



PT. SMART CAKRAWALA AVIATION

WORK ORDER

Form: SCA/MTC/030

Subject : Inspection 100 Hours / Annual Due at 200 FH.	No.	WO/003-SNC/XII/2021
	Date	22-Dec-2021
	A/C Reg.	PK-SNC MSN 1016
Reference : MP PILATUS PC-6 Rev. 0	Prepared By	TS
	Checked By	CI
	Approved By	TM
To : Engineer In Charge		
Description : <ol style="list-style-type: none">1. Perform Inspection 100 Hours / Annual Due at 200 FH.2. Make an entry in Maintenance Log.3. Return the Completed Work Order and Form to PPC. <p>#If any finding, please close the routine card, and transferred to inspection card.</p>		
Additional Work : - Removal LH & RH UNDERWING TANKS FOR COMPANY CONVENIENCE. BY		
Compliance Statement WORK CARRIED OUT FOUND SATISFACTORY	Sign & Date 08 JAN 2022 Company Lic. No.: 5709  (Engineer In Charge) 	Signature (Technical Manager)



PT. SMART CAKRAWALA AVIATION

CERTIFICATE RETURN TO SERVICE

SCHEDULED MAINTENANCE INSPECTION (CRS-SMI)

A/C TYPE : PILATUS PORTER PC-6	TTSN : 196:46
A/C REG : PK-SNC	TCSN : 171
MSN : 1016	DATE : 08 JAN 2022





TYPE OF INSPECTION : INSPECTION 100 HOURS / ANNUAL
 DUE AT : 200 HOURS
 REFF : MP PILATUS PC-6 REV. 0

EXCEPTION

NIL

AUTHORIZED PERSON

I hereby certify that this aircraft has been maintained accordance with CASR and Maintenance Program.
 Aircraft safe and airworthy for flight

NAME	CAT	AMEL/OTR NO	SIGN&STAMP	DATE
BRAMONO ONY.T	AIRFRAME & POWER PLANT	5709 SCA-16		08 /
	EIRA			JAN 2022

THE NEXT DUE TYPE OF INSPECTION : 100 hrs
 DUE AT : 300:00 hrs.

Form: SCA/MTC/049

AIRCRAFT CHECK WORK SUMMARY
(Form: SCA/MTC/051)

DATE OF ISSUED	JO/WO #	TYPE OF MAINTENANCE	DATE OF ACCOMPLISHED	
22-Dec-2021	WO/003-SNC/XII/2021	Inspection 100 Hrs / Annual	08 JANUARY 2022	
A/C Type		Mfg. Serial Number	A/C Registration	
PILATUS PORTER PC-6		1016	PK-SNC	
AIRCRAFT DATA				
Subject	Pos #	Serial Number (SN)	TTSN/TCSN	
Engine	#1	PCE-PG0567	196:46 / 171	
	#2	-		
Propeller/Rotor	#1	FY4989		
	#2	-	196 : 46	
Landing Gear	NLG			
	LH MLG		196:46 / 171	
	RH MLG		196:46 / 171	
PACKAGE COVERED				
No	Subject	Qty	Remark	
1	Non-Routine Card	-		
2	Inspection Card	-		
3	Work Order	1		
4	Summary Inspection List	1		
5	Material and Tool List	-		
6	Escalation form	-		
7	CRS (SMI / Unscheduled Maintenance)	1		
INSPECTION CARD (IC) LIST (Finding during maintenance)				
No	Taskcard Ref	Subject	Status Open Close	Name/ Sign & Stamp
<u>IC-001</u>				
<u>IC-002</u>				
<u>IC-003</u>				
<u>IC-004</u>				
<u>IC-005</u>				
<u>IC-006</u>				

<u>IC-007</u>					
<u>IC-008</u>					
<u>IC-009</u>					
<u>IC-010</u>					
<u>IC-011</u>					
<u>IC-012</u>					
<u>IC-013</u>					
<u>IC-014</u>					
<u>IC-015</u>					

Prepared by :
Technical Support

Checked by :
Chief Maintenance

Verified by :
Chief Inspector

Approved by :
Technical Manager

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SUMMARY INSPECTION ITEMS (Form: SCA/MTC/050)

WO Ref: WO/002-SNC/XII/2021

NO.	TASK CARD NO.	DESCRIPTION	DATE	EST MHR	NAME	STAMP
1	APPENDIX	ENGINE GROUND RUN CHECK SHEET	08/ JAN '22		BRANDS ON TRIPLO	
2	APPENDIX	100 HOURS / ANNUAL INSPECTION	08/ JAN '22		BRANDS ON TRIPLO	
3	APPENDIX	FUEL DISTRIBUTION SYSTEM - ADJUSTMENT TEST	07/2022 JAN		FEBRI HERMANA	
4	APPENDIX	FUEL INDICATING SYSTEM - ADJUSTMENT	07/2022 JAN		FEBRI HERMANA	
5	APPENDIX	FUEL SYSTEM UNDERWING TANK INSPECTION TRANSFER PUMP FILTERS	06/ JAN 2022		BRANDS ON TRIPLO	
6	APPENDIX	WHEEL AND BRAKES INSPECTION	08/ JAN 2022		BRANDS ON TRIPLO	
7	NRC-001	STARTER/GENERATOR BRUSHES - CHECK FOR WEAR	07/ JAN '22		BRANDS ON TRIPLO	
8	NRC-002	PROPELLER OVERSPEED GOVERNOR - CHECK OPERATION	08/ JAN '22		BRANDS ON TRIPLO	
9	SCA/MTC/023	EMERGENCY EQUIPMENT CHECK	08/ JAN '22		BRANDS ON TRIPLO	

	ENGINE GROUND RUN CHECK SHEET - PT6A-27 ENGINE WITH FOUR BLADE PROPELLER (HARTZELL STC SA377CH)
	 

WORK ORDER NO.		: WO / 003- SNC / x11 / 2021	
Aircraft Registration	PK-SNC	Aircraft Total Hours	196 : 46
Aircraft Serial No.	1016	Aircraft Total Landings	171
Engine Serial No.	PCE-PC 0567	Engine TSN / TSO	196 : 46
Propeller Serial No	FY 4989	Propeller TSN / TSO	196 : 46
Ambient Temp	35 °C	FBP (Field Barometric Pressure)	1009 In.Hg
Date	08 JAN 2021	Time	13 : 10 LT
Mechanic / Engineer	FEBRI HERMANA	Authorized Engineer	BRANDOT OT
Reason For Ground Run		AFTER MAINTENANCE	

Checks to be carried out. No:	1 2 4 5 7 8 9 10 11 12 13 14 15
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Engine Ground Run Check Frequency

Check Number	1	2	3	4	5	6	7	8	9	10	11	12	13	14	15
Each 100 / Yearly	x	x		x			x	x			x	x	x	x	x
Each 200									x						
Pre-Complete Overhaul	x	x	x	x		x	x	x	x	x	x	x	x	x	x
After Short Term Storage															x
After Long Term Storage	x	x	x	x		x	x	x	x	x	x	x	x	x	x

In addition the following check must be carried out after Installation, Repair and Adjustment of any of the following components.

Check Number	1	2	3	4	5	6	7	8	9	10	11	12	13	14	15
Engine Installation	x	x	x	x	x	x	x	x	x	x	x	x	x	x	x
Propeller Installation		x	x	x	x			x							
Fuel Control Unit	x				x	x	x	x		x	x				
HP Fuel Pump						x	x								
Fuel Nozzle						x	x								
Starting Flow Control	x				x		x	x							
Emer Fuel Control Actuator											x				
Prop Governor	x		x	x	x		x	x							
Prop Overspeed Governor									x						
Compressor Bleed Valve						x	x								
Engine Controls	x			x	x			x	x						
Low Pitch Warning Switch				x											
Suction Components														x	



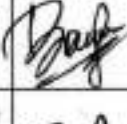

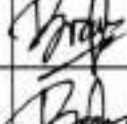

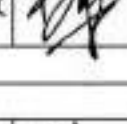

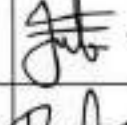

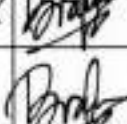

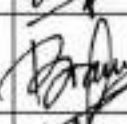

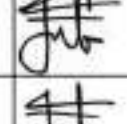

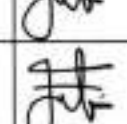

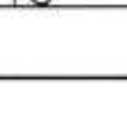
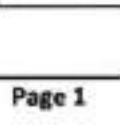


Use this sheet's to record engine run result, use in conjunction with task cards.















NO.	CHECK	TARGET	ACTUAL
ENGINE START			
	ITT (Troubleshoot If More Than 925°C)	Max. 1090 °C	532 °C ✓
	Cabin Heat	OFF	✓ OK?
1	Low Idle (Minimum Governing) Speed	51 - 53 % Ng	52.0 % Ng
	Fuel Pressure / Boost Pump OFF	Light out or 25 ± 5 psi	OK?
	ITT		542 °C ✓
	Oil Pressure		90 psi
	Oil Temperature		65 °C
2	Propeller Governor		
	Maximum Np	1980 - 2000 rpm (90.0 - 90.9 %)	1990 rpm
	Py Disconnected		90.9 % Ng
	Py Connected		91 % Ng
	Difference	Maximum 0.3% Ng	%
	Airbleed Link at Minimum	1900 - 1950 rpm (86.4 - 88.6 %)	rpm
	Aircraft with SB 161:		
	Propeller Control Lever at Minimum	1880 - 1900 rpm (85.5 - 86.4 %)	rpm
3 X	Propeller Fine Pitch Setting (High Idle)		
	Target Torque	psi	psi
	Power Lever to Give Np	1694 rpm (77 %)	rpm
	Basic High Idle	68 - 72% Ng	%Ng
4	Propeller Low Pitch Warning		
	PCL from Reverse to Detent	Light OFF	
		1 to 2 mm before Detent	1 mm ✓
5 X	Minimum Pitch in Flight		
	Ng	67 - 73 %	% Ng
	Np	1800 - 1950 rpm (81.8 - 88.6 %)	rpm
	Torque	4 - 7 psi	psi
6 X	FCU Maximum Governing Speed (Ng)	97.1 % Ng	% Ng
	(Trim stop deployed)		

