Appendix – FUEL SYSTEM UNDERWING TANK INSPECTION TRANSFER PUMP FILTERS

Ref. AMM Pilatus Porter Chapter 28-15-00

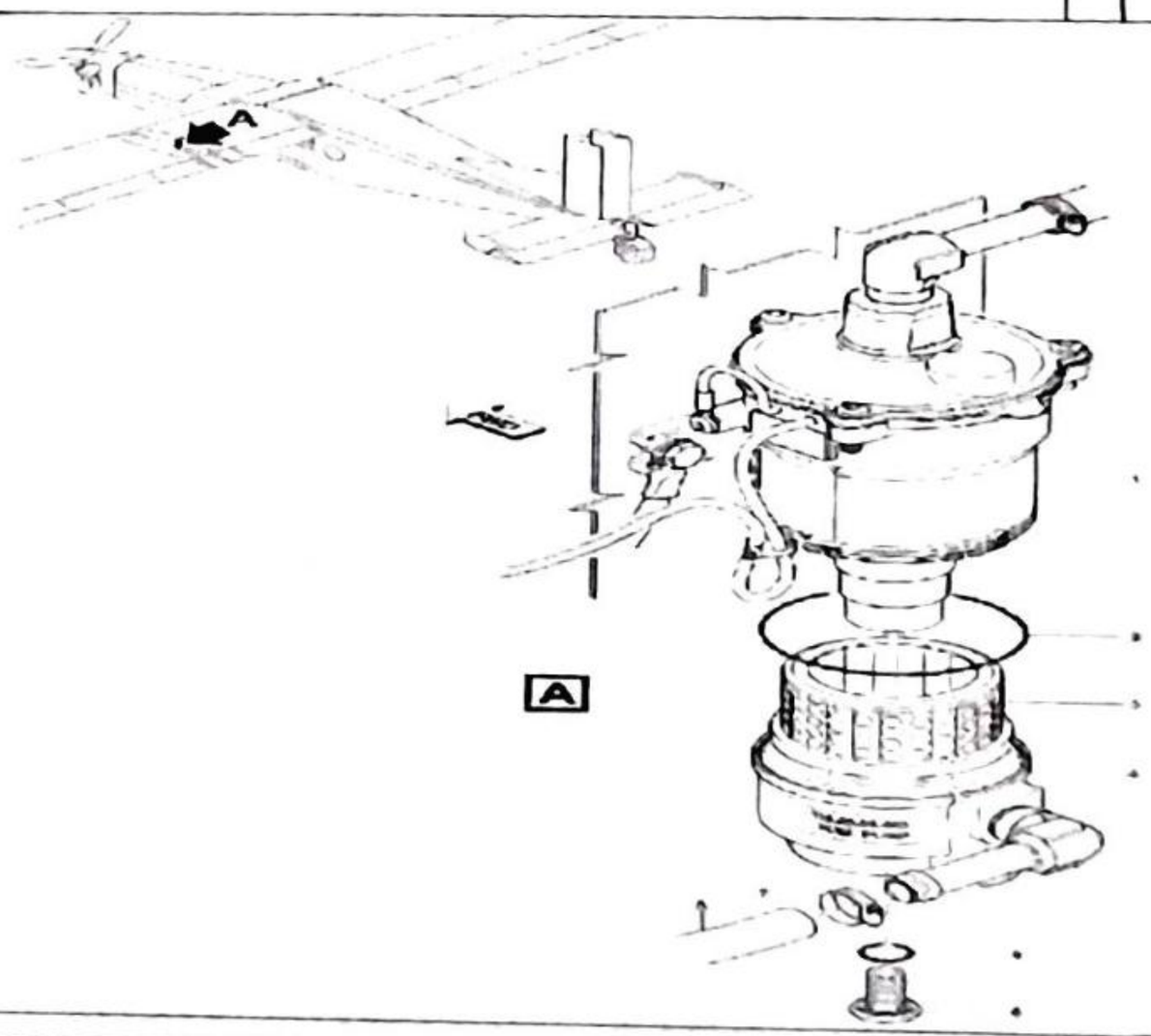
FUEL SYSTEM UNDERWING TANK INSPECTION TRANSFER PUMP FILTERS

Reg. Mark	: PK - SNB	Date	: 11/01/2022
MSN	: 1015	Station	: BEJ
TSN / CSN	: ~~4097~~ 409:00 /4164	WO No.	: WO/006-SNB/XII/2021

NO	TASK	SIGNATURE	
		SIGN	STAMP
Tools Equipment for Aircraft with Underwing Fuel system and Fuel Transfer Pump. P/N: 115.55.06.443 Regulated Air supply			
Expendable Parts **Part no:** / **Description** / **Fig. Item no.** 968.84.30.305 — Filter (if required) — Fig 701, item 3 946.91.27.355 — O-ring — Fig 701, item 5 968.84.30.309 — O-ring — Fig 701, item 2			
WARNING: OBEY THE SAFETY PRECAUTIONS GIVEN IN 28-00-00, PAGE BLOCK 201, WHEN YOU DO WORK ON THE FUEL SYSTEM.			
1	Open and install the safety clip to the circuit breaker **EXT FUEL**.		29
2	Remove the Access panel LB7, LB9, RB7 and RB8.		29
3	Loosen the clamp (item 7) and disconnect the tube (item 8) from the elbow of thepump inlet.		29
4	Remove the bolt (item 6), the bowl (item 4), the O-Ring (item 2) and filter (item 3)from the pump (item 1). **WARNING**: MAKE SURE YOUR HANDS ARE CLEAN BEFORE YOU CLEAN THE FILTER (3)DO NOT USE COTTON OR CLOTH TO CLEAN THE FILTER COTTON CAN CONTAMINATE THE FILTER.		29
5	Use a regulated air supply to blow through from the inside of the filter (item 3) toremove unwanted material.		29
6	If the filter (item 3) is damaged or cannot be cleaned, discard the filter and install anew one.		29
7	Remove and discard the O-Ring (item 5) from the bolt (item 6) and install a new O-ring (item 5).		29

Appendix – FUEL SYSTEM UNDERWING TANK INSPECTION TRANSFER PUMP FILTERS

NO	TASK	SIGNATURE	
		SIGN	STAMP
8	Put the filter (item 3), new O-ring (item 2) and bowl (item 4) in position and installthe bolt (item 6).		
9	Connect the tube (item 8) to the elbow of the pump outlet and tighten the clamp(item 7).		
10	Close the circuit breaker EXT FUEL.		
11	Energizer the aircraft electrical system (Ref. AMM 24-40-00, Page Block 1).		
12	Set the NORMAL-EMERG switch to EMERG and check that the pump operates.		
13	Check the pump for leaks. No leaks are permitted.		
14	Set the NORMAL-EMERG switch to NORMAL.		
15	Remove the electrical power from the aircraft (Ref. AMM 24-40-00, Page Block 1).		
16	Make sure that the work area is clean and clear of tools and other item.		
17	Install the access panel LB7, LB9, RB7 and RB8.		



PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER
ARIS KURNIAWAN	ENGINEER		3523
ROHPININDO. N S P	ENGINEER		9634
BASRI	MECHANIC		

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : ARIS KURNIAWAN Stamp : 29

Signature : _____ Place/Date : BERAU / 11/01/2022

WHEEL AND BRAKES INSPECTION SHEET OF PILATUS PORTER PC6

Reg. Mark	:	PK - SNB	Date	:	10/11/2024
MSN	:	1015	Station	:	BE1
TSN / CSN	:	409:00 / 464	WO No.	:	WO/006-SNB/XII/2021

