: Do not operate vacuum system with filter removed or vacuum line disconnected, as dust and other foreign matter may enter the system and damage the vacuum operated instruments.

B. Refer to Chapter 5, Inspection Time Limits for filter inspection intervals. Replace filter element when damaged and whenever it becomes sufficiently clogged to cause suction gage reading to drop below 4.5 inches Hg (mercury).

CAUTION: Smoking during system operation will cause premature filter clogging.

2. Servicing

A. Remove Air Filter (Refer to Figure 301).

(1) Unscrew bolt and washer from bottom of central air filter.

(2) Remove central air filter from filter bracket.

(3) Inspect for damage, deterioration and contamination. Clean or replace as required.

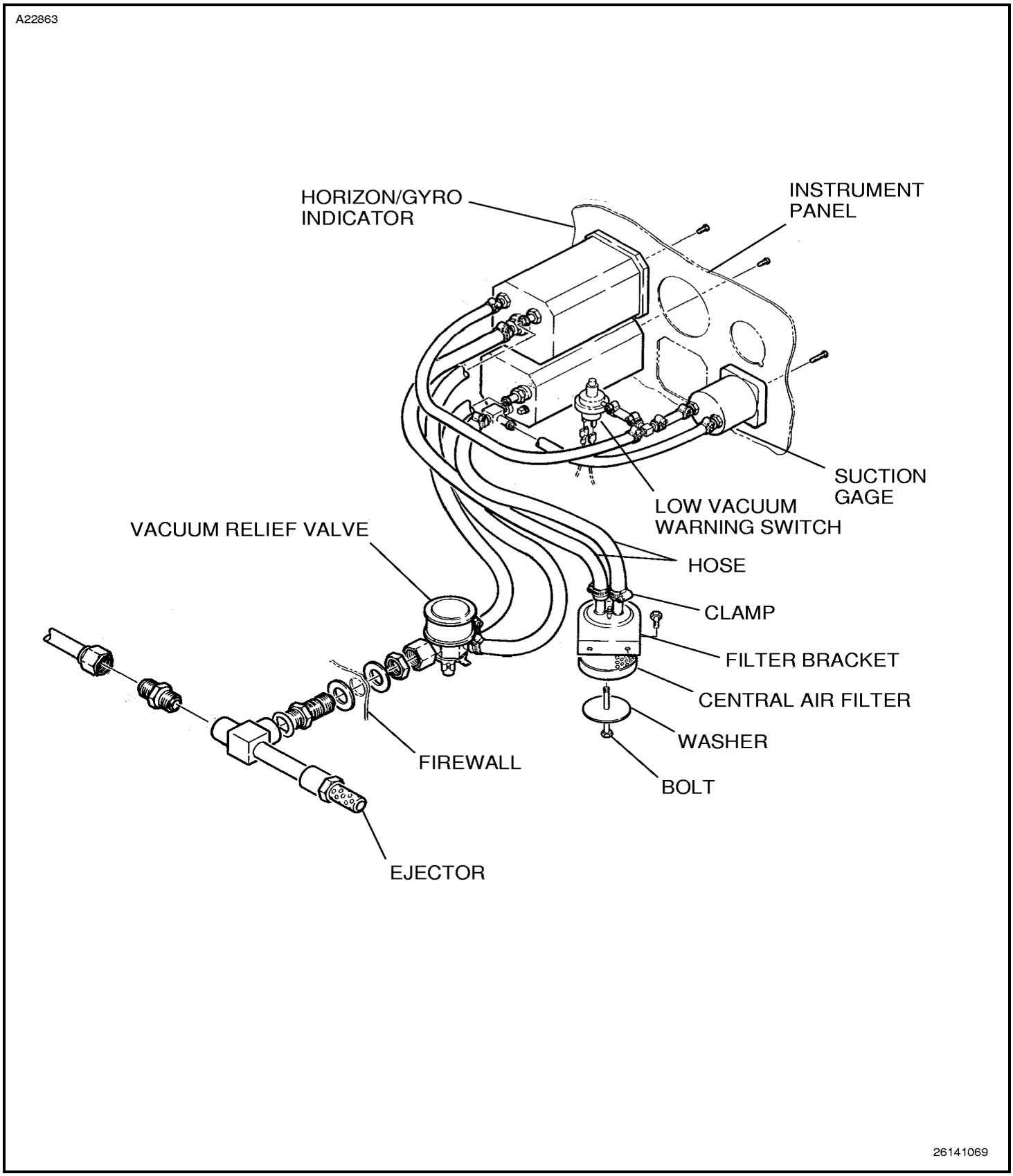
B. Install Air Filter (Refer to Figure 301).