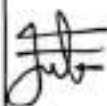

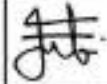

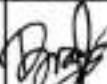

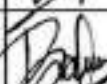



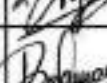



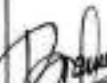



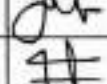

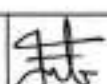

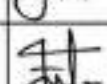

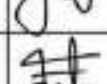

Ref. AMM Pilatus Porter PC6 Chapter 05-22-01, P&WC Maintenance Manual Model PT6A-27 Manual
Part No. 3013242 Chapter 72-00-00, Propeller Owner's Manual Hartzell (Manual 149)

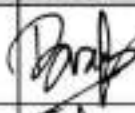

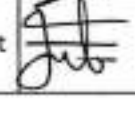



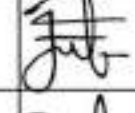

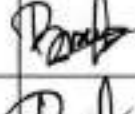

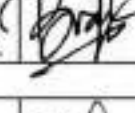

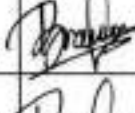



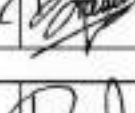

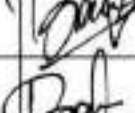

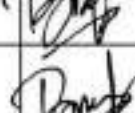

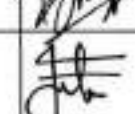

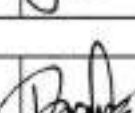

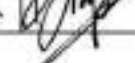

100 HOURS / ANNUAL INSPECTION

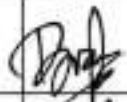

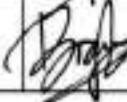



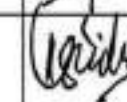

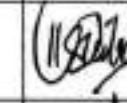

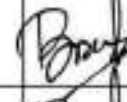

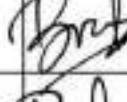

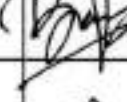







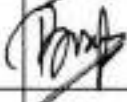

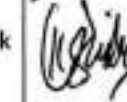

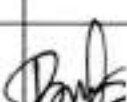



Reg. Mark : PK - SNC Date : 06 JAN 2022
MSN : 1016 Station : NAGNAG
TSN / CSN : 136 : 46 / 171 WO No. : WO/003-SNC/X11/2021





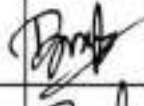

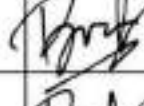

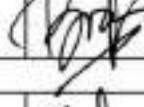

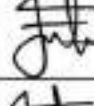

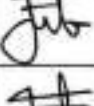

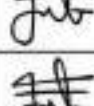

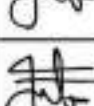

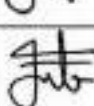

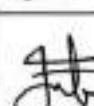

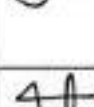

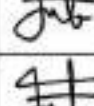



NO	TASK	SIGNATURE	
		SIGN	STAMP
1	Aircraft document Perform inspection document folder (onboard). Check content completeness of aircraft document. (Ref. CASR 91.25)		
2	Emergency equipment list Perform emergency equipment list. Form SCA/MTC/023. Make one copy and insert into the aircraft document folder.		
3	After engine run safety all screws bolts locknuts as applicable (duplicate inspection). Perform after engine run safety all screws bolts locknuts as applicable.		
4	After engine run check engine for signs of fuel, oil, air leaks. Perform after engine run check engine for signs of fuel, oil, air leaks.		
B. AIRFRAME			
Aircraft - General			
1	External surfaces Examine, particularly for fuel, oil and hydraulic leaks.		
2	Aircraft external Wash.		
3	Aircraft preparation Remove and examine the protective covers, blanks and restraints. Replace if damaged, torn or is not properly install.		
4	Placard and markings Examine and replace as necessary.		
5	Aircraft lifting Put the aircraft on jacks.		
6	Fuselage Remove access panels and fairings.		
7	Fuselage - internal Remove cockpit and cabin seats and interior fuselage linings.		



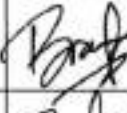

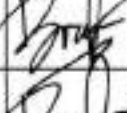

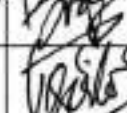

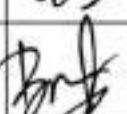

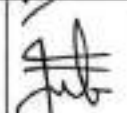

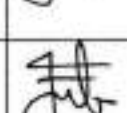

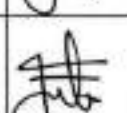

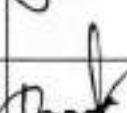









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8	Wings Remove access panels and fairings (not fuel tanks).	<i>Jul</i>	
9	Engine cowls Remove.	<i>Jul</i>	
10	Empennage Remove access panels and fairings.	<i>Jul</i>	
Chapter 21 - Air Conditioning			
1	Engine bleed air line and hoses Examine.	<i>Bray</i>	
2	Air inlet screens, filters and hoses Clean and examine.	<i>Bray</i>	
3	Mixer unit Examine.	<i>Jul</i>	
4	Butterfly vents - passenger cabin Examine.	<i>Jul</i>	
5	Emergency shut-off valve Examine.	<i>Jul</i>	
6	System component, pipes, cables, controls and linkages. Examine.	<i>Jul</i>	
7	System cables, controls and linkages Lubricate (Material No. P04-037).	<i>Jul</i>	
8	Air conditioning system Check operation during engine ground run checks.	<i>Jul</i>	
Chapter 24 - Electrical Power			
1	Generator voltage Check generator voltage at high idle under load <u>28.0</u> VDC.	<i>Bray</i>	
Chapter 25 - Equipment and Furnishings			
1	Pilot and Co-pilot seats Examine seat and seat attachments. Make sure that the seat adjustment mechanism operates correctly. Lubricate moving parts. (Material No. P04-011).	<i>Bray</i>	
2	Pilot and Co-pilot seats harnesses Examine. Inertial reel system - Operational test.	<i>Bray</i>	












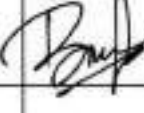





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3	Passenger seats Examine seats, seat attachments and seat harnesses. If seats with Torso Restraint System are installed, make sure the backrest release mechanism operates correctly. Lubricate moving parts (Material No. P04-028).		
4	Linings and curtains Examine.		
5	Emergency locator transmitter Examine Check battery expiry date <u>06 / 2026</u>		
6	Fire extinguisher Examine Check expiry date <u>18 NOV 2022</u>		
7	First aid kit Examine Check expiry date <u>18 NOV 2022</u>		
8	Crash axe Make sure it is stored correctly.		
9	Stretchers (if installed) Examine stretchers and mountings.	N/A	
10	Parachute dispatch system (if installed) Examine. Signal light system - Operational test.	N/A	
Chapter 27 - Flight Controls - General			
1	Control column Examine. Check for excessive play at Teflon bearing at base of column by pulling up and pushing down on column. Maximum play is 0,2 mm (0.008 in.).		
2	Control lock Examine.		
3	Rudder pedals Examine. Check for excessive play and full and free range of movement. Especially examine the brake pedal at the weld for cracks.		
Chapter 27 - Flight Controls - Ailerons			
1	Aileron control system Examine system including stops, cables, pulleys, guides, and bellcranks.		
2	Aileron controls Do a functional test.		
3	Aileron to rudder interconnect spring Examine.		



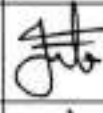

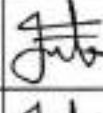

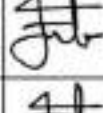

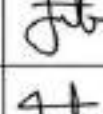

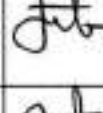

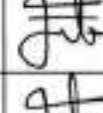

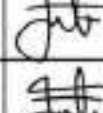

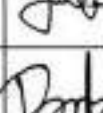

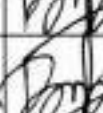

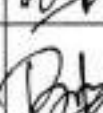

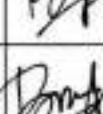

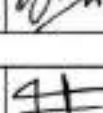

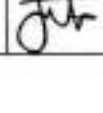

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4	Aileron trim tab electrical actuator (Electrical system) Examine.		
5	Aileron trim system (Mechanical or Electrical trim tab systems) Check neutral settings, sense and range of movement. Check cockpit indicator.		
Chapter 27 - Flight Controls - Rudder			
1	Rudder control system Examine system including stops, cables, pulleys, guides, and bellcranks.		
2	Rudder Do a functional test.		
3	Rudder trim tab electrical actuator (Electrical system) Examine.		
4	Rudder trim tab (Mechanical or Electrical trim tab systems) Do an inspection / check. Check neutral settings and range of movement. Check cockpit indicator.		
Chapter 27- Flight Controls - Elevator			
1	Elevator control system Examine system including stops, cables, pulleys, guides, and bellcranks.		
2	Elevator control system Do a functional test.		
3	Elevator balance tabs Check neutral settings, sense and range of movement.		
Chapter 27 - Flight Controls - Stabilizer			
1	Horizontal stabilizer trim actuator Electrical system Examine.		
2	Horizontal stabilizer trim actuator attachments Examine. On the Lugs, look for cracks and signs of excessive asymmetrical wear.		
3	Horizontal stabilizer trim system Examine.		
4	Electrical system Do a functional test.		
Chapter 27 - Flight Controls - Flaps			
1	Flap actuator and support bracket (Electrical system) Examine		

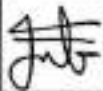

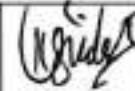



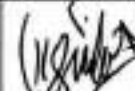



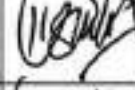

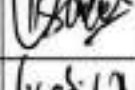

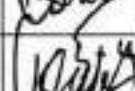

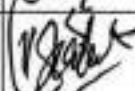

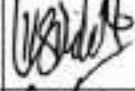

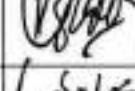

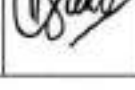

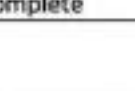

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2	Flap control system – Bellcranks, levers and push/pull rods (Elect. Sys) Examine		
3	Flaps Do a functional test.		
Chapter 28 - Fuel System			
1	Water collector tank and fuel filter Drain a minimum of 0,25 liters (0.5 pint) of fuel from each drain valve. Make sure that there is no water in the fuel.		
2	Fuel filter Examine.		
3	Fuel shut-off valve Examine.		
4	Main fuel tanks Examine vents, filler caps and seals.		
5	Fuel pipes and hoses Examine.		
6	Air maze fuel filter Examine inlet pipe and adjacent oil hose for chafing.		
7	Perform fuel filter clean (Airmaze). P/N: 968.35.21.147 S/N: NSN. P/N OFF :  P/N ON : 		
8	Fuel flow transmitter Examine.		
9	Engine driven fuel pump (EDP) Examine.		
10	Fuel system Set shut-off valve to OPEN and then set the AUX F PUMP to ON. Look for leaks on complete fuel system and unusual noise from the fuel pump. Set AUX F PUMP to OFF and then set shut-off valve to CLOSE.		
11	Fuel System Fuel distribution system test or adjustment.		
12	Fuel Indicating System Fuel indicating system test or adjustment.		

