PT. SMART CAKRAWALA AVIATION

WORK ORDER

Form: SCA/MTC/030

Subject : **Inspection 100 Hours / Annual & Add Task Due at 400 FH.**	No.	WO/006-SNB/XII/2021
	Date	22-Dec-2021
	A/C Reg.	PK-SNB MSN 1015
Reference : MP PILATUS PC-6 Rev. 0	Prepared By	TS
	Checked By	CI
	Approved By	TM

To : Engineer In Charge

Description :

1. Perform Inspection 100 Hours / Annual & Add Task Due at 400 FH.
2. Make an entry in Maintenance Log.
3. Return the Completed Work Order and Form to PPC.

#If any finding, please close the routine card, and transferred to inspection card.

Additional Work :

Compliance Statement	Sign & Date Company Lic. No.: (Engineer In Charge)	Signature (Technical Manager)

AIRCRAFT CHECK WORK SUMMARY
(Form: SCA/MTC/051)

DATE OF ISSUED	JO/WO #	TYPE OF MAINTENANCE	DATE OF ACCOMPLISHED
22-Dec-2021	WO/006-SNB/XII/2021	Inspection 100 Hrs / Annual	

A/C Type	Mfg. Serial Number	A/C Registration
PILATUS PORTER PC-6	1015	PK-SNB

AIRCRAFT DATA

Subject	Pos #	Serial Number (SN)	TTSN/TCSN
Engine	#1	PCE-PG0565	
	#2	-	
Propeller/Rotor	#1	FY4513	
	#2	-	
Landing Gear	NLG		
	LH MLG		
	RH MLG		

PACKAGE COVERED

No	Subject	Qty	Remark
1	Non-Routine Card	-	
2	Inspection Card	-	
3	Work Order	1	
4	Summary Inspection List	1	
5	Material and Tool List	-	
6	Escalation form	-	
7	CRS (SMI / Unscheduled Maintenance)	1	

INSPECTION CARD (IC) LIST (Finding during maintenance)

No	Taskcard Ref	Subject	Status		Name/ Sign & Stamp
			Open	Close	
IC-001					
IC-002					
IC-003					
IC-004					
IC-005					
IC-006					

IC-007					
IC-008					
IC-009					
IC-010					
IC-011					
IC-012					
IC-013					
IC-014					
IC-015					

Prepared by :	Checked by :	Verified by :	Approved by :
Technical Support	Chief Maintenance	Chief Inspector	Technical Manager
…….	…….	…….	…….

SUMMARY INSPECTION ITEMS
(Form: SCA/MTC/050)

WO Ref: WO/006-SNB/XII/2021

NO.	TASK CARD NO.	DESCRIPTION	DATE	EST MHR	NAME	STAMP
1	APPENDIX	ENGINE GROUND RUN CHECK SHEET				
2	APPENDIX	100 HOURS / ANNUAL INSPECTION				
3	APPENDIX	FUEL DISTRIBUTION SYSTEM – ADJUSTMENT TEST				
4	APPENDIX	FUEL INDICATING SYSTEM – ADJUSTMENT				
5	APPENDIX	FUEL SYSTEM UNDERWING TANK INSPECTION TRANSFER PUMP FILTERS				
6	APPENDIX	WHEEL AND BRAKES INSPECTION				
7	NRC-001	FUEL NOZZLE CHANGE				
8	NRC-002	BORESCOPE HOT SECTION INSPECTION				
9	NRC-003	STARTER/GENERATOR BRUSHES – CHECK FOR WEAR				
10	NRC-004	PROPELLER OVERSPEED GOVERNOR – CHECK OPERATION				
11	NRC-005	EMERGENCY BATTERY – OPERATIONAL TEST				
12	NRC-006	LEFT & RIGHT WING STRUT FITTING – VISUAL INSPECTION				
13	NRC-007	ENGINE DRIVEN FUEL PUMP – CHECK PUMP DRIVE SHAFT FOR BACKLASH				
14	NRC-008	COMPRESSOR – EXAMINE FOR CORROSION AND EROSION				

15	NRC-009	TURBINE – EXAMINE FOR CORROSION AND EROSION				
16	NRC-010	SPARK IGNITORS- FUNCTIONAL CHECK				
17	NRC-011	PROPELLER HARTZELL LUBRICATE				
16	SCA/MTC/023	EMERGENCY EQUIPMENT CHECK				

PT. SMART CAKRAWALA AVIATION

CERTIFICATE RETURN TO SERVICE

SCHEDULED MAINTENANCE INSPECTION
(CRS-SMI)

A/C TYPE	: PILATUS PORTER PC-6	TTSN	:
A/C REG	: PK-SNB	TCSN	:
MSN	: 1015	DATE	:

TYPE OF INSPECTION : INSPECTION 100 HOURS / ANNUAL & ADD TASK
DUE AT : 400 HOURS
REFF : MP PILATUS PC-6 REV. 0

EXCEPTION

AUTHORIZED PERSON

I hereby certify that this aircraft has been maintained accordance with CASR and Maintenance Program. Aircraft safe and airworthy for flight

NAME	CAT	AMEL/OTR NO	SIGN&STAMP	DATE
	AIRFRAME & POWER PLANT			
	EIRA			

THE NEXT DUE TYPE OF INSPECTION :

DUE AT :

Form: SCA/MTC/049

smartaviation	**INSPECTION CARD** **(Form: SCA/MTC/ 048)**	TECHNICAL DEPARTMENT

1. CARD #	2. JO/WO #	3. ORIGINATOR	4. CARD REF	5. DATE
6. A/C REG/MSN	7. A/C TYPE	8. TRADE	12. VENDOR ORDER #	
9. ZONE	10. STA	11. MTC TYPE		

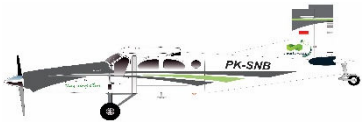
13. DESCRIPTION/DEFECT-IF FINDING OF CPCP INSPECTION, PLEASE COMPLETE SET. 20	14 PPC/ENG	15 DATE

16. CORRECTIVE ACTION		17 MECH	18 ENG. LIC	19 DATE
Performed at A/C TT : ..... A/C TC /LDG :				
20. CORROSION INFORMATION				
LOCATION	CAUSE OF DAMAGE			
	☐ Environment			
	☐ Internal Leakage			
CORROSION ☐ Isolated ☐ Widespread	☐ Chemical Spill			
CORROSION LVL ☐ 1 ☐ 2 ☐ 3	☐ LAV/Galley Spill			
PROPOSED ACTION ☐ Doublers	☐ Blocked Drain			
☐ Others	☐ Wet Insulation Blanket			
.....	☐ Other			
21. If the defect is RII, Please Sign this card finally by RII Inspector			INSP	DATE
NOTICE OF INSPECTOR				

22. PARTS REQUIRED						
PART DESCRIPTION	PART NO	QTY	SERIAL NO		STATUS	
			ON	OFF	CLOSE	OPEN

23. TOOLS REQUIRED			
DESCRIPTION	PART NO. / MODEL	NEXT CALIBRATION DATE	STATUS

ENGINE GROUND RUN CHECK SHEET - PT6A-27 ENGINE WITH FOUR BLADE PROPELLER (HARTZELL STC SA377CH)



PILATUS

HARTZELL

WORK ORDER NO.		**:**	
Aircraft Registration		**Aircraft Total Hours**	
Aircraft Serial No.		**Aircraft Total Landings**	
Engine Serial No.		**Engine TSN / TSO**	
Propeller Serial No		**Propeller TSN / TSO**	
Ambient Temp	**°C**	**FBP (Field Barometric Pressure)**	**In.Hg**
Date		**Time**	
Mechanic / Engineer		**Authorized Engineer**	
Reason For Ground Run			

Checks to be carried out. No:	**1 2 4 5 7 8 9 10 11 12 13 14 15**

Engine Ground Run Check Frequency

Check Number	1	2	3	4	5	6	7	8	9	10	11	12	13	14	15
Each 100 / Yearly	x	x		x			x	x			x	x	x	x	x
Each 200									x						
Pre-Complete Overhaul	x	x	x	x		x	x	x	x	x	x	x	x	x	x
After Short Term Storage															x
After Long Term Storage	x	x	x	x		x	x	x	x	x	x	x	x	x	x

In additional the following check must be carried out after Installation, Repair and Adjustment of any ofthe following components.

Check Number	1	2	3	4	5	6	7	8	9	10	11	12	13	14	15
Engine Installation	x	x	x	x	x	x	x	x	x	x	x	x	x	x	x
Propeller Installation		x	x	x	x			x							
Fuel Control Unit	x				x	x	x	x		x	x				
HP Fuel Pump						x	x								
Fuel Nozzle						x	x								
Starting Flow Control	x				x		x	x							
Emer Fuel Control Actuator											x				
Prop Governor	x		x	x	x		x	x							
Prop Overspeed Governor									x						
Compressor Bleed Valve						x	x								
Engine Controls	x			x	x			x	x						
Low Pitch Warning Switch				x											
Suction Components														x	

Use this sheet's to record engine run result, use in conjunction with task cards.