(1) Seat central air filter up and into filter bracket.

(2) Secure central air filter to filter bracket using bolt and washer.

(3) Check central air filter for unobstructed flow. A properly functioning filter should allow a reading of at least 4.5 inches Hg (mercury) on the instrument panel suction gage.

Figure 301 : Sheet 1 : Vacuum System Central Air Filter Servicing



NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/043-SNP/V/2023		**INSPECTION**	**PK-SNP**
5. CARD #	6. ATA SPEC	7. TRADE	8. STA
#03	**34**		
9. ZONE	10. PANEL		

11. DESCRIPTION

Perform Redundant Connection Check G1000 NXi

REFERENCE	[X] G1000 Manual 190 - 02128 - 04	[ ] EMM Ch	[ ] OTHER
RII (*)	[ ] Y	[X] N	MHR :

12. RESULT				MECH	ENG	INSP (*)
Performed at A/C TT : ..... A/C TC /LDG :						
FINDING	[ ] Y	[ ] N	ACT MHR :	DATE/TIME (DD/MM/YY)		
INSPECTION CARD (IC) #						

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS
FILTER ELEMENT	D9-18-1 / C294502-0201	1		

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/043-SNP/V/2023		**INSPECTION**	**PK-SNP**
5. CARD #	6. ATA SPEC	7. TRADE	8. STA
#04	**34**		
9. ZONE	10. PANEL		

11. DESCRIPTION

Perform the electrical bonding resistance check of G1000 NXi equipment

REFERENCE	[X] G1000 Manual 190 - 02128 - 04	[ ] EMM Ch	[ ] OTHER
RII (*)	[ ] Y	[X] N	MHR :

12. RESULT				MECH	ENG	INSP (*)
Performed at A/C TT : ..... A/C TC /LDG :						
FINDING	[ ] Y	[ ] N	ACT MHR :	DATE/TIME (DD/MM/YY)		
INSPECTION CARD (IC) #						

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS
FILTER ELEMENT	D9-18-1 / C294502-0201	1		

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS

4.4 Electrical Bonding Test

The following bonding tests are provided for G1000-equipped Nav III Series aircraft as a requirement beyond what is given in the 172, 182, or 206 Series Maintenance Manual.

4.4.1 Requirements

All G1000 equipment must be installed. Gain access for the procedure listed below in Section 4.4.3 as required and in accordance with the Cessna Nav III Series Maintenance Manuals.

4.4.2 Test Equipment

A milliohm meter and Kelvin probes are recommended for this test. However, an alternate method may be used to provide equivalent results by using the following procedure and a standard voltmeter, power supply with adjustable current limit, and ammeter. The test set up for this alternate method is described below.

All test equipment used for the bond checks must be calibrated.

1. Connect the positive lead of the power supply to airframe ground. Connect/touch the positive lead of the voltmeter to the same point.

NOTE

Ensure that the voltmeter and power supply probes do not touch, so as not to induce contact resistance.