NO	TASK	SIGNATURE	
		SIGN	STAMP
1	Perform Detail Visual Inspection with Flash Light, Mirror, and Magnifying Glass ofthe Brake Pedals and System for Cracks, Corrosion, and Security of Installation.		
2	Inspect Wheel and Brakes IAW ATA 32-40-00 and BERINGER Time Limits /Maintenance Checks MC-STC-002.		
3	Inspect hydraulic brake fluid reservoir, check brake fluid level, apply brakes,examine system for leaks, and service with MIL-PRF-5606 (ROYCO 756) hydraulic fluid as required.		
4	Inspect tire condition IAW ATA 12-14-32 and Michelin Aircraft Tire Care and Service Manual (Michelin Service Manual can be used as a guide line for all approved main tires but will not supersede manufacture inspection recommendations)		
5	Check Brake Disc Thickness Record R/H : 6.96 / L/H : 6.90 mm/inch. Minimum brake disc thickness **0.252 inch / 6.4 mm.**		
6	Examine brake disc condition for Coning, Groove and Bumps. See figure 2 asattached.		
7	Inspect Brake Pad for wear. Brake Pad must be changed before grooves are invisible. See figure 3 as attached. Friction material on Brake Pad minimumthickness **0.100 inch / 2.5 mm.**		
8	Check play between disc and key disc drive. Max play **0.024 inch / 0.6 mm**. Seefigure 3 as attached.		
9	Check Main wheels Tire. Examine and check inflation pressure 3,3 bar (~~49~~ 20 **psi**).		
10	Check Tail wheel. Examine and check for installation and inflation pressure 2,2bar (**47 psi**).		
11	Check and examine brake master cylinders for leaks and connections.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
12	After complete installation, check disc safety wire. Safety wire (**0.041**) must be inplace to prevent disc from sliding out the slots. See figure 1 as attached.		29
13	Record both Main Wheel Tire: S/N LH 19049268 and Hub S/N 489 S/N RH 19050098 and Hub S/N 490		29
14	Record Tail Wheel Tire S/N 12952559 Hub S/N 276		29

PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER
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ROHPINDO. N.S.P	ENGINEER		9634
BASRI	MECHANIC		

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Name : ARIS KURNIAWAN Stamp : 29

Signature : Place/Date : 11/01/2022

BERINGER
Aeropole 05130 TALLARD - FRANCE
Tel: +33 (0)4 92 20 16 19 Fax: +33 (0)4 92 52 69 66
e-mail : contact@beringer-aero.com

TIME LIMITS / MAINTENANCE CHECKS

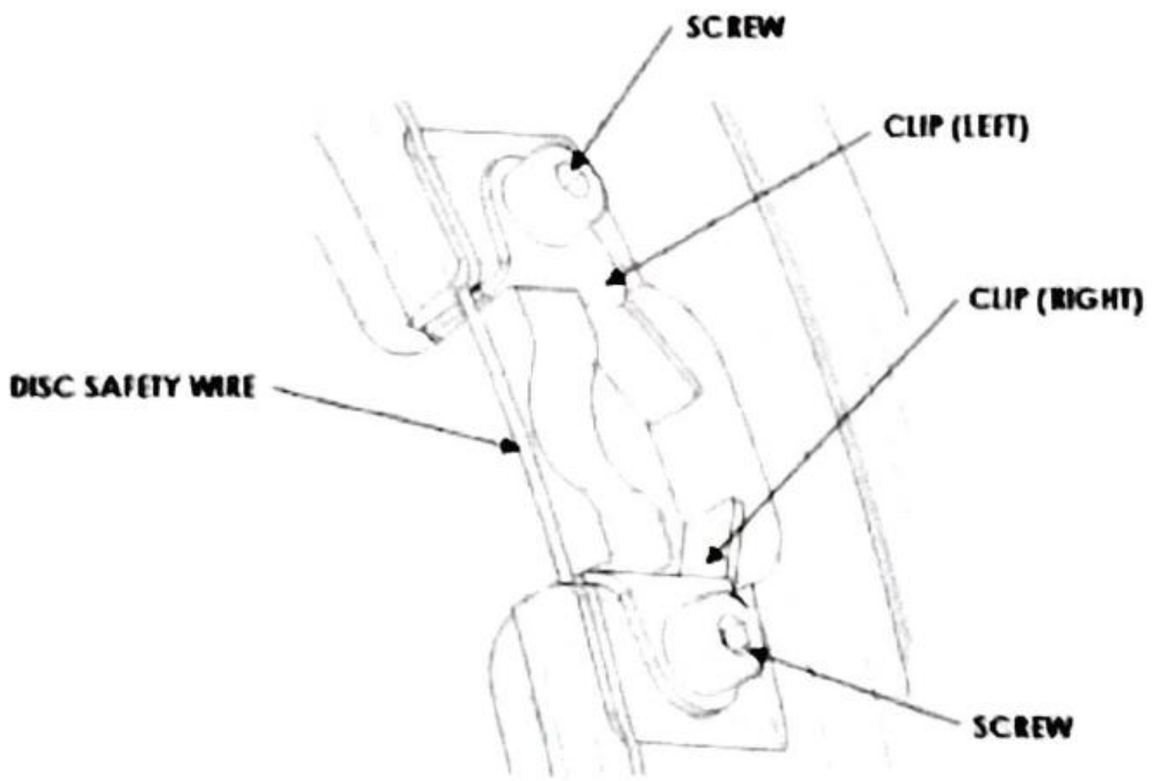
Number: BRG-ALTP-02
Reference: MC-STC-002

2. Scheduled maintenance checks

2.1. Flight maintenance checks

Next flight maintenance checks are in addition to PC-6 maintenance manual.

Additional flight maintenance checks	**Preflight inspection**
Component	**Operation**
Safety wire of brake disc	Visual inspection
Brake pads	Inspect for wear and damage



CAUTION: Disc safety wire must be in place, it prevents disc from sliding out the slots.

FIGURE 1

BERINGER
Aeropole, 05130 TALLARD - FRANCE
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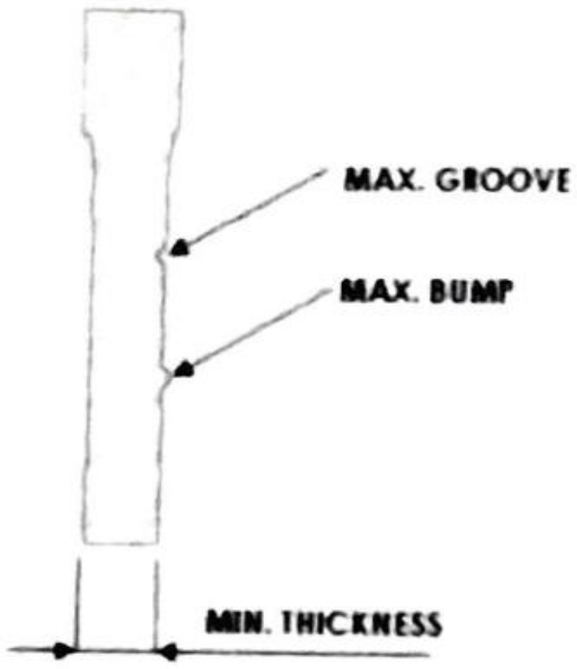
e-mail : contact@beringer-aero.com

TIME LIMITS / MAINTENANCE CHECKS

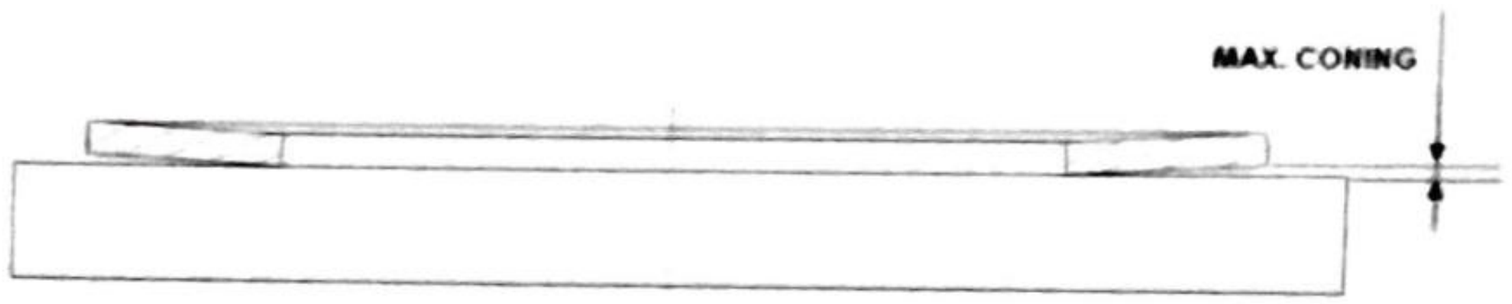
BRG-ALTP-02

MC-STC-002



DISC WEAR LIMITS:

Min. Thickness DSC-011	6.4mm	0.252 in
Min. Thickness DSC-011.2	~~7.0mm~~	~~0.275 in~~
	6.4mm	0.252 in
Max. Coning	0.3mm	0.012 in
Max. Groove	0.2mm	0.008 in
Max. Bump	0.2mm	0.008 in



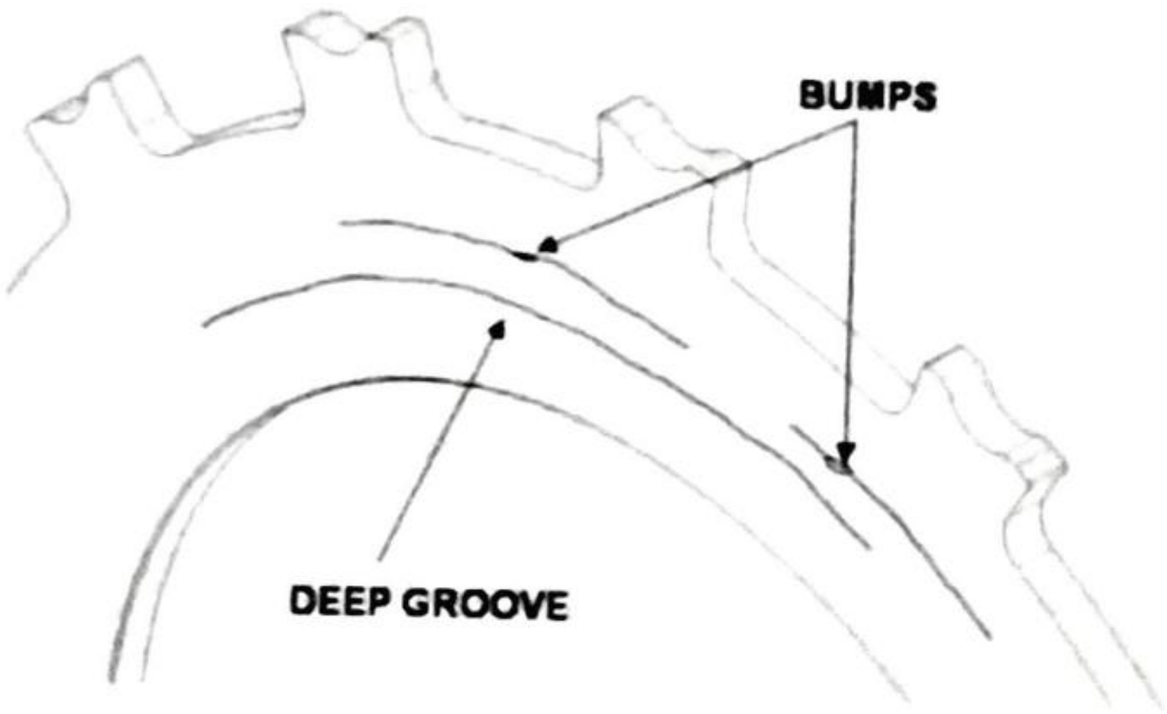


FIGURE 2

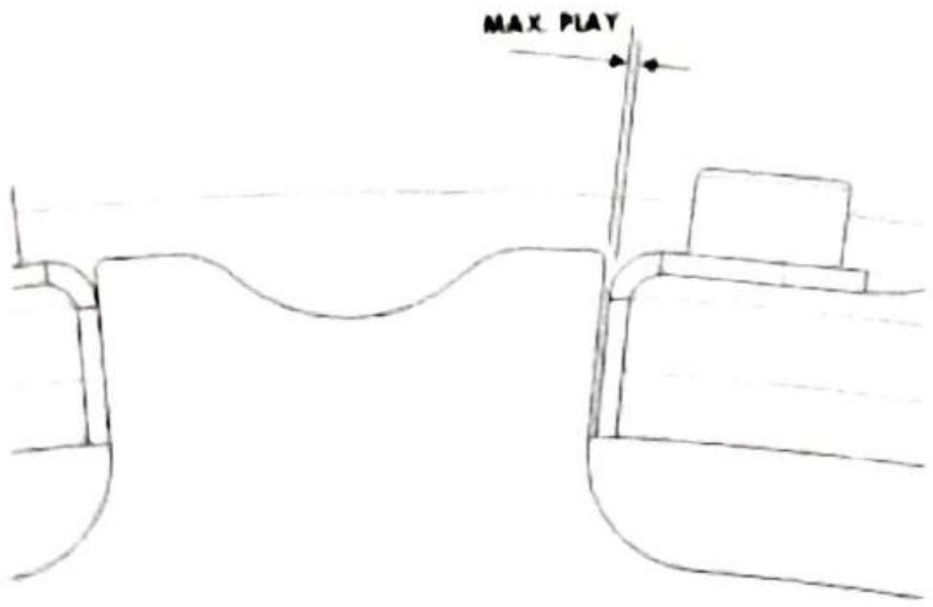
BERINGER
Aeropole, 05130 TALLARD - FRANCE
Tel: +33 (0)4 92 20 16 19 Fax: +33 (0)4 92 52 69 66
e-mail contact@beringer-aero.com

TIME LIMITS / MAINTENANCE CHECKS

Manual reference: BRG-ALTP-02
Reference document: MC-STC-002

CLIP WEAR LIMITS:

Max. Play	0.6mm	0.024 in



PAD WEAR LIMITS:

Min. Thickness	groove nearly invisible Friction material min. thickness 2.5mm (0.100 in)

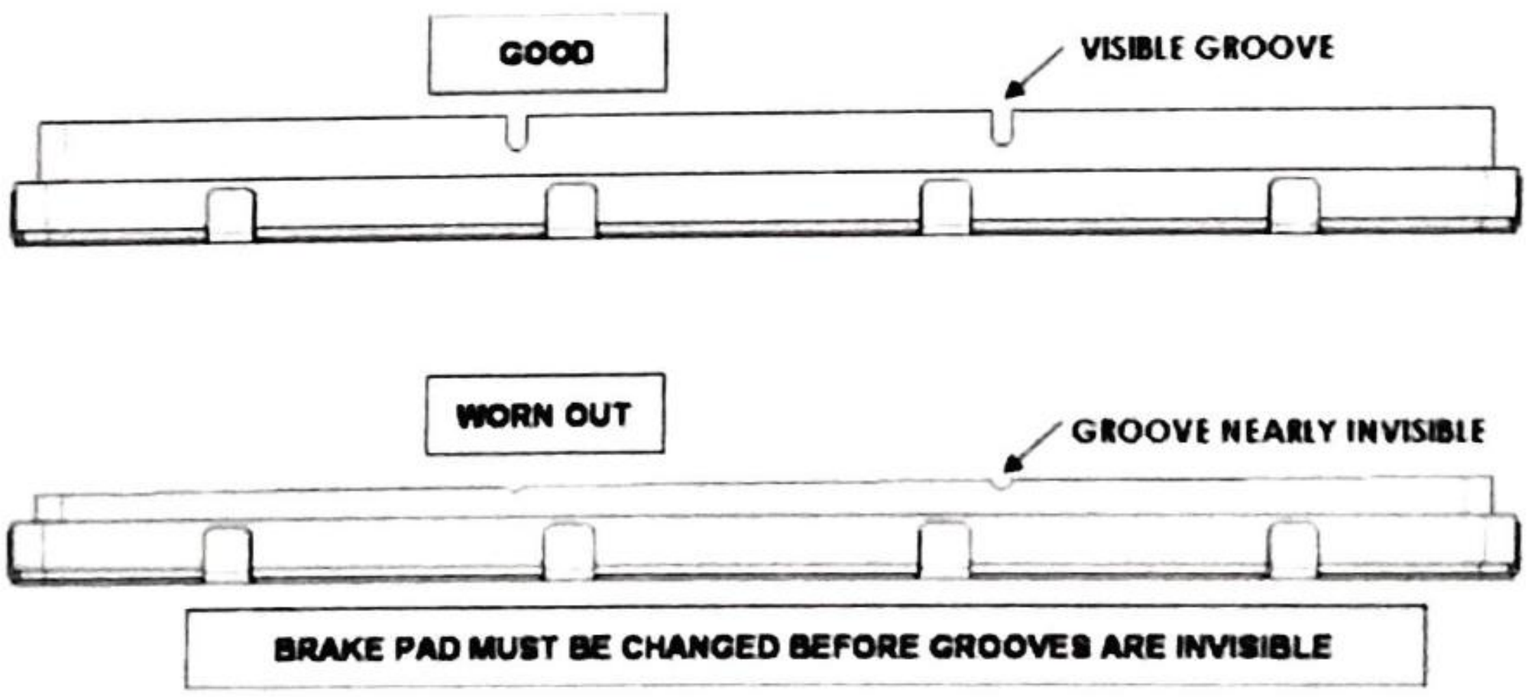


FIGURE 3

Instructions (Note Post SB 1372 Engines Sheet).
Each Listed Inspection Item is to be performed in accordance with the P & W Maintenance Manual 3013242 Chapter 73-10-05 Latest Revision and any other applicable publications