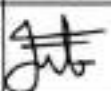

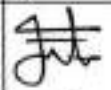

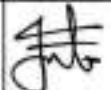

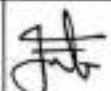



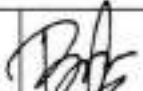



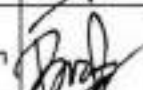





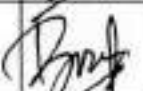

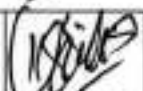

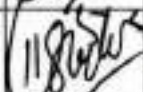

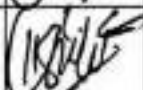



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Chapter 28 - Fuel System - Underwing Tanks			
1	Underwing tanks Examine.		
2	Transfer pump filters Examine and clean.		
3	Underwing tank system Check operation.		
4	Underwing Tank Fuel System Fuel system underwing tank inspection (if installed).		
5	Underwing Tank Fuel System Fuel system underwing tank inspection transfer pump filter.		
Chapter 32 – Landing Gear and Brakes			
1	Brakes Check brake pad wear. Visual Insp. Beringer.		
2	Main wheels rotation and debur discs. Perform main wheels LH and RH rotation and debur discs.		
3	Hydraulic pipes Examine.		
4	Brake system Check brake fluid level. Apply brakes, examine system for leaks.		
5	Park brake system Examine. Make sure system operates correctly		
6	Main wheel tires Examine.		
7	Main wheels Remove. Examine bearings, axles and wheels. Lubricate bearings and axels with grease (MIL - G - 81322). On installation rotate the wheel position LH to RH and vice versa		
8	Main wheels Perform main wheels inspection fill out the main wheel's inspection.		
9	Brake discs Examine. Check for wear.		







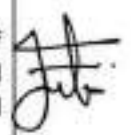

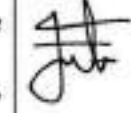

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10	V-struts Examine. Note: If you find damage that is more than 0,127 mm (0.005 in.) deep, reject the V-strut.		
11	V-struts If you find damage that is 0,127 mm (0.005 in.) deep or less, refer to Pilatus CMM 02270 for minor repair procedures.		
12	V-struts attachments Examine. Lubricate (Material No. P04-002)		
13	Main landing gear shock struts Examine. Lubricate (Material No. P04-002). Check fluid level.		
14	Main wheels Install. Inflate tire.		
15	Main wheel - dirt scraper Examine.		
16	Tail landing gear Examine. Make sure there are no cracks in the welded seams. Check the locking-lever pivot pins. Lubricate (Material No. P04-002)		
17	Tail wheel tire Examine.		
18	Tail wheel Remove. Examine bearings, axle and wheel. Lubricate bearings with grease (Material No. MIL-G-81322; Aeroshell 22, Royco22, Mobil 28)		
19	Tail wheel Install. Inflate tire.		
20	Steering system Examine. Check cable tension and range of movement.		
21	Debris guard Examine.		
22	Steering system Inspect steering cable tension with a turn buckle installed in the steering cable – adjust the turnbuckle to give a cable tension of minimum 32 lbs., maximal 35 lbs. and install two new locking clips.		





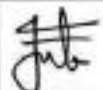









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Chapter 35 - Oxygen System			
1	Oxygen bottle(s) and attachment brackets (if installed) Examine.	7/1	
2	Oxygen system pipes, flexible tubes and fittings (if installed) Examine.	7/1	
3	Oxygen regulators (if installed) Examine.	7/1	
Chapter 52 - Doors			
1	Pilot, Co-pilot doors Examine. Remove safety wire. Make sure that the emergency release mechanism and latching mechanism operate correctly. Do the check of vertical play of the door Lubricate mechanism (Material No. P04-011). Install safety wire. (Material No. P02-021)		
2	Cabin RH / LH sliding door Examine door, sliding rails, rollers, stops and seals Make sure that the latching mechanism operates correctly. Lubricate mechanism. (Material No. P04-037)		
3	Cabin trap-door (if installed) Remove trap-door hatch cover. Examine doors, hinges, seal, and structural damage. Make sure that the latching mechanism and door release mechanism operate correctly. Test door for correct operation. Lubricate mechanism (Material No. P04-037)		
Chapter 53 - Fuselage			
1	Access panels and fasteners Examine.		
2	Fuselage - external Examine.		
3	Fuselage - internal Examine these structures as follows: - cockpit floor - cabin floor - cabin floor T-rails - door frames - accessible frames, stringers, and skin.		
4	Fuselage Make sure that the drain holes are not blocked.		
Chapter 55 - Stabilizers			

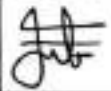

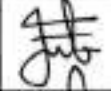





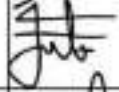



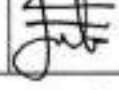









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1	Empennage Examine internal skin and structures as far as possible. Examine panels and fasteners. Make sure that the water drain holes are not blocked.		
2	Dorsal fin Examine.		
3	Vertical stabilizer Examine.		
4	Rudder - support structure Examine rudder support brackets, torque tube, control rod attachment points and attaching parts.		
5	Rudder Examine rudder skin and structure, balance weight attachment and mountings for static discharge wicks		
6	Rudder upper attachment Remove access panel EL4. Examine the attachment bolt and lockwire for security. Install access panel EL4.		
7	Rudder trim tab Examine tab, hinge, control rod attachment point and attaching parts.		
8	Horizontal stabilizer Inspection/ Check.		
9	Horizontal stabilizer actuator Examine the attachment brackets		
10	Elevator support structure Examine elevator support brackets, hinge bearings, control rod attachment points, control lever and attaching parts.		
11	Elevator Examine skin and structure, fixed tab (H4 only) and mountings for static discharge wicks		
12	Elevator attachments Remove access panels ET1 and EB1 Examine the attachment bolts and lock wire for security. Install access panels ET1 and EB1		
13	Elevator balance tab Examine tab, hinges, control attachment points and attaching parts. Lubricate hinges. (Material No. P04-011)		
Chapter 56 - Windows			
1	Windows and windshields Examine.		



















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2	Emergency window Examine.		
Chapter 57 - Wings			
1	Access panels and fasteners Examine.		
2	Wing - external Examine skin and structure, particularly in area of fuel tanks, all access hole and external component or equipment attaching points. Look for loose rivets along the main spar (this can indicate advanced corrosion of the spar cap).		
3	Wing - internal Examine internal skin and structure, particularly in the area of fuel tank, as far as possible. Look for signs of corrosion on the upper and lower main spar caps.		
4	Wings Make sure that the drain holes are not blocked.		
5	Wing struts - external Examine attachment brackets. Examine strut exterior.		
6	Wing struts - internal Examine.		
7	Wing tips Examine.		
8	Aileron support structure Examine aileron support brackets, hinge bearings, control rod attachment points and attaching parts.		
9	Ailerons Examine aileron skin and structure, balance arms and static discharge wicks.		
10	Aileron - balance tabs Examine balance tabs, tab control rods, rod ends, support brackets, hinges and attaching parts. Lubricate hinges (Material No. P04-011).		
11	Flap support structure Examine flap support brackets, hinge bearings, control rod attachments, actuator support bracket and attaching parts.		
12	Flaps Examine structure and skin. Use a mirror and light to examine the skin of the flaps and slats for cracks in the areas where the angles are attached.		
General - Close Up			
NOTE: Do these steps when the engine, electrical and avionic inspections are complete			

NO	TASK	SIGNATURE	
		SIGN	STAMP
1	Access panels and fairings Install.		
2	Fuselage - internal Install internal linings.		
3	Engine cowls Install.		
4	Aircraft Remove the aircraft from jacks.		
5	Aircraft Make sure that the work area is clean and clear of tools and other items.		
C. PROPELLER & ENGINE			
Chapter 61 - Propeller			
1	Spinner dome Remove.		
2	Propeller de-ice boots, slip-ring and brushes Examine.		
3	Slip-ring (Beta) Examine. Check gap between slip-ring and carbon block is no more than 0,50 mm (0.02 in.).		
4	Spinner body and backplate Examine.		
5	Blades Examine.		
6	Spinner Dome Install.		
Chapter 71 - Powerplant			
1	Power Recovery Wash Perform power recovery wash.		
2	Engine Run Perform engine run IAW HARTZELL STC SA377CH.		
3	Engine compartment Clean engine, engine compartment and cowlings.		
4	Engine compartment Examine. Make sure water drain holes are not blocked.		









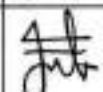







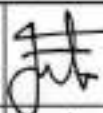

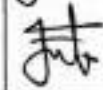

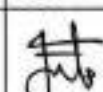

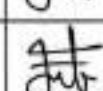

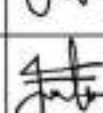

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5	Powerplant and accessories Examine		
6	Powerplant and accessories Inspect and pay particular attention to rear linkage cam box, fuel control unit arm, telescopic rod and rod end fittings. Disconnect rod ends and clean using solvent (PWC11-027) or (PWC11-031). Examine rod end for corrosion, roughness in rotation, side play and radial play. Lubricate with light grease (PWC04-001) or MIL-G-23827 after engine external wash. Reinstall rod ends and torque to specified value. (Ref.76-10-00) Check free movement and linkages.		
7	Powerplant and accessories Air inlet screen - Inspect cleanliness. (Ref.72-20-00) Inspect the air inlet screen wire mesh for cleanliness and/or damage. Screens with broken wire mesh must be replaced. Clean undamaged screens (Ref. Cleaning / Painting). Inspect the rubber sealing rims and flanges of the screen for security and damage.		
8	Powerplant and accessories Gas Generator Case - Inspect External surface, and fire seal mount ring brackets for cracks, distortion and corrosion. (Ref. 72-30-04) Examine for general condition, including cracks, distortion, corrosion and evidence of overheating. Minor corrosion on exposed surface of gas generator case may be removed. (Ref. Approved Repairs). If the condition of the corrosion exhibited on the exposed surfaces of the gas generator case indicates that further examination of the fuel manifold and igniter bosses is required, remove the fuel manifold adapters (Ref. 73-10-05, Removal/Installation) and spark igniters. (Ref. 74-20-00, Removal/Installation). Examine the mounting pads, fuel nozzle bosses and machined surfaces for corrosion and wear. Isolated corrosion pitting not closely grouped, less than 0.010 inch deep, not covering more than 75 percent of the surface is acceptable without repair.		
9	Powerplant and accessories Fireseal Mount Rings - Inspect Cracks and attachment of brackets and seals (Ref. 72-30-01/-02) Examine the rear fireseal mount ring halves for attachment, damage and condition. NOTE: For the external tubes/lines passing through the mount rings, refer to the relevant chapters in this manual. Examine the circumferential insulating strips for attachment. Loosened strips may be rebonded.		