2. Touch negative lead of power supply to each of the test points listed while performing Step 3. At each required point, configure the power supply to produce 1 amp before measuring voltage. (Use an ammeter to ensure current is within *minimum* of 1 amp ±100 milliamp at each point). Do not allow the reference current to exceed 1.5 amps for safety.
3. With the current set to 1A, the voltage reading will be the value of the bonding resistance. Set the voltmeter to measure millivolts and null the reading. Measure the voltage from airframe ground (step 1) to each required test points and record the voltage. (Perform Step 2 at each required point and ensure that *minimum* of 1 amp ±100 milliamp is present when measuring the voltage.)

TIP: When a 1A current is used all the millivolt readings are the same as milliohms, and required no further calculation of bond resistance. If 1 amp reference current cannot be maintained and is higher divide the measured voltage by current to get the resistance value. Alternatively, calculate the percentage increase in current and then reduce the measured voltage reading by the same percentage. Example: If the measured current is 1.2 amps, (20% high from the desired 1 amp current) and the measured voltage is 3.0 millivolts, then the value recorded will be 3 millivolts reduced by 20% which is 2.4 millivolts which is the same as 2.4 milliohms.

4.4.3 Electrical Bonding Procedure

Using one of the two measurement methods in Section 4.4.2 record the bonding measurement for the following equipment. Some equipment in the list is optional and may not be installed.

Ensure that the PFD unit, and MFD unit test points, no more than 40 mΩ is present. Ensure that at other unit test points, no more than 5 mΩ is present.

Pilot Compartment

- Metal case of PFD: _____ mΩ
- Metal case of MFD: _____ mΩ
- GDL 69A body (182 and 206 only): _____ mΩ

Rear Fuselage and Empennage

- GDL 69A body (172 only): _____ mΩ
- GSU 75 body: _____ mΩ
- GTX 335R or GTX 345R body: _____ mΩ

4.5 GSU 75 Earth Magnetic Field Updates

The GSU 75, utilizes an Earth magnetic field model which is updated once every five years. The update is expected to be available from Garmin every five years, as long as the GSU 75, remain Garmin-supported products.

The G1000 system alerts the operator that the magnetic field database is out of date by issuing the message “AHRS SERVICE – AHRS Magnetic-field model needs update”. Garmin will distribute updates as part of the navigation database cycle. If the IGRF model in the update is newer than the model installed, a prompt will appear for each GRS/GSU (Select “OK” to update the IGRF Magnetic Field Model). Use the FM Knob and ENT key to select “OK”.

4.6 G1000 Redundant Connection Check

Perform the following steps to verify the following:

(A) The primary (PRI) and secondary (SEC) power sources for the PFD, GSU (if installed) or GDC and GRS (if installed).

(B) The GSU (if installed) data path to PFD and MFD.

Note: Section B is not required if a GSU 75 is not installed. Refer to the Cessna Nav III Line Maintenance Manual for procedures if a GDC 74A and GRS 77 are installed..

(A) PRI and SEC power sources for PFD and GSU (if installed) check

1. Connect a ground power unit to the external power receptacle and turn on the ground power unit.
2. Set the BAT and AVIONICS MASTER PWR switches to ON.

4.4.3 Electrical Bonding Procedure

Using one of the two measurement methods in Section 4.4.2 record the bonding measurement for the following equipment. Some equipment in the list is optional and may not be installed.

Ensure that the PFD unit, and MFD unit test points, no more than 40 mΩ is present. Ensure that at other unit test points, no more than 5 mΩ is present.

Pilot Compartment

- Metal case of PFD: _____ mΩ
- Metal case of MFD: _____ mΩ
- GDL 69A body (182 and 206 only): _____ mΩ

Rear Fuselage and Empennage

- GDL 69A body (172 only): _____ mΩ
- GSU 75 body: _____ mΩ
- GTX 335R or GTX 345R body: _____ mΩ

4.5 GSU 75 Earth Magnetic Field Updates

The GSU 75, utilizes an Earth magnetic field model which is updated once every five years. The update is expected to be available from Garmin every five years, as long as the GSU 75, remain Garmin-supported products.