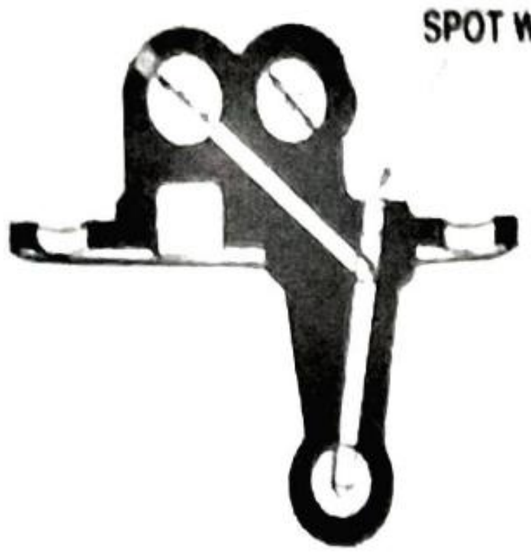
FUEL NOZZLE CHANGE WORK SHEET PT6A-27

Date Performed :	11-JAN-2022	**Engine S/N** :	PCE-P40565
Removed from Aircraft :	11-JAN-2022	**Engine TSN** :	409:60
Aircraft Total Hours :	409:00	**Engine TSO** :	409:00
Aircraft Total Cycle :	464	**Work Order Number** :	WO/006-SAB/XII/2021

NO	TASK	SIGNATURE	
		SIGN	STAMP
Procedure Before Removal			
1	To ease accessibility to the transfer tubes and manifold adapters adjacent to the spark igniters, disconnect the ignition leads (ref CMM 74-20-00/74-20-01). Release the ignition lead loop clamps from the support brackets at the center fire seal lower attachment brackets and move the leads clear. Install blanking caps on the spark igniters and lead connectors.		29
2	Disconnect both lines from the fuel inlet adapter or flow divider and install blanking caps.		29
Removal of Fuel Manifold Adapters			
	Note: The following procedure itemizes a removal sequence commencing with the No 8 fuel manifold inlet adapter which is best achieved by consideration of the Nos. 7, 8 and 9 adapters as a group. The procedural sequence may be modified by the operator as convenient for adapters at other locations.		
1	Using a suitable dye maker (PWC05-027) or (PWC05-046), number the position of each manifold adapter to identify its original position. (Refer to Figure 201) and toaid detecting hot section damage		29
2	Remove bolts securing transfer tube locking plate (6, Fig 202) and inlet manifoldadapter (item 2) to gas generator case. Remove locking plate (item 6).		29
3	Remove bolts securing locking plates (item 6) to the primary and secondary manifold adapters (item 5) adjacent to the inlet manifold adapter (item 2). Remove locking plates (item 6).		29
4	Hold all the three adapters, then move the interconnecting fuel transfer tubes (item 1) into adapter bores (item 5). Use the puller (PWC54246) or pusher (PWC32366)to move the fuel transfer tubes in a clockwise direction, away from the inlet manifold adapter bores.		29
5	Remove the inlet manifold adapter (item 2) (with flow divider and dump valve (item 14) installed). Use the pusher (PWC32366) or puller (PWC54246) to remove thefuel transfer tubes (item 1) from the adjacent adapters (item 5). Remove and discardpreformed packings (item 13) from the transfer tubes.		29

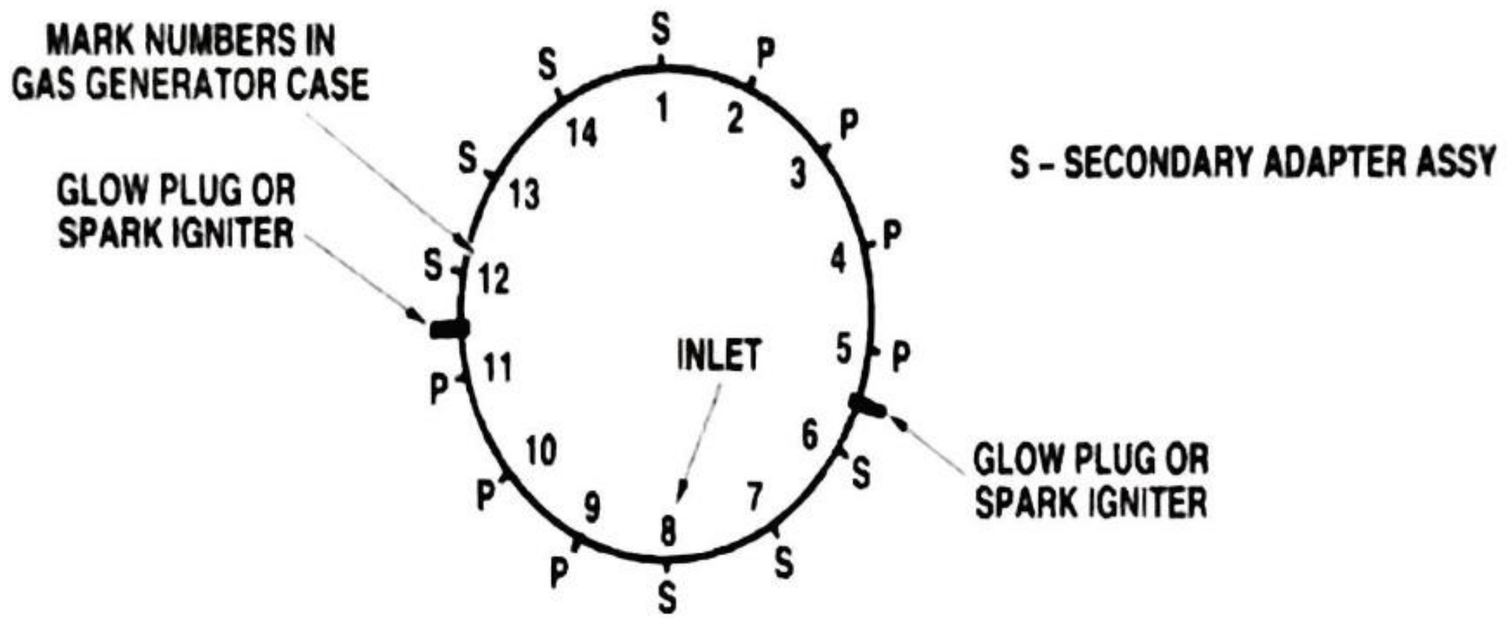
NO	TASK	SIGNATURE	
		SIGN	STAMP
1	With a 10X magnifying glass verify that each manifold adapter assembly carries thecorrect detail fuel nozzle assembly tip part number		29
2	Install the elbows (Item 9) on the inlet adapter (starting control installation only) inthe same positions as noted on removal		29
3	Install the fuel nozzle (Item 8, Fig 202) in the fuel manifold adapters (Item 2 and 5)with new keywasher (item 7) at each location		29
	Use engine oil (PWC03-001) and torque the nozzle assemblies 45 to lbf. In.		29
	NOTE: Leak test and function test of each Nozzle and Adapter Assembly may perform OFF WING in House or approvedoverhaul/repair vendor as required. A company serviceable tag with a copy of the Approved Parts Tag FAA Form 8130-3, EASA formone or equivalent for new parts.		
4	**CAUTION:** DO NOT USE SHARP EDGE TOOLS TO BEND OR SET KEYWASHER TABS.		
	On completion of installation lock each key washer on respective nozzle assembly. Do not exceed specified torque to align flat on tip with key washer.		29
5	Install the manifold adapters and fuel transfer tubes as follows.		29
	NOTE: Primary fuel manifold adapters are identified by a single weld blob on the larger mounting flange. Other weld blobs appearing on the knuckle section of the adapters should be ignored. (SeeFig 201)		
	a) Assemble Sheaths (Item 3) on all adapters (Items 2 & Item 5). Make sure eachlocating pin engages hole in each sheath.		29
	b) With Nozzle adapter and sheath pressed together by hand, check clearance between adapter and sheath flanges. Maximum gap allowed is 0.003 inch (Ref. Fig 203 upper drawing) Larger gap suggests either or both parts are distorted. Iffound Send distorted parts to parts P &W for repair.		29
	c) Carefully check gap between fuel nozzle tip and side hole in sheath; clearance of 0.020 inch is required (Ref. Fig 203). If clearance is less at any point, either orboth parts are distorted. If found Send distorted parts to parts Pratt and Whitney for repair.		29
	d) Lubricate and install preformed packings (13, Fig. 202) on all fuel transfer tubes (1) using a thin layer of engine oil (PWC03-001). Position fuel transfer tubes intoports on one side of manifold adapters (2 and 5). Fully insert fuel transfer tubes in their respective ports until the bottom of manifold adapter is reached.		29