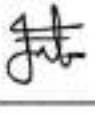







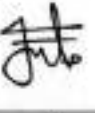





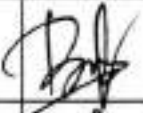



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10	<p>Powerplant and accessories</p> <p>Exhaust Duct - Inspect cracks and distortion. (Ref. 72-50-05, Maintenance practices)</p> <p>Examine the outer surface condition for buckling, ripples or similar distortion. Inspect outer surface, particularly in vicinity of flanges A and C for cracking in metal skin, welds, or flange bolt holes. Inspect exhaust port flanges for cracking.</p> <p>Cracks not exceeding 0.500 inch in length and do not progress into the stitch weld or cracks in a tangential direction not exceeding 1.000 inch long are acceptable provided they are stop drilled with a 1/16 (0.0625) inch drill.</p> <p>Check for the integrity of internal structure through the exhaust ports. NOTE: Refer to the Aircraft Maintenance Manual for removal/installation of the exhaust stubs.</p> <p>Examine the internal structure as far as possible for cracks, looseness and distortion.</p> <p>Inspect Engines that exhibit interior welds (Ref. 72-50-05, Maintenance Practices) visually inspect the forward area of the exhaust duct for cracks, from the propeller reduction gearbox mounting flange to 2 inches aft around the entire circumference of the duct. Exhaust ducts are considered serviceable provided.</p>		
11	<p>Powerplant and accessories</p> <p>Accessories - Inspect attachment of accessories and linkages, air, oil, fuel lines (Ref. 73-10-07/-08) or (Ref. 70-00-00, Standard Practices Inspection).</p> <p>Inspect Fuel, Oil and Air Tubes from scratches, Nick, chafing, dents, pitting, rust and strainer.</p> <p>Inspect Security of pneumatic lines (Ref. 73-10-07/-08)</p> <p>Examine tube assemblies (Ref. 70-00-00, STANDARD PRACTICES - INSPECTION).</p> <p>Blend out minor damage that does not exceed specified limits. Replace the tube assemblies damaged beyond specified limits.</p> <p>Inspect heated rear pneumatic line.</p>		
12	Engine External Examine.		
13	Engine flexible and rigid pipes Examine.		
14	Engine cowlings and seals Examine.		
15	Firescreens and seals Examine.		
16	Shock mounts Examine.		

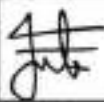

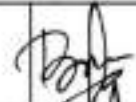

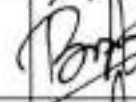





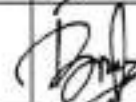

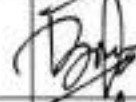

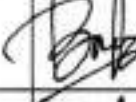

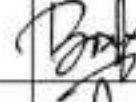

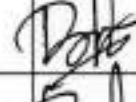

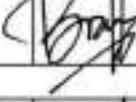

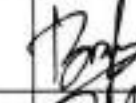

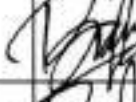



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17	Support ring Examine.		
18	Support struts Examine.		
19	Electrical harnesses Examine.		
Chapter 72 - Engine			
1	Compressor inlet screen Clean, Examine.		
2	Gas generator case Examine.		
3	Propeller shaft oil seal Examine, look for oil leaks		
4	Accessories Examine.		
Chapter 73 - Engine Fuel and Control			
1	HP fuel pump Examine.		
2	HP fuel pump outlet filter Examine, replace if contaminated		
3	Fuel HP Outlet Filter Perform fuel HP outlet filter replacement. P/N : AN6235-3A or ALTERNATIVE P/N. P/N OFF : <u>AN6235-3A</u> P/N ON : <u>AN6235-3D</u>		
4	Fuel control unit Examine Check for leaks from vent. (Ref. P&WC EMM 73-20-00) Check flow divider and dump valve for installation and leaks (Ref. EMM 73-10-06). Check FCU for installation, linkages and pneumatic tubes (Ref. EMM 73-20-00). Evidence of FCU bearing washout indicated by traces of blue dye effluent is caused by a mixture of bearing grease and fuel. For post-SB1472 engines fitted with a manual override on the fuel control, check FCU Manual Override System for static operation (Ref. EMM 71-00-00).		










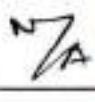

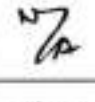

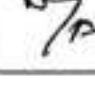



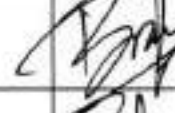



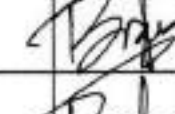

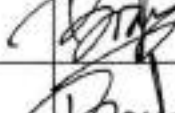

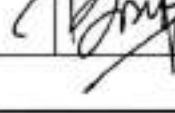

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5	Fuel control unit Perform SIL NO. PT6A-221R01 – FCU Health Monitoring - Deceleration Check. Ref. P&WC PT6A-27 MM 71-00-00		
6	Pneumatic System Check P3 filter for installation. Clean or replace filter, dependent on condition, service experience or environment.		
7	Starting flow control unit Examine.		
8	Propeller governor Examine.		
9	Air pipes Examine.		
10	Fuel pipes Examine.		
11	Gas generator case drain valves Examine.		
12	Igniter exciter Examine and check ignition system/current regulator for installation and condition (Ref.74-10-01 and 74-10-02) Inspect the ignition exciters for signs of damage and general condition. Inspect the input and output connectors for damage, paying particular attention to the connector threads for corrosion. Inspect the cover and box of the regulator for general condition. A cracked or distorted mounting bracket on the box, or loose components on the box or cover, must be repaired at an overhaul facility. Inspect the seal on the box and the sealing gasket on the cover for general condition. A loose seal or gasket may be rebonded using adhesive cement (PWC08-010).		
13	Ignition cables Examine and check ignition cable for chafing, wear and installation (Ref.74-20-01) Inspect cables for signs of damage to braiding and general condition. Inspect cable coupling nuts for corrosion. Inspect central conductor and insulation for contamination and burning. Do retention test on igniter end of cable only: <ul style="list-style-type: none"> - Connect contact with tool (Ref. Table 201). - Contact must hold a 0.125 lb. weight. - If contact does not hold weight, ship cable to an authorized repair shop for inner cable replacement. 		

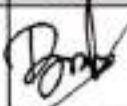

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14	Spark igniters Examine and check spark igniters/glow plugs for cleanliness and erosion. Check function (Ref. 74-20-02 and 74-00-00). Inspect the exterior cylindrical area of the firing end of the igniter shell for chafing wear. Wear is acceptable to a depth of 0.015 inch. Inspect the igniter shell and electrode for erosion (Ref. Fig. 207 and Table 202). If erosion equals or exceeds amounts shown, reject the spark igniter. Do a functional test on acceptable and replacement spark igniters (Ref. 74-00-00, Adjustment/Test).		
15	Interconnect rod Inspect accessible lockwire and safety cable for security and installation of the interconnect rod.		
16	Idle control system Examine.		
17	Power control system Examine.		
18	Propeller control system Examine.		
19	Engine controls Lubricate rod ends with grease. (Material No. P04-002).		
20	Emergency fuel control system Examine. Do a functional test.		
Chapter 78 - Exhaust			
1	Exhaust duct Examine.		
2	Exhaust stubs Examine.		
Chapter 79 - Oil			
1	Oil cooler system Examine. Flap - Do an operational test.		
2	Oil filter Examine and clean.		
3	Chip detector Do a functional test. Check Magnetic Detectors for continuity.		

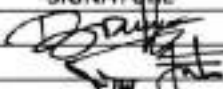


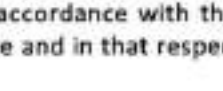
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4	Scavenge Oil pump Examine.		
5	Oil filler cap and dipstick Examine.		
6	Oil separator (Aircraft with SB75) Examine.		
General			
1	Powerplant Make sure that the work area is clean and clear of tools and other items.		
2	Powerplant Do a functional test.		
3	Powerplant (Post P&WC SB 1568 only) Do a deceleration check. NOTE: Not required if FCU is identified with 'RES2' or 'SB 73-3', or with a serial number that has the letter 'F' as a prefix.		
D. ELECTRICS AND INSTRUMENTS			
Chapter 21 - Air Conditioning			
1	Cockpit blower motor Examine and operational test.		
2	Cabin blower motor Examine and operational test.		
Chapter 24 - Electrical Power			
1	Battery mountings Examine attachment fittings, ventilation hoses, cable connectors, wiring.		
2	External power receptacle Examine.		
3	Starter/Generator Examine.		
4	Starter/Generator Examine QAD adaptor and clamp.		
5	Starter and power generation relays. Examine. Functionally test during engine ground run.		

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6	Voltage regulator Examine. Functionally test during engine ground run.		
7	Cockpit - switches and circuit breakers Examine. Make sure that placards are readable.		
8	Cables, plugs, connectors, relays, terminal blocks Examine in these areas: - engine compartment - cockpit - fuselage - empennage - wings		
9	Bonding Examine bonding leads in these areas: - engine compartment - cockpit - fuselage - empennage - wings - landing gear		
Chapter 27 - Flying Controls			
1	Aileron trim actuator Examine. Operational test.		
2	Rudder trim actuator Examine. Operational test.		
3	Flap actuator Examine. Operational test.		
4	Horizontal stabilizer actuator Examine. Operational test.		
Chapter 28 - Fuel			
1	Auxiliary fuel pump Operational test.		
2	Underwing fuel pumps. (if installed) Operational test.		

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Chapter 30 - Ice and Rain Protection			
1	Pitot tube and static port heaters Operational test.		
Chapter 31 - Indicating/Recording			
1	Instrument panel shockmounts Examine.		
2	Instruments Examine.		
3	Annunciator panel Examine.		
Chapter 33 - Lights			
1	Navigation lights Examine. Operational test.		
2	Anti-collision strobe lights or beacons Examine. Operational test.		
3	Landing lights Examine. Operational test.		
4	Cockpit lights Examine. Operational test.		
5	Instrument lights Examine. Operational test.		
6	Warning lights Examine. Operational test.		
7	Passenger cabin lights Examine. Operational test.		
Chapter 34 - Navigation			
1	Pitot tube Examine.		
2	Static ports Examine.		
3	Pipes - pitot, static and vacuum Examine.		