The G1000 system alerts the operator that the magnetic field database is out of date by issuing the message “AHRS SERVICE – AHRS Magnetic-field model needs update”. Garmin will distribute updates as part of the navigation database cycle. If the IGRF model in the update is newer than the model installed, a prompt will appear for each GRS/GSU (Select “OK” to update the IGRF Magnetic Field Model). Use the FM Knob and ENT key to select “OK”.

4.6 G1000 Redundant Connection Check

Perform the following steps to verify the following:

(A) The primary (PRI) and secondary (SEC) power sources for the PFD, GSU (if installed) or GDC and GRS (if installed).

(B) The GSU (if installed) data path to PFD and MFD.

Note: Section B is not required if a GSU 75 is not installed. Refer to the Cessna Nav III Line Maintenance Manual for procedures if a GDC 74A and GRS 77 are installed..

(A) PRI and SEC power sources for PFD and GSU (if installed) check

1. Connect a ground power unit to the external power receptacle and turn on the ground power unit.
2. Set the BAT and AVIONICS MASTER PWR switches to ON.

3. With the G1000 system in normal mode, pull the following circuit breakers on the circuit breaker panel:
    - PFD (ESS BUS)
    - ADC/AHRS (ESS BUS)
    - NAV 1/ENG (ESS BUS)
4. Wait at least 5 seconds, then verify the data on the PFD remains valid and the following alert messages are not present:
    - GIA 1 FAIL (or any related message such as COM1, NAV1, XPDR1, GPS1)
    - AHRS 1 FAIL
    - ADC 1 FAIL
5. Close the circuit breakers listed in step 3.
6. Repeat step 4.
7. Pull the following circuit breakers on the circuit breaker panel:
    - PFD (AVN BUS 1)
    - ADC/AHRS (AVN BUS 1)
    - NAV 1/ENG (AVN BUS 1)
8. Wait at least 5 seconds, then repeat step 4.
9. Close the circuit breakers listed in step 7.

(B) AHRS and ADC data path check (when GSU 75 installed)

1. Verify there are no AHRS loss of data messages, such as:
    - AHRS not receiving any GPS information
    - AHRS not receiving backup GPS information
    - AHRS using backup GPS source
2. Place the G1000 system in configuration mode.
3. In the GIA page group, go to the GIA RS-232/ARINC 429 CONFIG page.
4. With GIA1 selected in the SELECT UNIT window, verify the following indicators are green checks:
    - RS232 Channel 1 (GDC72 #1)
    - RS232 Channel 6 (GRS79 #1)
    - ARINC 429 IN 6 (GSU75 #1)

(Ref. Section 5.1.1.2.)

5. With GIA2 selected in the SELECT UNIT window, verify the following indicators are green checks:
    - RS232 Channel 2 (GRS79 #1)
    - ARINC 429 IN 5 (GSU75 #1)

(Ref. Section 5.1.1.2.)

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/043-SNP/V/2023		**REPLACEMENT COMPONENT**	**PK-SNP**
5. CARD #	6. ATA SPEC	7. TRADE	8. STA
05	**34**		
9. ZONE	10. PANEL		

11. DESCRIPTION

PERFORM ELT BATTERY REPLACEMENT
P/N OFF: 452-0133 S/N OFF:377841-009
P/N ON: S/N ON:

REFERENCE	[X] CMM C406	[X] AMM 34-54-00	[ ] OTHER
RII (*)	[ ] Y	[X] N	MHR :

12. RESULT				MECH	ENG	INSP (*)
Performed at A/C TT : ..... A/C TC /LDG :						
FINDING	[ ] Y	[ ] N	ACT MHR :	DATE/TIME (DD/MM/YY)		
INSPECTION CARD (IC) #						

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS
FILTER ELEMENT	D9-18-1 / C294502-0201	1		