NO.	CHECK	TARGET	ACTUAL
	ENGINE START		
	ITT (Troubleshoot If More Than 925□C)	Max. 1090 □C	□C
	Cabin Heat	OFF	OK?
1	Low Idle (Minimum Governing) Speed	51 - 53 % Ng	% Ng
	Fuel Pressure / Boost Pump OFF	Light out or 25 ± 5 psi	OK?
	ITT		□C
	Oil Pressure		psi
	Oil Temperature		□C
2	Propeller Governor		
	Maximum Np	1980 - 2000 rpm (90.0 - 90.9 %)	rpm
	Py Disconnected		% Ng
	Py Connected		% Ng
	Difference	Maximum 0.3% Ng	%
	Airbleed Link at Minimum	1900 - 1950 rpm (86.4 - 88.6 %)	rpm
	Aircraft with SB 161:		
	Propeller Control Lever at Minimum	1880 - 1900 rpm (85.5 - 86.4 %)	rpm
3	Propeller Fine Pitch Setting (High Idle)		
	Target Torque	psi	psi
	Power Lever to Give Np	1694 rpm (77 %)	rpm
	Basic High Idle	68 - 72% Ng	%Ng
4	Propeller Low Pitch Warning		
	PCL from Reverse to Detent	Light OFF	
		1 to 2 mm before Detent	mm
5	Minimum Pitch in Flight		
	Ng	67 - 73 %	% Ng
	Np	1800 - 1950 rpm (81.8 - 88.6 %)	rpm
	Torque	4 - 7 psi	psi
6	FCU Maximum Governing Speed (Ng)	97.1 % Ng	% Ng
	(Trim stop deployed)		

NO.	CHECK	TARGET	ACTUAL
7	Engine Performance	Ref: AMM 71-00-00	
	Target Torque Pressure	psi	psi
	Fuel flow (Actual minus 23 lb / hr or 3.4 gal / hr)	lb / hr	lb / hr
	Target Ng	% Ng	% Ng
	Maximum ITT	^0C	^0C
8	Reverse Power Setting		
	Np	1880 - 1925 rpm (85.5 - 87.5 %)	rpm
	Torque	psi	psi
9	Propeller Overspeed Governor		
	Test Lever Selected to:		
	TEST	1880 - 1920 rpm (85.5 - 87.3 %)	rpm
	NORMAL	1980 - 2000 rpm (90.0 - 90.9 %)	rpm
10	Acceleration		
	64 % – 90 % Ng	2.5 – 4 secs	secs
	Deceleration		
	85% to 60% Ng or low idle speed(Whichever comes first)	Maximum 6-12 sec (Dependent upon altitude)	secs altitude (kFt)
	Manual Override (MOR) (Aircraft with SB 164)		
	Use Toggle Switch In Small Increment	Increase to 15% above Idle	OK?
	(REF. to WARNINGS and CAUTIONS in Check 11)	(Max Increase less than 4 % per Second)	
		Decrease To Idle	OK?
		(Max Decrease less Than 4% per Second)	
12	Oil Pressure	80 -100 psi	psi
13	Generator (Ref. 24-30-00)	Online by 60% Ng	% Ng
14	Suction (High Idle)	4.5 – 5.2 in. Hg	in. Hg
15	Engine Rundown Time After Stop	MIN 30 secs	secs
Additional			
Generator Check (High Idle Under Load)		27.75 – 28.25 VDC	VDC
After Engine Run			
Check Eng. For Signs of Fuel/Oil/Air Leaks		NO LEAKS FOUND	OK?
Safety All Screws, Bolts, Locknuts as Req.			OK?

Ref. AMM Pilatus Porter PC6 Chapter 05-22-01, P&WC Maintenance Manual Model PT6A-27 Manual Part No. 3013242Chapter 72-00-00, Propeller Owner's Manual Hartzell (Manual 149)

100 HOURS / ANNUAL INSPECTION

Reg. Mark	:	PK -	Date	:	
MSN	:		Station	:	
TSN / CSN	:		WO No.	:	