NO	TASK	SIGNATURE	
		SIGN	STAMP
4	Vertical speed indicator Reset to zero.		
5	Airspeed indicator Check, calibrate if necessary.		
6	Gyro operated instruments Operational test.		
7	Magnetic compass Check correction card date validity _____		
Chapter 37 – Vacuum			
1	Vacuum system suction regulator Clean filter.		
2	Vacuum system Examine. Replace if air filter is contaminated.		
3	Vacuum system pressure regulator Examine.		
4	Vacuum system ejector Examine.		
E. AVIONICS			
Chapter 23 - Communications and Chapter 34 - Navigation			
1	Antennas Examine.		
2	Headsets and microphones Clean, Examine.		
3	Avionic equipment Examine.		
4	Avionic connectors and cables Examine.		
5	Avionic equipment racks and shock mounts Examine.		
6	All Avionics systems Examine switches and circuit breakers.		

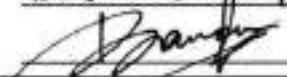
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7	All Avionics systems Operational test.		

PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER
DONALD O.T	ENGINEER		5709
FEBRI HERMANAN	ENGINEER		6445
M. AZIL ANDIKA	ENGINEER		9633
Atya P.	Engineer		9505

RETURN TO SERVICE



















The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : DONALD O.T Stamp : 















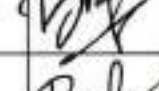

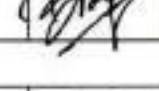

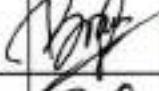

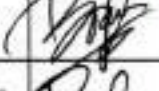

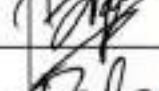





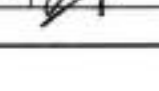

Signature :  Place/Date : WANGEE / 08 / 01 / 2022

Ref. AMM Pilatus Porter Chapter 28-20-00
FUEL DISTRIBUTION SYSTEM – ADJUSTMENT TEST

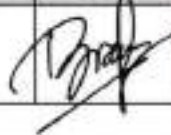

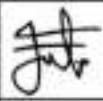

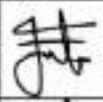

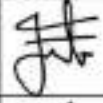

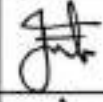

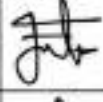

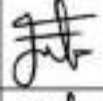

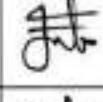

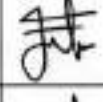

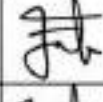

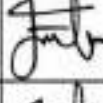

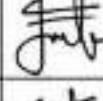

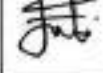

Reg. Mark : PK-SNC Date : 06-JAN-2022
MSN : 1016 Station : NABIRE
TSN / CSN : PG: 46 / 171 WO No. : WO/003-SNC/X11/2021

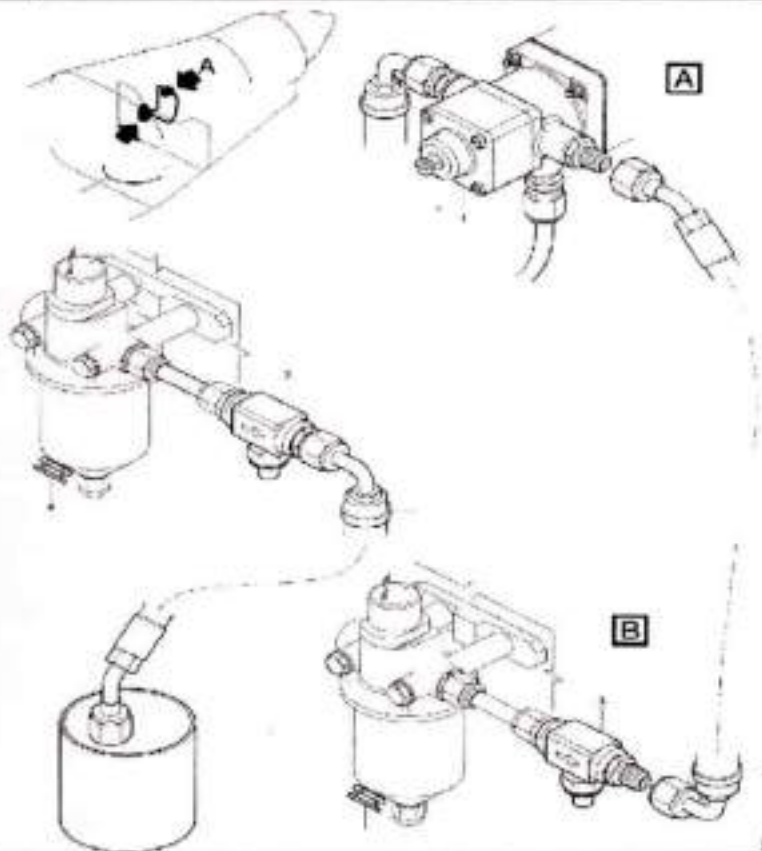
NO	TASK	SIGNATURE	
		SIGN	STAMP
Tools and Equipment			
Part No.	Description	Remarks	
	Stopwatch		
	Fuel container with measured graduation in liters or US gals	Minimum capacity 40 liters (10 US gals)	
Procedure			
A. Job Set Up			
1	Make sure that the aircraft is tail down ($10^{\circ} \pm 1^{\circ}$ nose up).		
2	Make sure that the aircraft is refueled to maximum (Ref. 12-11-28, page Block301).		
3	Set the fuel-system valve lever to CLOSED.		
4	Open the fuel-filter access panel PB3.		
5	Open the engine access panel PL1.		
B. Preparation			
1	Put the fuel container below the fuel filter.		
2	Open the filter drain valve (4) and let the fuel drain.		
3	When the flow of fuel stops, close the drain valve (4).		
4	Remove the outlet hose (2) from between the fuel flow transmitter (3) and the Engine Driven Pump (EDP) (1).		



Appendix – FUEL DISTRIBUTION SYSTEM – ADJUSTMENT TEST


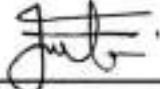
NO	TASK	SIGNATURE	
		SIGN	STAMP
5	Turn the outlet hose through 180 degree, then install the outlet hose (2) to the fuel flow transmitter (3) with the other end through the access panel PB3.		
6	Put the fuel container below the disconnected end of the outlet hose (2). Do not extend the length of the hose for the test.		
7	Set the fuel-system valve-lever to OPEN until you get a constant flow of fuel in to the container.		
8	Set the fuel-system valve-lever to CLOSED and empty the container.		
C. Gravity Flow System			
1	Put the fuel container below the disconnect end of the outlet hose.		
2	Set the fuel-system valve-lever to OPEN for 5 minutes.		
3	Set the fuel-system valve-lever to CLOSED.		
4	Make sure that there is not less than 6,95 liters (1,84 US gals) of fuel in the container. 7,5 liters.		
5	Empty the container.		
D. Auxiliary Fuel Pump System			
1	Put the fuel container below the disconnected end of the outlet hose		
2	Energize the aircraft electrical system.		
3	Set the AUX F PUMP switch to ON and immediately set the fuel-system valve lever to OPEN.		
4	After 5 minutes, set the fuel-system valve lever to CLOSED and the AUX F PUMP switch to OFF.		
5	De-energize the aircraft electrical system.		
6	Make sure that there is not less than 22,1 liters (5,84 US gals) of fuel in the container.		

Appendix – FUEL DISTRIBUTION SYSTEM – ADJUSTMENT TEST

NO	TASK	SIGNATURE	
		SIGN	STAMP
7	Empty the container.		
E. Close Up (Ref. Fig. 501)			
1	Remove the outlet hose (2) from the fuel flow transmitter (3).		
2	Install the outlet hose (2) between the fuel flow transmitter (3) and the EDP (1)		
3	Energize the aircraft electrical system.		
4	Set the fuel-system valve lever to OPEN.		
5	Set the AUX F PUMP switch to ON.		
6	Do leak checks at the outlet hose (2) and the tee adapter (3) connection. No leaks are permitted.		
7	Set the AUX F PUMP to OFF.		
8	Set the fuel-system valve lever to CLOSED.		
9	De-energize the aircraft electrical system.		
10	Discard the fuel as given in the local regulations.		
11	Make sure that the work area is clean and clear of tools and other items.		
12	Close the fuel-filter access panel PB3 and the engine access panel PL1.		