NO	TASK	SIGNATURE	
		SIGN	STAMP
	CAUTION: TO AVOID POSSIBLE OF THE STAINLESS STEEL GASKETS, IT IS ADVISABLE TO SLIDE THE GASKET OVER THE SHEATH, CAREFULLY ALIGN THE HOLES AND INSERT BOTH BOLTS, THE WHOLE ASSEMBLY CAN THEN BE MATED WITH THE PADS ON THE GAS GENERATOR CASE TUS AVOIDING ANY ATTEMPT TO LEVER THE GASKETS INTO ALIGNMENT WITH THE BOLTS.		
	e) Position the Pre SB 1276/ Post SB 1276 gasket (Item 4) over the sheath (Item3) on the inlet manifold adapter (Item 2) and align the bolt holes or mounting studs. The sheath flange must be flat. NOTE: The gasket may be put on either side. For consistency, all the gasketsshould be installed with the flat side against the gas generator. NOTE: **Post-SB1167**: For engines with the conversion coated gas generatorcase only, lightly coat both faces of the gasket (4) with corrosion- preventive compound (PWC09-003).		29
	f) Position the gaskets (4) over the sheaths (3) on the primary and secondarymanifold adapters (5) and align on the bolt holes.		29
	g) Start with each side of the inlet manifold adapter (2). Install the remaining manifold adapters (5) and sheaths (3) on the gas generator case at the locations specified before. Use the pusher (PWC32366) or puller (PWC54246)to engage fuel transfer tubes (1) to interconnect with the adjacent adapter at each location.		29
	h) Install locking plate (Item 6) and bolts to the gas generator case. Torque boltsfinger tight.		29
	When all remaining manifold adapters (5) are positioned, remove bolts or locknuts, as applicable, from inlet manifold adapter (2). Assemble the locking plate (6) and reinstall the bolts or locknuts.		29
6	**CAUTION:** MAKE SURE ALL 14 MANIFOLD ADAPTER LOCKING PLATES ARE CORRECTLY INSTALLED.		
	CAUTION: PRIOR TO TIGHTENING BOLTS OR LOCKNUTS, CHECK SEATING OF METAL GASKET RELATIVE TO SHEATH AND BOSS. TIGHTENING SHOULD BE DONE EVENLY ON EACH ADAPTER TO PROVIDE FULL SEATING POTENTIAL OF METAL GASKET.		
7	Tighten all adapter mounting bolts, in a sequence, 15 to 20 lb.in. Retighten 32 to 36lb. in. in the same sequence. **Secure bolts with lock wire.**		29
	NOTE: After torqueing, a 0.001 in. gap is allowed between the adapter and sheath flanges (Ref. Fig. 203 lower drawing).		
8	Remove the blanking caps from the fuel delivery lines and connect the lines to theelbows. Tighten the coupling nuts 90 to 100 lb.in. and fasten with lockwire.		29

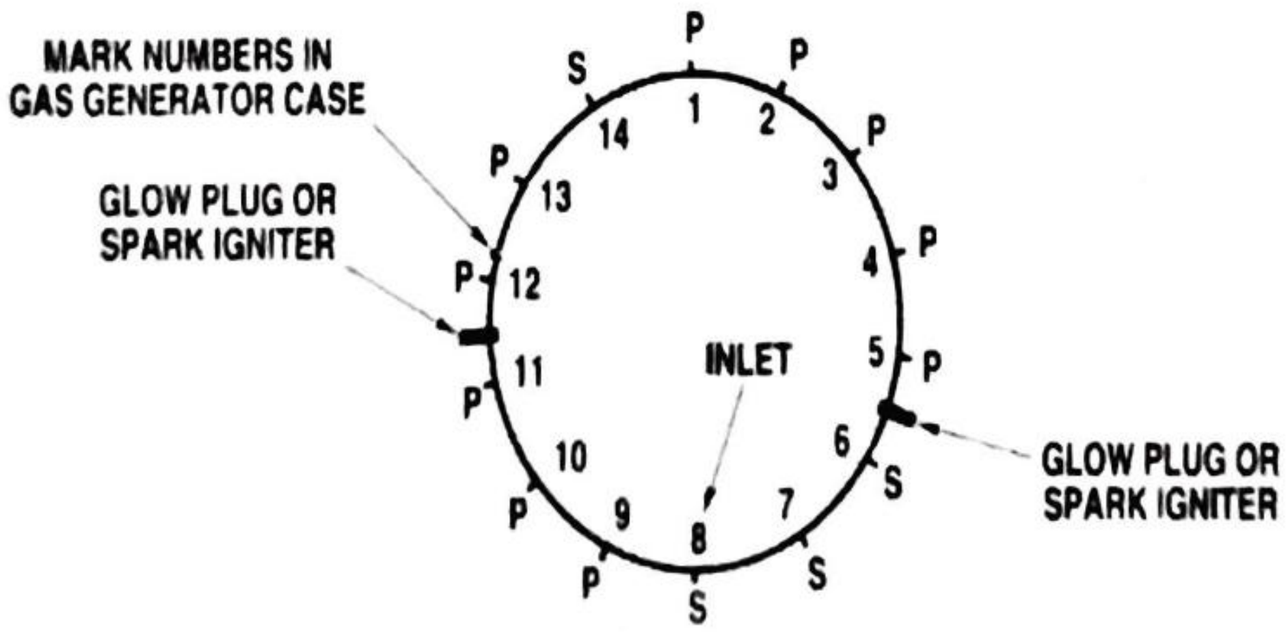


PRIMARY FUEL MANIFOLD
ADAPTER ASSEMBLY

SECONDARY FUEL MANIFOLD
ADAPTER ASSEMBLY



PRE-SB1372



POST-SB1372

FUEL MANIFOLDS AS VIEWED FROM REAR OF ENGINE

Figure 201 - Identification and Location of Fuel Manifold Adapters

NO	TASK	SIGNATURE	
		SIGN	STAMP
9	Remove the blanking caps from the spark igniters and from harness leads and connect the leads to the igniters. Tighten the connections finger tight, plus 45 degrees and fasten with lock wire. Secure the ignition lead loop clamps to the support brackets at the center fire seal lower mounts and tighten the nuts 32 to 36 lb.in.		29
10	Check function of fuel manifold installation (Ref. Adjustment/Test).		29

Installed Parts Record. (Post SB1372 worksheet)

Part Number	Description	Serial Number/s	Qty	Remarks
3019754	Secondary Inlet Adapter Manifold	UNK	1	NO DEFFECT FOUND
3011152	Primary Inlet Adapters	UNK	10	NO DEFFECT FOUND
3011154	Secondary Inlet Adapters	UNK	3	NO DEFFECT FOUND
3019491	Fuel Nozzle Sheaths	UNK	14	NO DEFFECT FOUND

PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER
ARIS KURNIAWAN	ENGINEER		9523
ROHPININDO. N. S.P	ENGINEER		9634

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : ARIS KURNIAWAN Stamp : 29

Signature : Place/Date : BERAU / 12/01/2022

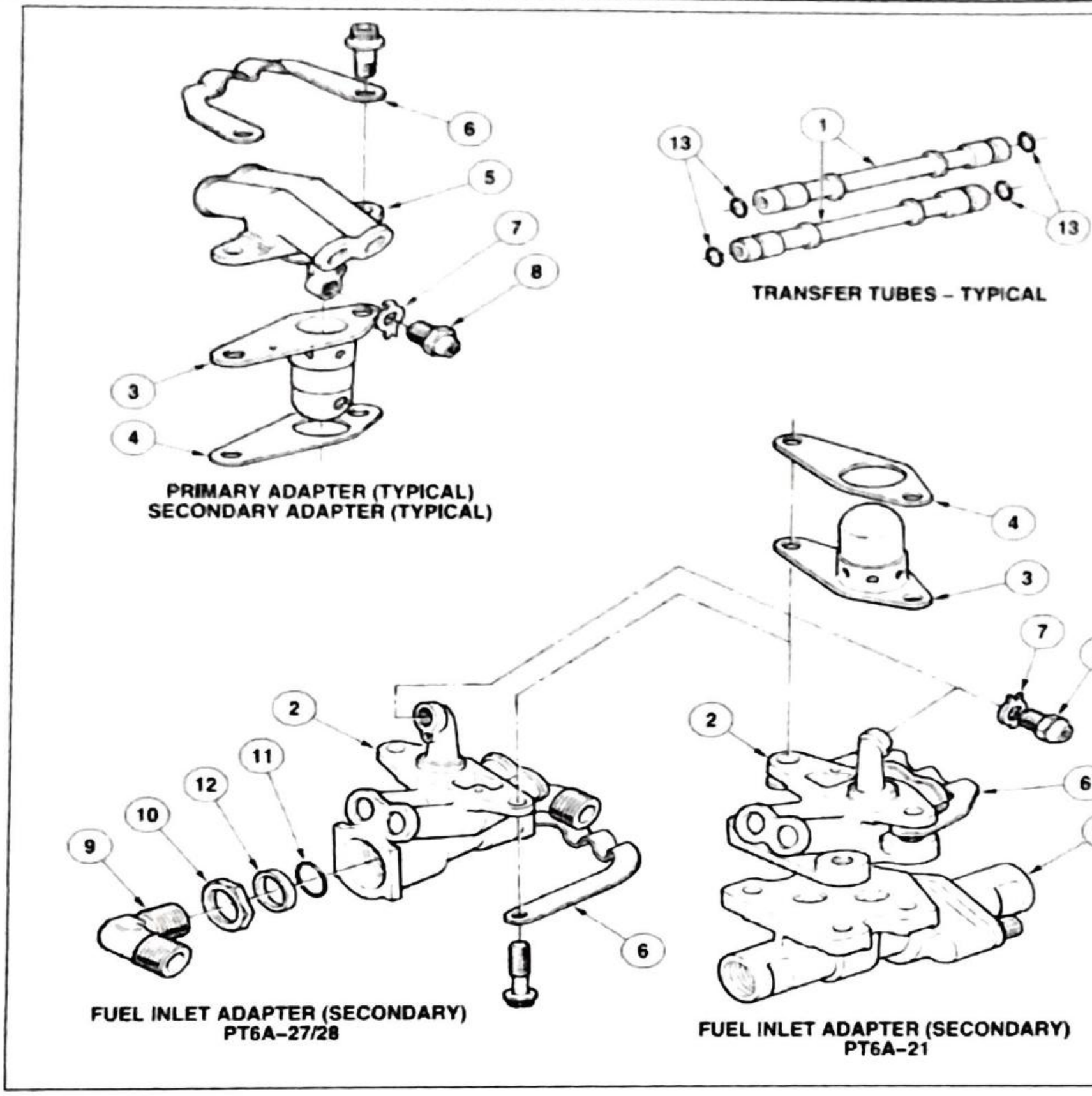


Figure 202 - Removal/ Installation of Fuel Manifold Adapter

Key to Figure 202

1. Fuel Transfer Tube
2. Fuel Manifold Inlet Adapter
3. Sheath
4. Gasket
5. Fuel Manifold Adapter
6. Lockplate
7. Keywasher
8. Fuel Nozzle
9. Elbow
10. Locknut
11. Preformed Packing
12. Back-up ring
13. Preformed Packing
14. Flow Divider and Dump or Purge Valve

NO	TASK	SIGNATURE	
		SIGN	STAMP
9	Remove the blanking caps from the spark igniters and from harness leads and connect the leads to the igniters. Tighten the connections finger tight, plus 45 degrees and fasten with lock wire. Secure the ignition lead loop clamps to the support brackets at the center fire seal lower mounts and tighten the nuts 32 to 36 lb.in.		
10	Check function of fuel manifold installation (Ref. Adjustment/Test).		

Installed Parts Record. (Post SB1372 worksheet)

Part Number	Description	Serial Number/s	Qty	Remarks
3014754	Secondary Inlet Adapter Manifold	UNK	1	NO DEFFECT FOUND
3011152	Primary Inlet Adapters	UNK	10	NO DEFFECT FOUND
3011154	Secondary Inlet Adapters	UNK	3	NO DEFFECT FOUND
3014491	Fuel Nozzle Sheaths	UNK	14	NO DEFFECT FOUND

PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER
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ROHPINDO. N S.P	ENGINEER		9634

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Signature : Place/Date : BERAU / 12/01/2022

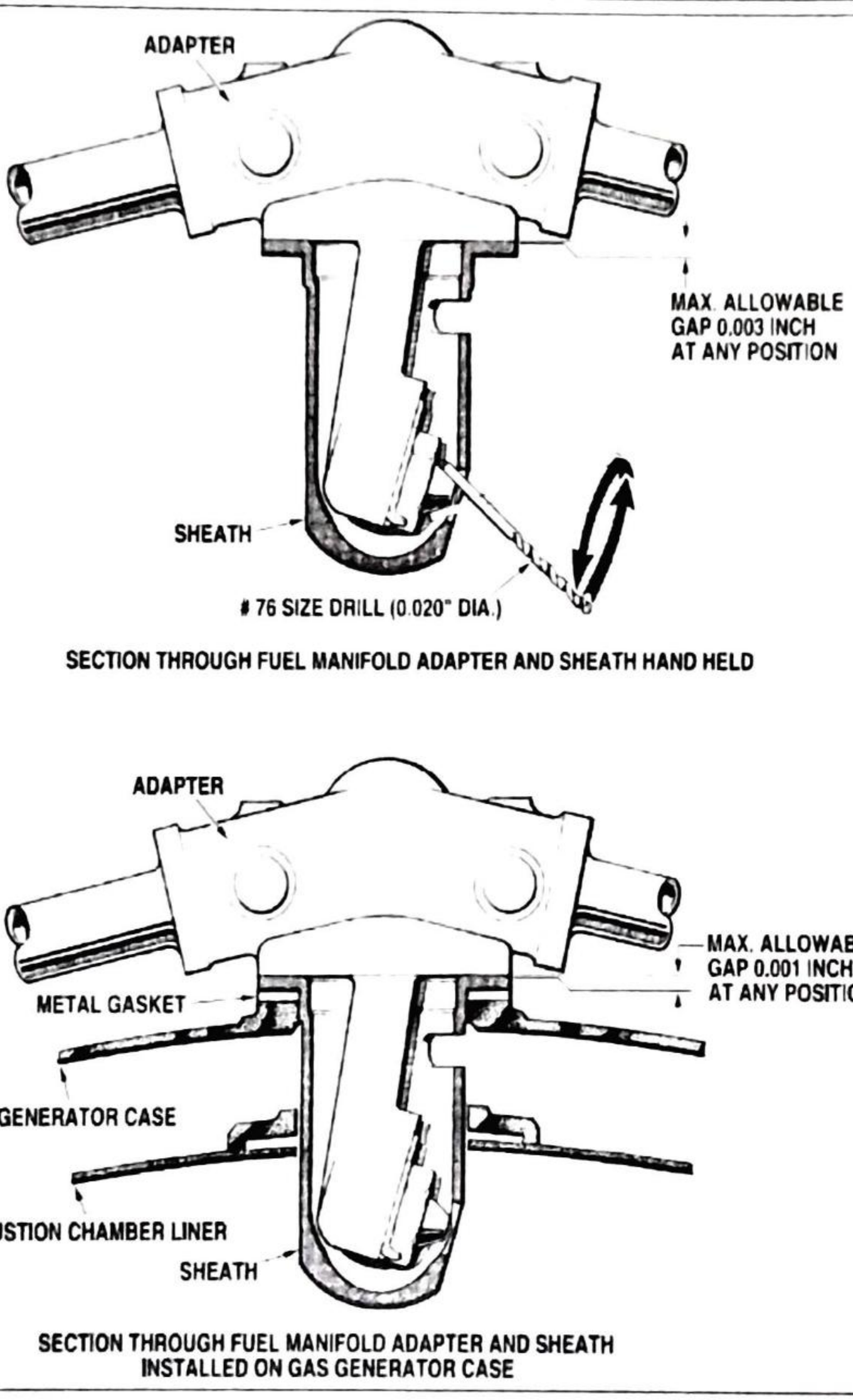


Figure 203 Fuel Manifold Installation - Clearance Checks

Engine Borescope Hot Section Inspection

Work No: **WO/006-SNB/XII/2021**

Engine Serial Number	Date	Base / Location	Aircraft Registration
PCE-PG0565	**29 December 2021**	**MALINAU**	**PK-SNB**