NO	TASK	SIGNATURE	
		SIGN	STAMP
1	Aircraft document Perform inspection document folder (onboard). Check content completeness of aircraft document. (Ref. CASR 91.25)		
2	Emergency equipment list Perform emergency equipment list. Form SCA/MTC/023. Make one copy and insert into the aircraft document folder.		
3	After engine run safety all screws bolts locknuts as applicable (duplicate inspection). Perform after engine run safety all screws bolts locknuts as applicable.		
4	After engine run check engine for signs of fuel, oil, air leaks. Perform after engine run check engine for signs of fuel, oil, air leaks.		
B. AIRFRAME			
Aircraft - General			
1	External surfaces Examine, particularly for fuel, oil and hydraulic leaks.		
2	Aircraft external Wash.		
3	Aircraft preparation Remove and examine the protective covers, blanks and restraints. Replace if damaged, torn or is not properly install.		
4	Placard and markings Examine and replace as necessary.		
5	Aircraft lifting Put the aircraft on jacks.		
6	Fuselage Remove access panels and fairings.		
7	Fuselage - internal Remove cockpit and cabin seats and interior fuselage linings.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
8	Wings Remove access panels and fairings (not fuel tanks).		
9	Engine cowls Remove.		
10	Empennage Remove access panels and fairings.		
Chapter 21 - Air Conditioning			
1	Engine bleed air line and hoses Examine.		
2	Air inlet screens, filters and hoses Clean and examine.		
3	Mixer unit Examine.		
4	Butterfly vents - passenger cabin Examine.		
5	Emergency shut-off valve Examine.		
6	System component, pipes, cables, controls and linkages. Examine.		
7	System cables, controls and linkages Lubricate (Material No. P04-037).		
8	Air conditioning system Check operation during engine ground run checks.		
Chapter 24 - Electrical Power			
1	Generator voltage Check generator voltage at high idle under load_____VDC.		
Chapter 25 - Equipment and Furnishings			
1	Pilot and Co-pilot seats Examine seat and seat attachments. Make sure that the seat adjustment mechanism operates correctly. Lubricate moving parts. (Material No. P04-011).		
2	Pilot and Co-pilot seats harnesses Examine. Inertial reel system - Operational test.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
3	Passenger seats Examine seats, seat attachments and seat harnesses. If seats with Torso Restraint System are installed, make sure the backrest release mechanism operates correctly. Lubricate moving parts (Material No. P04-028).		
4	Linings and curtains Examine.		
5	Emergency locator transmitter Examine Check battery expiry date _____		
6	Fire extinguisher Examine Check expiry date _____		
7	First aid kit Examine Check expiry date _____		
8	Crash axe Make sure it is stored correctly.		
9	Stretchers (if Installed) Examine stretchers and mountings.		
10	Parachute dispatch system (if Installed) Examine. Signal light system - Operational test.		
Chapter 27 - Flight Controls - General			
1	Control column Examine. Check for excessive play at Teflon bearing at base of column by pulling up and pushing down on column. Maximum play is 0,2 mm (0.008 in.).		
2	Control lock Examine.		
3	Rudder pedals Examine. Check for excessive play and full and free range of movement. Especially examine the brake pedal at the weld for cracks		
Chapter 27 - Flight Controls - Ailerons			
1	Aileron control system Examine system including stops, cables, pulleys, guides, and bellcranks.		
2	Aileron controls Do a functional test.		
3	Aileron to rudder interconnect spring Examine.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
4	Aileron trim tab electrical actuator (Electrical system) Examine.		
5	Aileron trim system (Mechanical or Electrical trim tab systems) Check neutral settings, sense and range of movement. Check cockpit indicator.		
Chapter 27 - Flight Controls - Rudder			
1	Rudder control system Examine system including stops, cables, pulleys, guides, and bellcranks.		
2	Rudder Do a functional test.		
3	Rudder trim lab electrical actuator (Electrical system) Examine.		
4	Rudder trim tab (Mechanical or Electrical trim tab systems) Do an inspection / check. Check neutral settings and range of movement. Check cockpit indicator.		
Chapter 27- Flight Controls - Elevator			
1	Elevator control system Examine system including stops, cables, pulleys, guides, and bellcranks.		
2	Elevator control system Do a functional test.		
3	Elevator balance tabs Check neutral settings, sense and range of movement.		
Chapter 27 - Flight Controls - Stabilizer			
1	Horizontal stabilizer trim actuator Electrical system Examine.		
2	Horizontal stabilizer trim actuator attachments Examine. On the Lugs, look for cracks and signs of excessive asymmetrical wear.		
3	Horizontal stabilizer trim system Examine.		
4	Electrical system Do a functional test.		
Chapter 27 - Flight Controls - Flaps			
1	Flap actuator and support bracket (Electrical system) Examine		