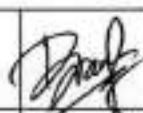

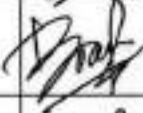

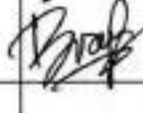

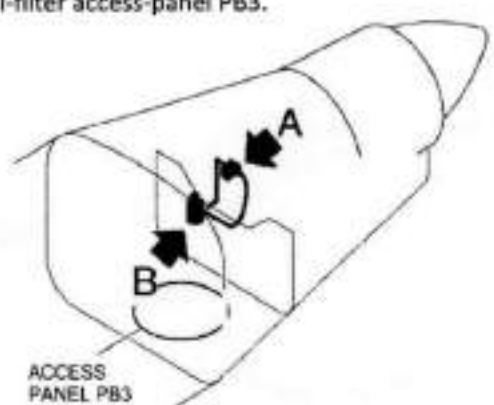






NO	TASK	SIGNATURE	
		SIGN	STAMP
			

PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER
BRANDON ONY T	ENGINEER		5709
FEBRI HERMAWAN	ENGINEER		60445

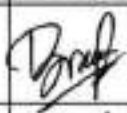

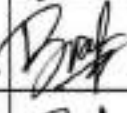



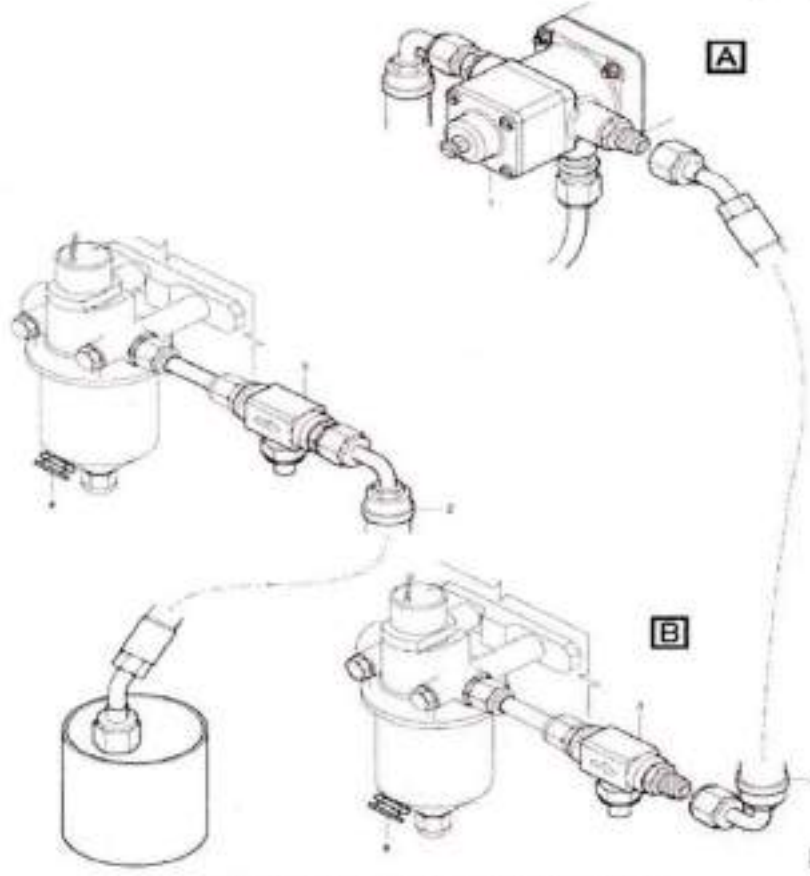


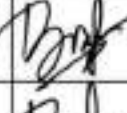

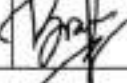

RETURN TO SERVICE			
The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.			
Name	: FEBRI HERMAWAN	Stamp	: 
Signature	: 	Place/Date	: NABIRE / 07-JAN-22

Ref. AMM Pilatus Porter Chapter 28-40-00
FUEL INDICATING SYSTEM – ADJUSTMENT/TEST







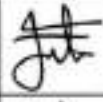

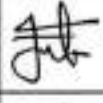

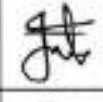

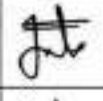

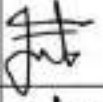

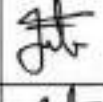

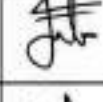

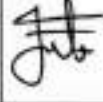

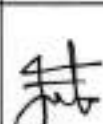

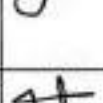

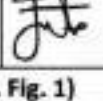

Reg. Mark	: PK - SNC	Date	: 07 JAN 2022
MSN	: 1016	Station	: NABIDE
TSN / CSN	: 196:46 / 171	WO No.	: WD/003-SNC/KH/2021











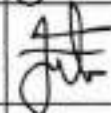

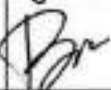





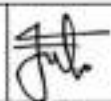

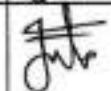

NO	TASK	SIGNATURE										
		SIGN	STAMP									
Tools and Equipment <table border="0"> <tr> <td>Part No.</td> <td>Description</td> <td>Remarks</td> </tr> <tr> <td></td> <td>Stopwatch</td> <td></td> </tr> <tr> <td></td> <td>Fuel container with measured graduation in liters or US gals</td> <td>Minimum capacity 40 liters (10 US gals)</td> </tr> </table>				Part No.	Description	Remarks		Stopwatch			Fuel container with measured graduation in liters or US gals	Minimum capacity 40 liters (10 US gals)
Part No.	Description	Remarks										
	Stopwatch											
	Fuel container with measured graduation in liters or US gals	Minimum capacity 40 liters (10 US gals)										
Procedure												
A. Job Set Up												
1	Make sure that the aircraft is tail down ($10^{\circ} \pm 1^{\circ}$ nose up)											
2	Make sure that the aircraft is refuelled to maximum (Ref. 12-11-28, Page Block 301)											
3	Set the fuel-system valve-lever to CLOSED											
4	Open the fuel-filter access-panel PB3. 											
5	Open the engine access panel PL1											
B. Preparation (Ref. Fig. 1)												
1	Put the fuel container below the fuel filter.											

Appendix – FUEL INDICATING SYSTEM – ADJUSTMENT/TEST























NO	TASK	SIGNATURE	
		SIGN	STAMP
2	Open the filter drain valve (4) and let the fuel drain.		
3	When the flow of fuel stops, close the drain valve (4).		
4	Remove the outlet hose (2) from between the fuel flow transmitter (3) and the EngineDriven Pump (EDP) (1).		
 <p style="text-align: center;">Figure 1 fuel Distribution – Adjusment/Test</p>			
5	Turn the outlet hose through 180 degrees, then install the outlet hose (2) to the fuelflow transmitter (3) with the other end through the access panel PB3.		
6	Put the container below the disconnect end of the outlet hose (2). Do not extend the length of the hose for the test.		
7	Energize the aircraft electrical system.		


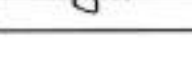
Appendix – FUEL INDICATING SYSTEM – ADJUSTMENT/TEST

NO	TASK	SIGNATURE	
		SIGN	STAMP
8	Set the fuel-system valve-lever to OPEN until you get a constant flow of fuel into the container.		
9	Set the fuel-system valve-lever to CLOSED and empty the container.		
10	Reset the fuel used totalizer to zero.		
C. Low Fuel Flow Indication Check (Ref. Fig. 1)			
1	Set the fuel-system valve-lever to OPEN.		
2	When the fuel flow has stabilized, record the fuel flow indication. Fuel flow: <u>26 GPH</u>		
3	After 5 minutes set the fuel-system valve-lever to CLOSED		
4	Record the fuel used from the fuel used totalizer. Fuel used: <u>2 us Gall</u>		
5	Measure the quantity of the fuel in the container. Quantity: <u>7.5 liter (2 US Gallon)</u>		
6	Make sure the difference between the fuel used totalizer indication and the quantity of fuel in the container is not more than ± 1 liter (± 0.26 US gals).		
7	Calculate the fuel rate for 1 hour as follows: - Quantity of fuel in container after 5 minutes $\times 12 =$ Flow rate/hour.		
8	Compare the calculated fuel flow with the actual fuel flow recorded at step (2): - The difference between the calculated fuel flow and the actual fuel flow must be the same $\pm 12,5$ liter/hour ($\pm 3,3$ gals/hour)		
9	If the difference between the actual and calculated fuel flow is out of limits, replace the indicators, signal conditioner or fuel flow transmitter. NOTE: IT IS RECOMMENDED THAT THE FUEL FLOW TRANSMITTER IS REPLACED FIRST.		
10	Empty the container.		
D. High Fuel Flow Indication Check Using the Auxilliary Fuel Pump System (Ref. Fig. 1)			
1	Set the AUX F PUMP switch to ON.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
2	<p>Move the fuel-system valve-lever towards the CLOSED position to get an indicated fuel flow of between 189 and 285 liters/hour (50 and 75 US gals/hour). Record the indicated fuel flow.</p> <p>Fuel flow: <u>60 GPH (227 liters/hour)</u> <u>74 GPH 280 ltr/hours</u></p> <p>NOTE: TO MAKE THE TEST EASIER, YOU CAN MAKE AN ORIFICE TO GET THE CORRECT FUEL FLOW. USE A METAL BLANK DRILLED WITH A 2,4 mm (3,32 in) HOLE INSTALLED IN THE END OF THE DISCONNECTED OUTLET HOSE (2). THIS WILL GIVE THE REQUIRED FUEL FLOW WITH THE FUEL-SYSTEM VALVE-LEVER FULLY "OPEN".</p>		
3	<p>When the fuel flow is correct and constant, get a second person and do these steps at the same time.</p> <p>Reset the fuel totalizer.</p> <p>Record the quantity of fuel in the container.</p> <p>Start the stopwatch.</p>		
4	After 5 minutes, set the fuel-system valve-lever to CLOSED and the AUX F PUMP switch to OFF.		
5	<p>Record the fuel used from the fuel used totalizer.</p> <p>Fuel flow: <u>6 US Gallon (totalizer) 22.71 ltr.</u></p>		
6	<p>Measure the quantity of the fuel in the container and subtract the quantity recorded in Step (3).</p> <p>Quantity: <u>23.5 liters = 6.2 USG</u> <u>5.28 25.7 USG (23.5 liters)</u></p>		
7	Make sure the difference between the fuel used totalizer indication and the quantity of fuel in the container is not more than ± 1 liter ($\pm 0,26$ US gals).		
8	<p>Calculate and record the flow rate for 1 hour as follows:</p> <p>Quantity of fuel measured at step (6) x 12 = Fuel flow rate /hour.</p>		
9	<p>Make sure that the flow rate recorded at step (8) is not more than $\pm 12,5$ liters/hour ($\pm 3,3$ US gals/hour) different to the fuel flow indication recorded in Step (2).</p> <p><u>74.5 60 GPH</u></p>		
10	<p>If the difference between the actual and calculated fuel flow is out of limits, replace the indicators, signal conditioner or fuel flow transmitter.</p> <p>NOTE: IT IS RECOMMENDED THAT THE FUEL FLOW TRANSMITTER IS REPLACED FIRST.</p>		
E. Close Up (Ref. Fig. 1)			
1	If necessary, remove the orifice from the outlet hose (2).		
2	Remove the outlet hose (2) from the fuel flow transmitter (3).		

Appendix – FUEL INDICATING SYSTEM – ADJUSTMENT/TEST

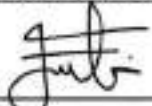
NO	TASK	SIGNATURE	
		SIGN	STAMP
3	Install the outlet hose (2) between the fuel flow transmitter (3) and the EDP (1).		
4	Energize the aircraft electrical system.		
5	Set the fuel-system valve lever to OPEN.		
6	Set the AUX F PUMP switch ON.		
7	Do a leak check at the outlet hose (2) and the tee adapter (3) connection.No leaks are permitted.		
8	Set the AUX F PUMP switch to OFF.		
9	De-energized the aircraft electrical system.		
10	Discard the fuel in accordance with local regulation.		
11	Make sure that the work area is clean and clear of tools and other items.		
12	Close the fuel-filter access panel PB3.		
13	Close the engine access panel PL1.		

PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER
BRUNO OMY.T	ENGINEER		5709
FEBRI HERMANA	ENGINEER		6445

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : FEBRI HERMAWAN Stamp : 









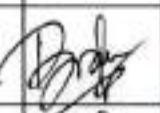





Signature :  Place/Date : NABIRE / 07-JAN-2022







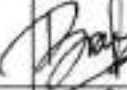

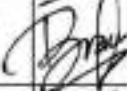

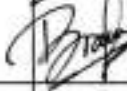

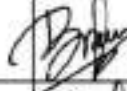

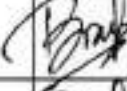

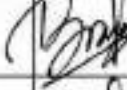

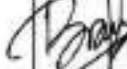

Appendix – FUEL SYSTEM UNDERWING TANK INSPECTION TRANSFER PUMP FILTERS

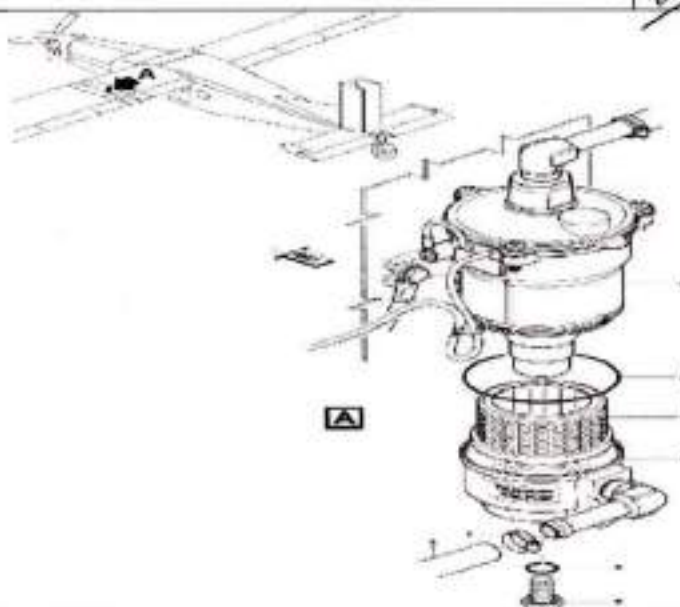
Ref. AMM Pilatus Porter Chapter 28-15-00

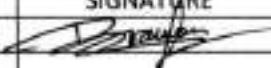
FUEL SYSTEM UNDERWING TANK INSPECTION TRANSFER PUMP FILTERS

Reg. Mark	: PK - SNC	Date	: 06 JAN 2022
MSN	: 1016	Station	: NABIRE
TSN / CSN	: 196 : 46 / 171	WO No.	: WO / 003 - SNC / XU / 2021

NO	TASK	SIGNATURE	
		SIGN	STAMP
Tools Equipment for Aircraft with Underwing Fuel system and Fuel Transfer Pump. P/N: 115.55.06.443 Regulated Air supply			
Expendable Parts			
Part no:	Description	Fig. Item no.	
968.84.30.305	Filter (if required)	Fig 701, item 3	
946.91.27.355	O-ring	Fig 701, item 5	
968.84.30.309	O-ring	Fig 701, item 2	
WARNING: OBEY THE SAFETY PRECAUTIONS GIVEN IN 28-00-00, PAGE BLOCK 201, WHEN YOU DO WORK ON THE FUEL SYSTEM.			
1	Open and install the safety clip to the circuit breaker EXT FUEL .		
2	Remove the Access panel LB7, LB9, RB7 and RB8.		
3	Loosen the clamp (item 7) and disconnect the tube (item 8) from the elbow of the pump inlet.		
4	Remove the bolt (item 6), the bowl (item 4), the O-Ring (item 2) and filter (item 3) from the pump (item 1). WARNING: MAKE SURE YOUR HANDS ARE CLEAN BEFORE YOU CLEAN THE FILTER (3) DO NOT USE COTTON OR CLOTH TO CLEAN THE FILTER COTTON CAN CONTAMINATE THE FILTER.		
5	Use a regulated air supply to blow through from the inside of the filter (item 3) to remove unwanted material.		
6	If the filter (item 3) is damaged or cannot be cleaned, discard the filter and install a new one.		
7	Remove and discard the O-Ring (item 5) from the bolt (item 6) and install a new O-ring (item 5).		

NO	TASK	SIGNATURE	
		SIGN	STAMP
8	Put the filter (item 3), new O-ring (item 2) and bowl (item 4) in position and install the bolt (item 6).		
9	Connect the tube (item 8) to the elbow of the pump outlet and tighten the clamp (item 7).		
10	Close the circuit breaker EXT FUEL.		
11	Energize the aircraft electrical system (Ref. AMM 24-40-00, Page Block 1).		
12	Set the NORMAL-EMERG switch to EMERG and check that the pump operates.		
13	Check the pump for leaks. No leaks are permitted.		
14	Set the NORMAL-EMERG switch to NORMAL.		
15	Remove the electrical power from the aircraft (Ref. AMM 24-40-00, Page Block 1).		
16	Make sure that the work area is clean and clear of tools and other item.		
17	Install the access panel LB7, LB9, RB7 and RB8.		



PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER
BRAMOND ONY.T	ENGINEER		5709

RETURN TO SERVICE



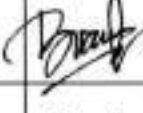





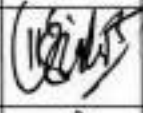

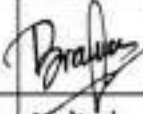

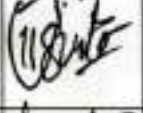

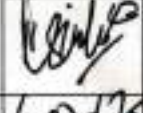

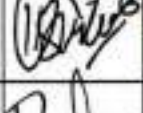



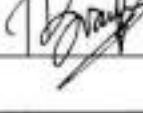

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.







Name : BRAMOND ONY.T Stamp : 

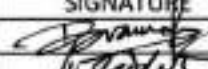
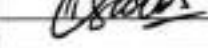
Signature :  Place/Date : WABIRE / 03-01-2022



WHEEL AND BRAKES INSPECTION SHEET OF PILATUS PORTER PC6


Reg. Mark : PK - SNC Date : 08 JAN 2021
 MSN : 1016 Station : NABIRE
 TSN / CSN : 196:46 / 171 WO No. : WO/003-SNC/x11/2021

NO	TASK	SIGNATURE	
		SIGN	STAMP
1	Perform Detail Visual Inspection with Flash Light, Mirror, and Magnifying Glass of the Brake Pedals and System for Cracks, Corrosion, and Security of Installation.		
2	Inspect Wheel and Brakes IAW ATA 32-40-00 and BERINGER Time Limits /Maintenance Checks MC-STC-002.		
3	Inspect hydraulic brake fluid reservoir, check brake fluid level, apply brakes, examine system for leaks, and service with MIL-PRF-5606 (ROYCO 756) hydraulic fluid as required.		
4	Inspect tire condition IAW ATA 12-14-32 and Michelin Aircraft Tire Care and Service Manual (Michelin Service Manual can be used as a guide line for all approved main tires but will not supersede manufacture inspection recommendations)		
5	Check Brake Disc Thickness Record $6.95 \frac{FH}{10} 0.271$ mm/inch. $6.96 \frac{LH}{10} 0.274$ Minimum brake disc thickness 0.252 inch / 6.4 mm.		
6	Examine brake disc condition for Coning, Groove and Bumps. See figure 2 as attached.		
7	Inspect Brake Pad for wear. Brake Pad must be changed before grooves are invisible. See figure 3 as attached. Friction material on Brake Pad minimum thickness 0.100 inch / 2.5 mm. $5.7 \frac{mm}{10} / 5.5 \frac{mm}{10}$		
8	Check play between disc and key disc drive. Max play 0.024 inch / 0.6 mm. See figure 3 as attached.		
9	Check Main wheels Tire. Examine and check inflation pressure 3,3 bar (47 psi). <u>20 PSI</u>		
10	Check Tail wheel. Examine and check for installation and inflation pressure 2,2bar (47 psi).		
11	Check and examine brake master cylinders for leaks and connections.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
12	After complete installation, check disc safety wire. Safety wire (0.041) must be in place to prevent disc from sliding out the slots. See figure 1 as attached.		
13	Record both Main Wheel Tire: S/N LH <u>19049269</u> and Hub S/N <u>492</u> S/N RH <u>19025074</u> and Hub S/N <u>493</u>		
14	Record Tail Wheel Tire S/N <u>83432003</u> Hub S/N <u>277</u>		

PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER
BRAMONO DNY.T	ENGINEER		5709
AGUS SUATMAN	ENGINEER		6444

RETURN TO SERVICE			
The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.			
Name	: <u>BRAMONO DNY.T</u>	Stamp	: 
Signature	: 	Place/Date	: <u>WABUP / 08 JAN 2022</u>

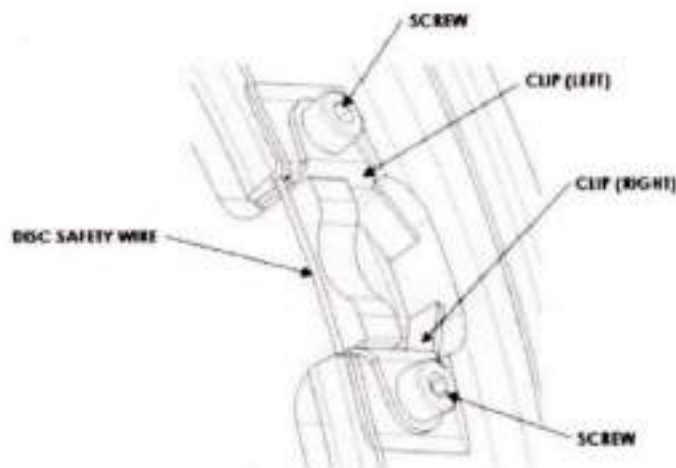
 Aéroports, 05130 TALLARD - FRANCE Tel: +33 (0)4 92 20 16 19 Fax: +33 (0)4 92 52 69 66 e-mail: contact@beringer-aero.com	TIME LIMITS / MAINTENANCE CHECKS	Manual reference: BRG-ALTP-02 Reference document: MC-STC-002
--------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------	-------------------------------------------------	-------------------------------------------------------------------------------------

2. Scheduled maintenance checks

2.1. Flight maintenance checks

Next flight maintenance checks are in addition to PC-6 maintenance manual.

Additional flight maintenance checks	Preflight inspection
Component	Operation
Safety wire of brake disc	Visual inspection
Brake pads	Inspect for wear and damage



CAUTION: Disc safety wire must be in place, it prevents disc from sliding out the slots.