Aircraft Total Time	Aircraft Total Cycle	Reason For Borescope
400:57	**455**	**DUE FOR INSPECTION**

Note:
Record any discrepancies found during inspection, and/or take photographic evidence.
If None, then write No Findings. If you find defects, please quote EMM (Engine Maintenance Manual) Reference and Limitations.

Item	Work Description	SIGN	STAMP
1	Remove fuel manifold adapter as necessary (Ref. 73-10-05).		
2	Perform inspection of the First Stage Compressor. Defects: **NO DEFECT FOUND** If defects found, quote MM Limitation and References :		

Photo of First Stage Compressor 1st Quadrant

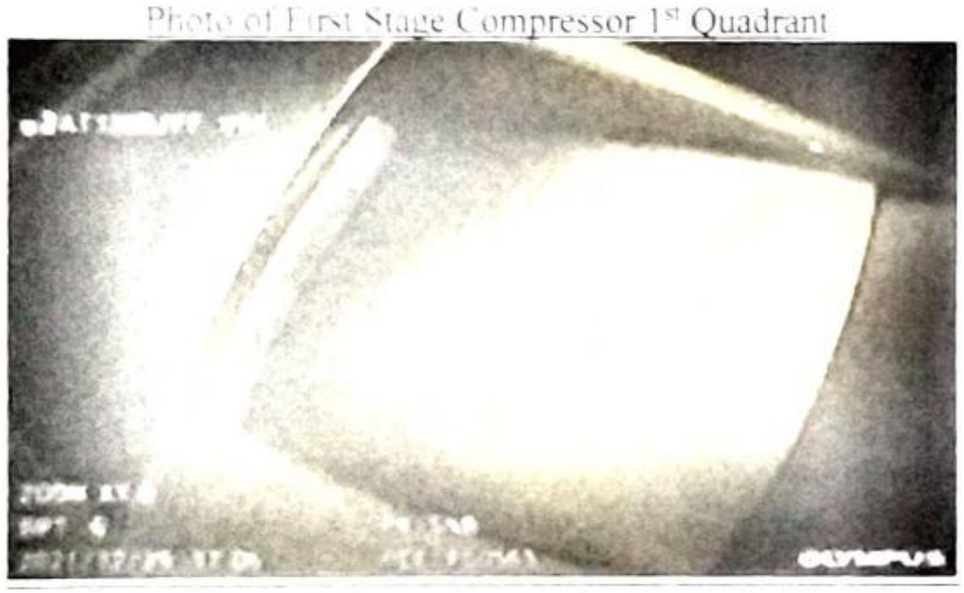


Photo of First Stage Compressor 2nd Quadrant

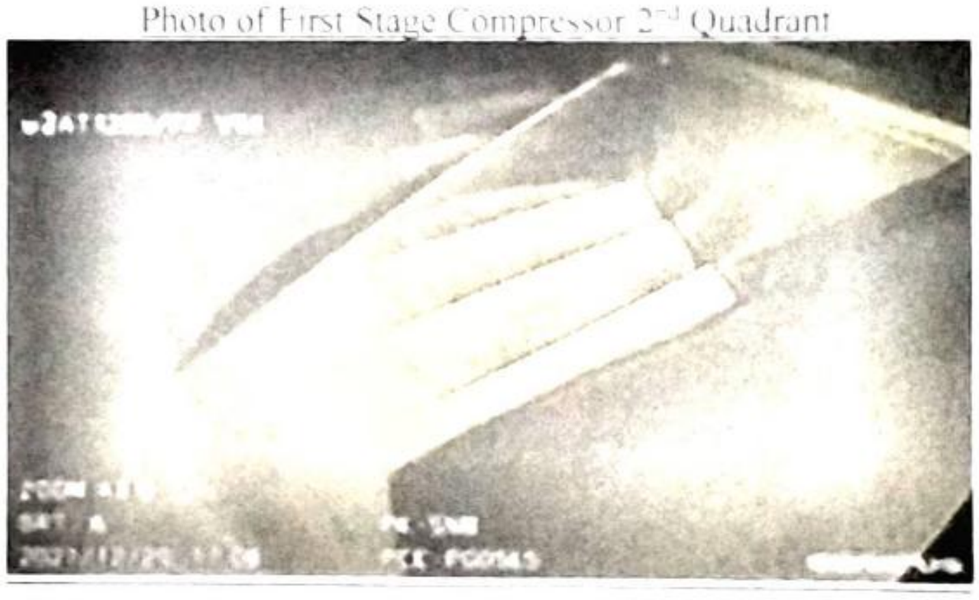


Photo of First Stage Compressor 3rd Quadrant

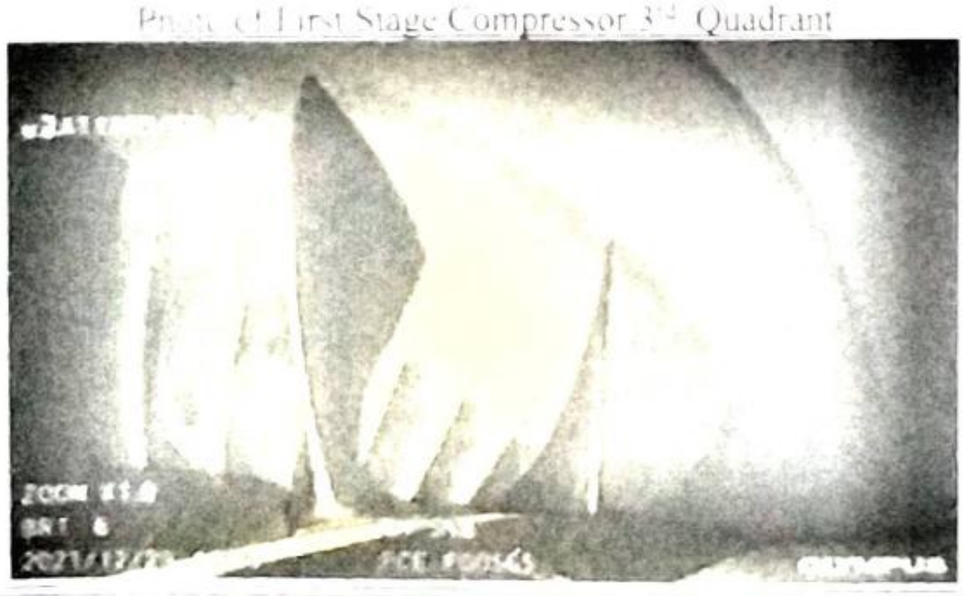


Photo of First Stage Compressor 4th Quadrant

		SIGN	STAMP
3	Perform inspection of Combustion Chamber Liner Assembly. Defects: **NO DEFECT FOUND** If defects found, quote MM Limitation and References :		