NO	TASK	SIGNATURE	
		SIGN	STAMP
2	Flap control system – Bellcranks, levers and push/pull rods (Elect. Sys) Examine		
3	Flaps Do a functional test.		
Chapter 28 - Fuel System			
1	Water collector tank and fuel filter Drain a minimum of 0,25 liters (0.5 pint) of fuel from each drain valve. Make sure that there is no water in the fuel.		
2	Fuel filter Examine.		
3	Fuel shut-off valve Examine.		
4	Main fuel tanks Examine vents, filler caps and seals.		
5	Fuel pipes and hoses Examine.		
6	Air maze fuel filter Examine inlet pipe and adjacent oil hose for chafing.		
7	Perform fuel filter clean (Airmaze). P/N: 968.35.21.147 S/N: NSN. P/N OFF : …… P/N ON : ……		
8	Fuel flow transmitter Examine.		
9	Engine driven fuel pump (EDP) Examine.		
10	Fuel system Set shut-off valve to OPEN and then set the AUX F PUMP to ON. Look for leaks on complete fuel system and unusual noise from the fuel pump. Set AUX F PUMP to OFF and then set shut-off valve to CLOSE.		
11	Fuel System Fuel distribution system test or adjustment.		
12	Fuel Indicating System Fuel indicating system test or adjustment.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
Chapter 28 - Fuel System - Underwing Tanks			
1	Underwing tanks Examine.		
2	Transfer pump filters Examine and clean.		
3	Underwing tank system Check operation.		
4	Underwing Tank Fuel System Fuel system underwing tank inspection (if installed).		
5	Underwing Tank Fuel System Fuel system underwing tank inspection transfer pump filter.		
Chapter 32 – Landing Gear and Brakes			
1	Brakes Check brake pad wear. Visual Insp. Beringer.		
2	Main wheels rotation and debur discs. Perform main wheels LH and RH rotation and debur discs.		
3	Hydraulic pipes Examine.		
4	Brake system Check brake fluid level. Apply brakes, examine system for leaks.		
5	Park brake system Examine. Make sure system operates correctly		
6	Main wheel tires Examine.		
7	Main wheels Remove. Examine bearings, axles and wheels. Lubricate bearings and axels with grease (MIL - G - 81322). On installation rotate the wheel position LH to RH and vice versa		
8	Main wheels Perform main wheels inspection fill out the main wheel's inspection.		
9	Brake discs Examine. Check for wear.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
10	V-struts Examine. Note: If you find damage that is more than 0,127 mm (0.005 in.) deep, reject the V-strut.		
11	V-struts If you find damage that is 0,127 mm (0.005 in.) deep or less, refer to Pilatus CMM 02270 for minor repair procedures.		
12	V-struts attachments Examine. Lubricate (Material No. P04-002)		
13	Main landing gear shock struts Examine. Lubricate (Material No. P04-002). Check fluid level.		
14	Main wheels Install. Inflate tire.		
15	Main wheel - dirt scraper Examine.		
16	Tail landing gear Examine. Make sure there are no cracks in the welded seams. Check the locking-lever pivot pins. Lubricate (Material No. P04-002)		
17	Tail wheel tire Examine.		
18	Tail wheel Remove. Examine bearings, axle and wheel. Lubricate bearings with grease (Material No. MIL-G-81322; Aeroshell 22, Royco22, Mobil 28)		
19	Tail wheel Install. Inflate tire.		
20	Steering system Examine. Check cable tension and range of movement.		
21	Debris guard Examine.		
22	Steering system Inspect steering cable tension with a turn buckle installed in the steering cable – adjust the turnbuckle to give a cable tension of minimum 32 lbs., maximal 35 lbs. and install two new locking clips.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
Chapter 35 - Oxygen System			
1	Oxygen bottle(s) and attachment brackets (if installed) Examine.		
2	Oxygen system pipes, flexible tubes and fittings (if installed) Examine.		
3	Oxygen regulators (if installed) Examine.		
Chapter 52 - Doors			
1	Pilot, Co-pilot doors Examine. Remove safety wire. Make sure that the emergency release mechanism and latching mechanism operate correctly. Do the check of vertical play of the door Lubricate mechanism (Material No. P04-011). Install safety wire. (Material No. P02-021)		
2	Cabin RH / LH sliding door Examine door, sliding rails, rollers, stops and seals Make sure that the latching mechanism operates correctly. Lubricate mechanism. (Material No. P04-037)		
3	Cabin trap-door (if Installed) Remove trap-door hatch cover. Examine doors, hinges, seal, and structural damage. Make sure that the latching mechanism and door release mechanism operate correctly. Test door for correct operation. Lubricate mechanism (Material No. P04-037)		
Chapter 53 - Fuselage			
1	Access panels and fasteners Examine.		
2	Fuselage - external Examine.		
3	Fuselage - internal Examine these structures as follows: - cockpit floor - cabin floor - cabin floor T-rails - door frames - accessible frames, stringers, and skin.		
4	Fuselage Make sure that the drain holes are not blocked.		
Chapter 55 - Stabilizers			