FIGURE 1

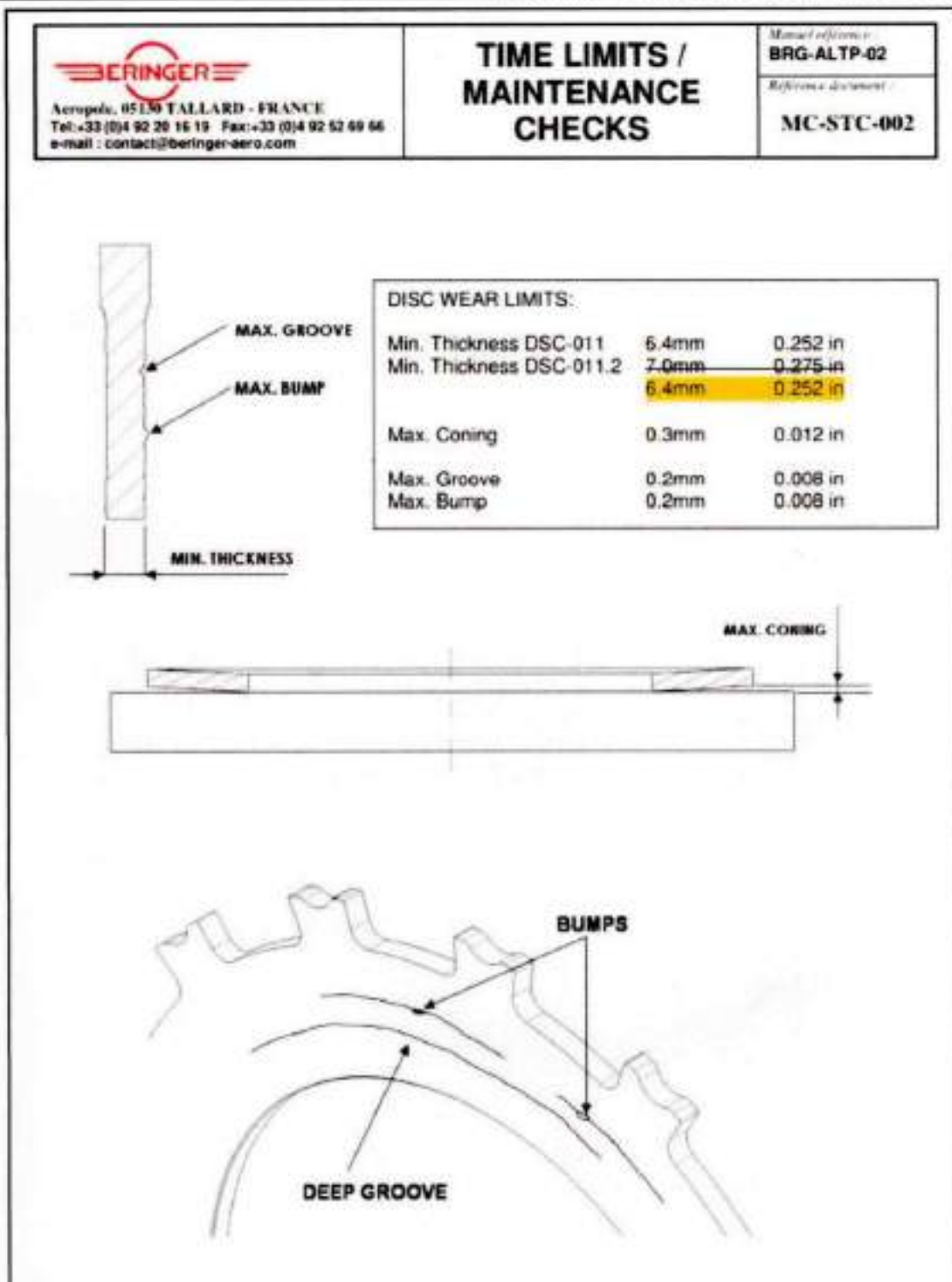

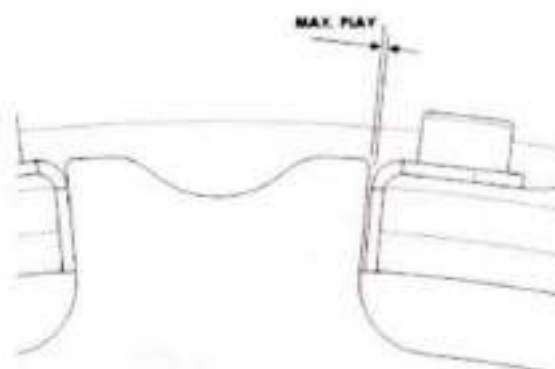


FIGURE 2

 Acropole, 95130 TALLARD - FRANCE Tel: +33 (0)4 92 20 16 19 Fax: +33 (0)4 92 52 69 66 e-mail: contact@beringer-aero.com	TIME LIMITS / MAINTENANCE CHECKS	Manual reference : BRG-ALTP-02 Reference document : MC-STC-002
-------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------	-------------------------------------------------	---------------------------------------------------------------------------------------

CLIP WEAR LIMITS:

Max. Play 0.6mm 0.024 in



PAD WEAR LIMITS:

Min. Thickness groove nearly invisible
 Friction material min. thickness 2.5mm (0.100 in)

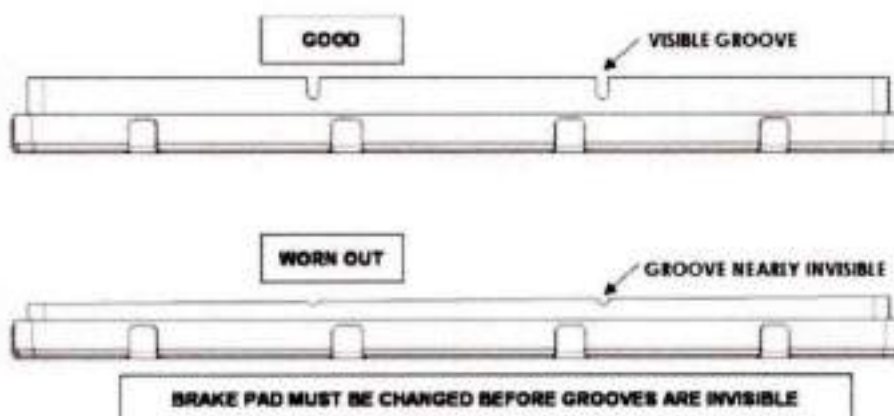


FIGURE 3

7

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/003-SNC/XII/2021	22-Dec-2021	CHECK FOR WEAR	PK-SNC/1016
5. CARD #	6. ATA SPEC	7. TRADE	8. STA
001	24		
9. ZONE	10. PANEL	-	

11. DESCRIPTION

PERFORM CHECK FOR WEAR STARTER GENERATOR BRUSHES

REFERENCE

☒ AMM Ch. 24-31-11, Page block 601

☐

☐ OTHER

R/II (*)

☐ Y

☒ N

MHR :

12. RESULT

FOUND BRUSHES WEAR STILL WITHIN LIMIT
APPROX 85% LIFE LIMIT.
REINSTALLED AND SECURED AS REQUIRED.

Performed at A/C TT : 196:46

A/C TC /LDG : 171

MECH

ENG

INSP (*)

7A



7A

FINDING

☐ Y

☒ N

ACT MHR : 2 MHR

DATE/TIME
(DD/MM/YY)

INSPECTION CARD (IC) #

07

01

2022

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS
7A	7A	-	-	-

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS
7A	7A	-	-

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/003-SNC/X/2021	22-Dec-2021	CHECK OPERATION	PK-SNC/1016
5. CARD #	6. ATA SPEC	7. TRADE	8. STA
002	71		
9. ZONE	10. PANEL		

11. DESCRIPTION

PERFORM CHECK OPERATION OF PROPELLER OVERSPEED GOVERNOR

REFERENCE

☒ AMM Ch. 71-00-00

☐

☐ OTHER

R/I (*)

☐ Y

☒ N

MHR :

12. RESULT

CARRIED OUT DURING ENGINE RUN UP
FOUND INDICATION OVERSPEED GOVERNING
PROPELLER AT 1880 RPM.

Performed at A/C TT : 196:46 A/C TC /LDG : 171

MECH

ENG

INSP (*)

N/A



N/A

FINDING

☐ Y

☒ N

ACT MHR : 1 MHR

DATE/TIME
(DD/MM/YY)

INSPECTION CARD (IC) #

08

01

2022

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS
N/A	N/A	-	-	-

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS
N/A	N/A	-	-



#1




Parts Used Sheet

[illegible]



EMERGENCY EQUIPMENT LIST INSPECTION & MONITOR

**PT. SMART CAKRAWALA
AVIATION
DEPARTMENT TEKNIK
Form: SCA/MTC/023**

DATE : 08 JAN 2022	A/C REG : PK-SNC
A/C TYPE : PC B / B2-H4 PIAZUS PORTER	CHECKER : BRAMONO ST SIGN: 

No.	Description	P/N	S/N	Next Insp.	Remarks
1	Pilot Life Vest	P0107A-101	12062000022	OCT 2022	SERVICEABLE
2	Co-Pilot Life Vest	P0107A-101W	17060500010	OCT 2022	SERVICEABLE
3	Pax Life Vest	P0107A-101	12062000025	OCT 2022	SERVICEABLE
4	Pax Life Vest	P10107A-101	12062000034	OCT 2022	SERVICEABLE
5	Pax Life Vest	P10174-101W	17060500025	OCT 2022	SERVICEABLE
6	Pax Life Vest	63600-167	L157063	OCT 2022	SERVICEABLE
7	Pax Life Vest	63600-167	L273003	OCT 2022	SERVICEABLE
8	Pax Life Vest	D 21343-195	L849507	OCT 2022	SERVICEABLE
9	Pax Life Vest	S 21250-6300	42218-1161	OCT 2022	SERVICEABLE
10	Pax Life Vest				
11	Pax Life Vest				
12	Pax Life Vest				
13	First Aid Kit	FAK /ANC 1	LJ 1108200	AUG 2023	SERVICEABLE
14	Crash Axe Installed	BIN 14924 A	AS 53	O/C	SERVICEABLE
15	Fire Extinguisher	BA51015R-5	119254	NOV 2022	SERVICEABLE
16	Life Raft (If Installed)				
17	Survival Kit (If Installed)	10001-1	NSN	APRIL 2024	SERVICEABLE
OTHERS					