Photo of Combustion Chamber 1st Quadrant

Photo of Combustion Chamber 2nd Quadrant

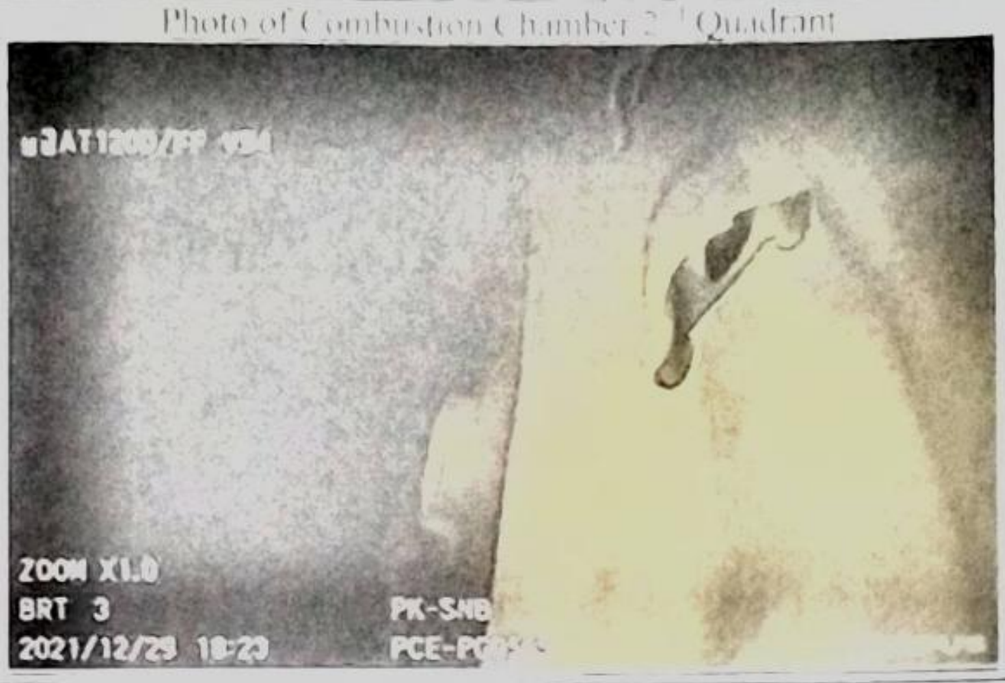


Photo of Combustion Chamber 3rd Quadrant

Photo of Combustion Chamber 4th Quadrant

		SIGN	STAMP
4	Perform Inspection of CT-Stator assembly. Defects: **NO DEFECT FOUND** If defects found, quote MM Limitation and References :		

Photo of CT Stator 1st Quadrant

Photo of CT Stator 2nd Quadrant

Photo of C.I Station 1 Quadrant

		SIGN	STAMP
5	Perform inspection of C.I blades and shroud segments. Defects. **NO DEFECT FOUND** If defects found, quote MM Limitation and References :		

Photo of Leading Edge C.I Blade 1 Quadrant

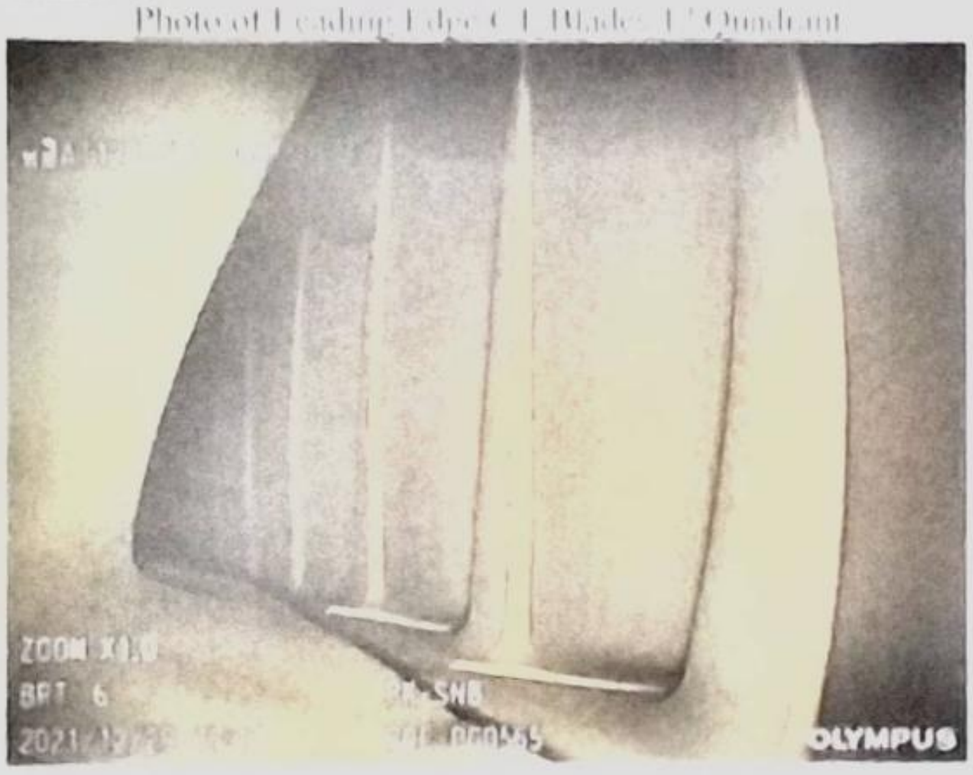


Photo of Leading Edge C.I Blade 2 Quadrant

Photo of Leading Edge C.I Blade 3 Quadrant

		SIGN	STAMP
6	Perform inspection Trailing Edge CT-Blades Defects: **NO DEFECT FOUND** If defects found, quote MM Limitation and References :		

Photo of Trailing Edge CT-Blades 1st Q

Photo of Trailing Edge CT-Blades 2nd Q

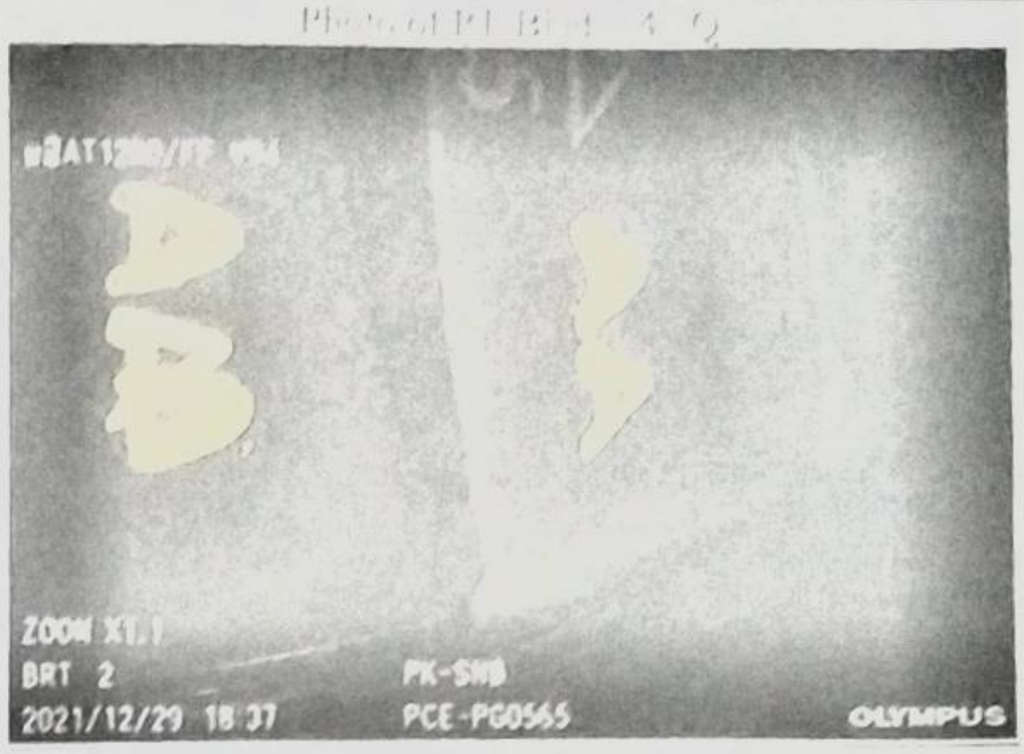
		SIGN	STAMP
7	Perform inspection of PT-Blades Defects: **NO DEFECT FOUND** If defects found, quote MM Limitation and References :		

Photo of PT-Blades 1st Q

Photo of PT-Blades 2nd Q

Photo of PT Blade 1 - 2

Photo of PT Blade 1 - 2



		SIGN	STAMP
8	Install fuel manifold adapter(s) (Ref. 73-10-05).		
9	Perform fuel leak check post fuel nozzle installation		

BORESCOPE PERFORMED BY

Name: **BRAMONO ONY TRIANTO** Signature : _____ Stamp : _____