NO	TASK	SIGNATURE	
		SIGN	STAMP
1	Empennage Examine internal skin and structures as far as possible. Examine panels and fasteners. Make sure that the water drain holes are not blocked.		
2	Dorsal fin Examine.		
3	Vertical stabilizer Examine.		
4	Rudder - support structure Examine rudder support brackets, torque tube, control rod attachment points and attaching parts.		
5	Rudder Examine rudder skin and structure, balance weight attachment and mountings for static discharge wicks		
6	Rudder upper attachment Remove access panel EL4. Examine the attachment bolt and lockwire for security. Install access panel EL4.		
7	Rudder trim tab Examine tab. hinge, control rod attachment point and attaching parts.		
8	Horizontal stabilizer Inspection/ Check.		
9	Horizontal stabilizer actuator Examine the attachment brackets		
10	Elevator support structure Examine elevator support brackets, hinge bearings, control rod attachment points, control lever and attaching parts		
11	Elevator Examine skin and structure, fixed tab (H4 only) and mountings for static discharge wicks		
12	Elevator attachments Remove access panels ET1 and EB1 Examine the attachment bolts and lock wire for security. Install access panels ET1 and EB1		
13	Elevator balance tab Examine tab, hinges, control attachment points and attaching parts. Lubricate hinges. (Material No. P04-011)		
Chapter 56 - Windows			
1	Windows and windshields Examine.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
2	Emergency window Examine.		
Chapter 57 - Wings			
1	Access panels and fasteners Examine.		
2	Wing - external Examine skin and structure, particularly in area of fuel tanks, all access hole and external component or equipment attaching points. Look for loose rivets along the main spar (this can indicate advanced corrosion of the spar cap).		
3	Wing - internal Examine internal skin and structure, particularly in the area of fuel tank, as far as possible. Look for signs of corrosion on the upper and lower main spar caps.		
4	Wings Make sure that the drain holes are not blocked.		
5	Wing struts - external Examine attachment brackets. Examine strut exterior.		
6	Wing struts - internal Examine.		
7	Wing tips Examine.		
8	Aileron support structure Examine aileron support brackets, hinge bearings, control rod attachment points and attaching parts.		
9	Ailerons Examine aileron skin and structure, balance arms and static discharge wicks.		
10	Aileron - balance tabs Examine balance tabs, tab control rods, rod ends, support brackets, hinges and attaching parts. Lubricate hinges (Material No. P04-011).		
11	Flap support structure Examine flap support brackets, hinge bearings, control rod attachments, actuator support bracket and attaching parts.		
12	Flaps Examine structure and skin. Use a mirror and light to examine the skin of the flaps and slats for cracks in the areas where the angles are attached.		
General - Close Up NOTE: Do these steps when the engine, electrical and avionic inspections are complete			