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS

Reg. Mark : PK -

WO/FML No. : WO/032-SNP/XII/2022

PRE – INSPECTION	
Location	
Date	
Cycle	
Filed Barometric	
OAT	
Altitude	

POST – INSPECTION	
Location	
Date	
Cycle	
Filed Barometric	
OAT	
Altitude	

PRE – INSPECTION		
	Target	Actual
Tq		
Np		
ITT	°C	°C
Ng	%	%
Wf		
Oil Press		°C
Oil Temp		°C
Start Temp		°C

POST– INSPECTION		
	Target	Actual
Tq		
Np		
ITT	°C	°C
Ng	%	%
Wf		
Oil Press		°C
Oil Temp		°C
Start Temp		°C

Engine Run Up Checks

Inertial ☐ **EPL** ☐ **OVG** ☐ **Stby Alt** ☐ **BOV** ☐ **Brake** ☐ **Randown** ☐

NOTE:

1. Brake system at Torque 2000 ft-lbs.
2. Inertial Separator at Torque 400 ft-lbs.
3. EPL check can't exceed 4% Ng per second.
4. Standby Alt at 80% Ng.
5. Low idle at 55.5 - 57% 40Amps.
6. High idle at 64 - 66% Ng 40Amps

Engine Performance Target Table (Cessna C208B EX)

OAT (°C)	27	28	29	30	31	32	33	34	35	36	37	38	39	40	41
Tq (ft.lbs)	2397	2397	2397	2397	2397	2397	2397	2397	2397	2397	2397	2397	2397	2397	2397
Np	1900	1900	1900	1900	1900	1900	1900	1900	1900	1900	1900	1900	1900	1900	1900
ITT (°C)	835	837	839	841	841	841	841	841	841	842	843	844	846	846	846
Ng (%)	102.7	102.7	102.7	102.7	102.7	102.7	102.7	102.6	102.6	102.6	102.6	102.6	102.6	102.6	102.5
WF (PPH)	578	578	578	578	578	578	578	570	565	565	560	560	555	548	548

Note:

1. Make sure that inertial separator in normal condition, no bleed air extracted from the engine and air condition OFF.
2. This table only applies to altitude 0-500 feet MSL. For higher altitude, refer to EMM 72-00-00.
3. Max fuel flow is 580 lb/hr fuel flow is not more than 15 lbs/hr higher than the value shown in table.
4. If parameters are outside the target performance table to EMM chapter 71-00-00.

REMARKS:

PERFORMED BY			
Name	Sign & Stamp	Date	Location

	EMERGENCY EQUIPMENT LIST INSPECTION & MONITOR	**PT. SMART CAKRAWALA AVIATION** DEPARTMENT TEKNIK Form: **SCA/MTC/023**

DATE :	**A/C REG : PK-SNP**
A/C TYPE : C208B	**CHECKER : SIGN:**

No.	Description	P/N	S/N	Next Insp.	Remarks
1	Pilot Life Vest				
2	Co-Pilot Life Vest				
3	Pax Life Vest				
4	Pax Life Vest				
5	Pax Life Vest				
6	Pax Life Vest				
7	Pax Life Vest				
8	Pax Life Vest				
9	Pax Life Vest				
10	Pax Life Vest				
11	Pax Life Vest				
12	Pax Life Vest				
13	Firt Aid Kit				
14	Crash Axe Installed				
15	Fire Extinguisher				
16	Life Raft (If Installed)				
17	Survival Kit (If Installed				
OTHERS					

Additional Work Sheet Insp. 1600 Hrs & Add.Task

Aircraft Registration: **PK-SNP**

WO# Nr: **WO/043-SNP/V/2023**

Parts Used Sheet

Special Tool Used

No.	Description	Due Date	Engineer

smartaviation

Additional Work Sheet Inspection 1600 Hrs & Add.Task

Aircraft Registration: **PK-SNP**

WO# Nr: **WO/043-SNP/V/2023**

Parts Used Sheet

Part Used

Date	Part Nr.	Serial Nr.	Description	Quantity	Engineer