NO	TASK	SIGNATURE	
		SIGN	STAMP
1	Access panels and fairings Install.		
2	Fuselage - internal Install internal linings.		
3	Engine cowls Install.		
4	Aircraft Remove the aircraft from jacks.		
5	Aircraft Make sure that the work area is clean and clear of tools and other items.		
C. PROPELLER & ENGINE **Chapter 61 - Propeller**			
1	Spinner dome Remove.		
2	Propeller de-ice boots, slip-ring and brushes Examine.		
3	Slip-ring (Beta) Examine. Check gap between slip-ring and carbon block is no more than 0,50 mm (0.02 in.).		
4	Spinner body and backplate Examine.		
5	Blades Examine.		
6	Spinner Dome Install.		
Chapter 71 - Powerplant			
1	Power Recovery Wash Perform power recovery wash.		
2	Engine Run Perform engine run IAW HARTZELL STC SA377CH.		
3	Engine compartment Clean engine, engine compartment and cowlings.		
4	Engine compartment Examine. Make sure water drain holes are not blocked.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
5	Powerplant and accessories Examine		
6	Powerplant and accessories Inspect and pay particular attention to rear linkage cam box, fuel control unit arm, telescopic rod and rod end fittings. Disconnect rod ends and clean using solvent (PWC11-027) or (PWC11-031). Examine rod end for corrosion, roughness in rotation, side play and radial play. Lubricate with light grease (PWC04-001) or MIL-G-23827 after engine external wash. Reinstall rod ends and torque to specified value. (Ref.76-10-00) Check free movement and linkages.		
7	Powerplant and accessories Air inlet screen - Inspect cleanliness. (Ref.72-20-00) Inspect the air inlet screen wire mesh for cleanliness and/or damage. Screens with broken wire mesh must be replaced. Clean undamaged screens (Ref. Cleaning / Painting). Inspect the rubber sealing rims and flanges of the screen for security and damage.		
8	Powerplant and accessories Gas Generator Case - Inspect External surface, and fire seal mount ring brackets for cracks, distortion and corrosion. (Ref. 72-30-04) Examine for general condition, including cracks, distortion, corrosion and evidence of overheating. Minor corrosion on exposed surface of gas generator case may be removed. (Ref. Approved Repairs). If the condition of the corrosion exhibited on the exposed surfaces of the gas generator case indicates that further examination of the fuel manifold and igniter bosses is required, remove the fuel manifold adapters (Ref. 73-10-05, Removal/Installation) and spark igniters. (Ref. 74-20-00, Removal/Installation). Examine the mounting pads, fuel nozzle bosses and machined surfaces for corrosion and wear. Isolated corrosion pitting not closely grouped, less than 0.010 inch deep, not covering more than 75 percent of the surface is acceptable without repair.		
9	Powerplant and accessories Fireseal Mount Rings - Inspect Cracks and attachment of brackets and seals (Ref. 72-30-01/-02) Examine the rear fireseal mount ring halves for attachment, damage and condition. NOTE: For the external tubes/lines passing through the mount rings, refer to the relevant chapters in this manual. Examine the circumferential insulating strips for attachment. Loosened strips may be rebonded.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
10	Powerplant and accessories Exhaust Duct - Inspect cracks and distortion. (Ref. 72-50-05, Maintenance practices) Examine the outer surface condition for buckling, ripples or similar distortion. Inspect outer surface, particularly in vicinity of flanges A and C for cracking in metal skin, welds, or flange bolt holes. Inspect exhaust port flanges for cracking. Cracks not exceeding 0.500 inch in length and do not progress into the stitch weld or cracks in a tangential direction not exceeding 1.000 inch long are acceptable provided they are stop drilled with a 1/16 (0.0625) inch drill. Check for the integrity of internal structure through the exhaust ports. NOTE: Refer to the Aircraft Maintenance Manual for removal/ installation of the exhaust stubs. Examine the internal structure as far as possible for cracks, looseness and distortion. Inspect Engines that exhibit interior welds (Ref. 72-50-05, Maintenance Practices) visually inspect the forward area of the exhaust duct for cracks, from the propeller reduction gearbox mounting flange to 2 inches aft around the entire circumference of the duct. Exhaust ducts are considered serviceable provided.		
11	Powerplant and accessories Accessories - Inspect attachment of accessories and linkages, air, oil, fuel lines (Ref. 73-10-07/-08) or (Ref. 70-00-00, Standard Practices Inspection). Inspect Fuel, Oil and Air Tubes from scratches, Nick, chafing, dents, pitting, rust and strainer. Inspect Security of pneumatic lines (Ref. 73-10-07/-08) Examine tube assemblies (Ref. 70-00-00, STANDARD PRACTICES - INSPECTION). Blend out minor damage that does not exceed specified limits. Replace the tube assemblies damaged beyond specified limits. Inspect heated rear pneumatic line.		
12	Engine External Examine.		
13	Engine flexible and rigid pipes Examine.		
14	Engine cowling and seals Examine.		
15	Fireshields and seals Examine.		
16	Shock mounts Examine.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
17	Support ring Examine.		
18	Support struts Examine.		
19	Electrical harnesses Examine.		
Chapter 72 - Engine			
1	Compressor inlet screen Clean. Examine.		
2	Gas generator case Examine.		
3	Propeller shaft oil seal Examine, look for oil leaks		
4	Accessories Examine.		
Chapter 73 - Engine Fuel and Control			
1	HP fuel pump Examine.		
2	HP fuel pump outlet filter Examine, replace if contaminated		
3	Fuel HP Outlet Filter Perform fuel HP outlet filter replacement. P/N : AN6235-3A or ALTERNATIVE P/N. P/N OFF : ..... P/N ON :		
4	Fuel control unit Examine Check for leaks from vent. (Ref. P&WC EMM 73-20-00) Check flow divider and dump valve for installation and leaks (Ref. EMM 73-10-06). Check FCU for installation, linkages and pneumatic tubes (Ref. EMM 73-20-00). Evidence of FCU bearing washout indicated by traces of blue dye effluent is caused by a mixture of bearing grease and fuel. For post-SB1472 engines fitted with a manual override on the fuel control, check FCU Manual Override System for static operation (Ref.EMM 71-00-00).		

NO	TASK	SIGNATURE	
		SIGN	STAMP
5	Fuel control unit Perform SIL NO. PT6A-221R01 – FCU Health Monitoring - Deceleration Check. Ref. P&WC PT6A-27 MM 71-00-00		
6	Pneumatic System Check P3 filter for installation. Clean or replace filter, dependent on condition, service experience or environment.		
7	Starting flow control unit Examine.		
8	Propeller governor Examine.		
9	Air pipes Examine.		
10	Fuel pipes Examine.		
11	Gas generator case drain valves Examine.		
12	Igniter exciter Examine and check ignition system/current regulator for installation and condition (Ref.74-10-01 and 74-10-02) Inspect the ignition exciters for signs of damage and general condition. Inspect the input and output connectors for damage, paying particular attention to the connector threads for corrosion. Inspect the cover and box of the regulator for general condition. A cracked or distorted mounting bracket on the box, or loose components on the box or cover, must be repaired at an overhaul facility. Inspect the seal on the box and the sealing gasket on the cover for general condition. A loose seal or gasket may be rebonded using adhesive cement (PWC08-010).		
13	Ignition cables Examine and check ignition cable for chafing, wear and installation (Ref.74-20-01) Inspect cables for signs of damage to braiding and general condition. Inspect cable coupling nuts for corrosion. Inspect central conductor and insulation for contamination and burning. Do retention test on igniter end of cable only: - Connect contact with tool (Ref. Table 201). - Contact must hold a 0.125 lb. weight. - If contact does not hold weight, ship cable to an authorized repair shop for inner cable replacement.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
14	Spark igniters Examine and check spark igniters/glow plugs for cleanliness and erosion. Check function (Ref. 74-20-02 and 74-00-00). Inspect the exterior cylindrical area of the firing end of the igniter shell for chafing wear. Wear is acceptable to a depth of 0.015 inch. Inspect the igniter shell and electrode for erosion (Ref. Fig. 207 and Table 202). If erosion equals or exceeds amounts shown, reject the spark igniter. Do a functional test on acceptable and replacement spark igniters (Ref. 74-00-00, Adjustment/Test).		
15	Interconnect rod Inspect accessible lockwire and safety cable for security and installation of the interconnect rod.		
16	Idle control system Examine.		
17	Power control system Examine.		
18	Propeller control system Examine.		
19	Engine controls Lubricate rod ends with grease. (Material No. P04-002).		
20	Emergency fuel control system Examine. Do a functional test.		
Chapter 78 - Exhaust			
1	Exhaust duct Examine.		
2	Exhaust stubs Examine.		
Chapter 79 - Oil			
1	Oil cooler system Examine. Flap - Do an operational test.		
2	Oil filter Examine and clean.		
3	Chip detector Do a functional test. Check Magnetic Detectors for continuity.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
4	Scavenge Oil pump Examine.		
5	Oil filler cap and dipstick Examine.		
6	Oil separator (Aircraft with SB75) Examine.		
General			
1	Powerplant Make sure that the work area is clean and clear of tools and other items.		
2	Powerplant Do a functional test.		
3	Powerplant (Post P&WC SB 1568 only) Do a deceleration check. NOTE: Not required if FCU is identified with 'RE52' or 'SB 73-3', or with a serial number that has the letter 'F' as a prefix.		
D. ELECTRICS AND INSTRUMENTS **Chapter 21 - Air Conditioning**			
1	Cockpit blower motor Examine and operational test.		
2	Cabin blower motor Examine and operational test.		
Chapter 24 - Electrical Power			
1	Battery mountings Examine attachment fittings, ventilation hoses, cable connectors, wiring.		
2	External power receptacle Examine.		
3	Starter/Generator Examine.		
4	Starter/Generator Examine QAD adaptor and clamp.		
5	Starter and power generation relays Examine. Functionally test during engine ground run.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
6	Voltage regulator Examine. Functionally test during engine ground run.		
7	Cockpit - switches and circuit breakers Examine. Make sure that placards are readable.		
8	Cables, plugs, connectors, relays, terminal blocks Examine in these areas: - engine compartment - cockpit - fuselage - empennage - wings		
9	Bonding Examine bonding leads in these areas: - engine compartment - cockpit - fuselage - empennage - wings - landing gear		
Chapter 27 - Flying Controls			
1	Aileron trim actuator Examine. Operational test.		
2	Rudder trim actuator Examine. Operational test.		
3	Flap actuator Examine. Operational test.		
4	Horizontal stabilizer actuator Examine. Operational test.		
Chapter 28 - Fuel			
1	Auxiliary fuel pump Operational test.		
2	Underwing fuel pumps. (if installed) Operational test.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
Chapter 30 - Ice and Rain Protection			
1	Pitot tube and static port heaters Operational test.		
Chapter 31 - Indicating/Recording			
1	Instrument panel shockmounts Examine.		
2	Instruments Examine.		
3	Annunciator panel Examine.		
Chapter 33 - Lights			
1	Navigation lights Examine. Operational test.		
2	Anti-collision strobe lights or beacons Examine. Operational test.		
3	Landing lights Examine. Operational test.		
4	Cockpit lights Examine. Operational test.		
5	Instrument lights Examine. Operational test.		
6	Warning lights Examine. Operational test.		
7	Passenger cabin lights Examine. Operational test.		
Chapter 34 - Navigation			
1	Pitot tube Examine.		
2	Static ports Examine.		
3	Pipes - pitot, static and vacuum Examine.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
4	Vertical speed indicator Reset to zero.		
5	Airspeed indicator Check, calibrate if necessary.		
6	Gyro operated instruments Operational test.		
7	Magnetic compass Check correction card date validity _____		
Chapter 37 – Vacuum			
1	Vacuum system suction regulator Clean filter.		
2	Vacuum system Examine. Replace if air filter is contaminated.		
3	Vacuum system pressure regulator Examine.		
4	Vacuum system ejector Examine.		
E. AVIONICS **Chapter 23 - Communications and Chapter 34 - Navigation**			
1	Antennas Examine.		
2	Headsets and microphones Clean. Examine.		
3	Avionic equipment Examine.		
4	Avionic connectors and cables Examine.		
5	Avionic equipment racks and shock mounts Examine.		
6	All Avionics systems Examine switches and circuit breakers.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
7	All Avionics systems Operational test.		

PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Ref. AMM Pilatus Porter Chapter 28-20-00
FUEL DISTRIBUTION SYSTEM – ADJUSTMENT TEST

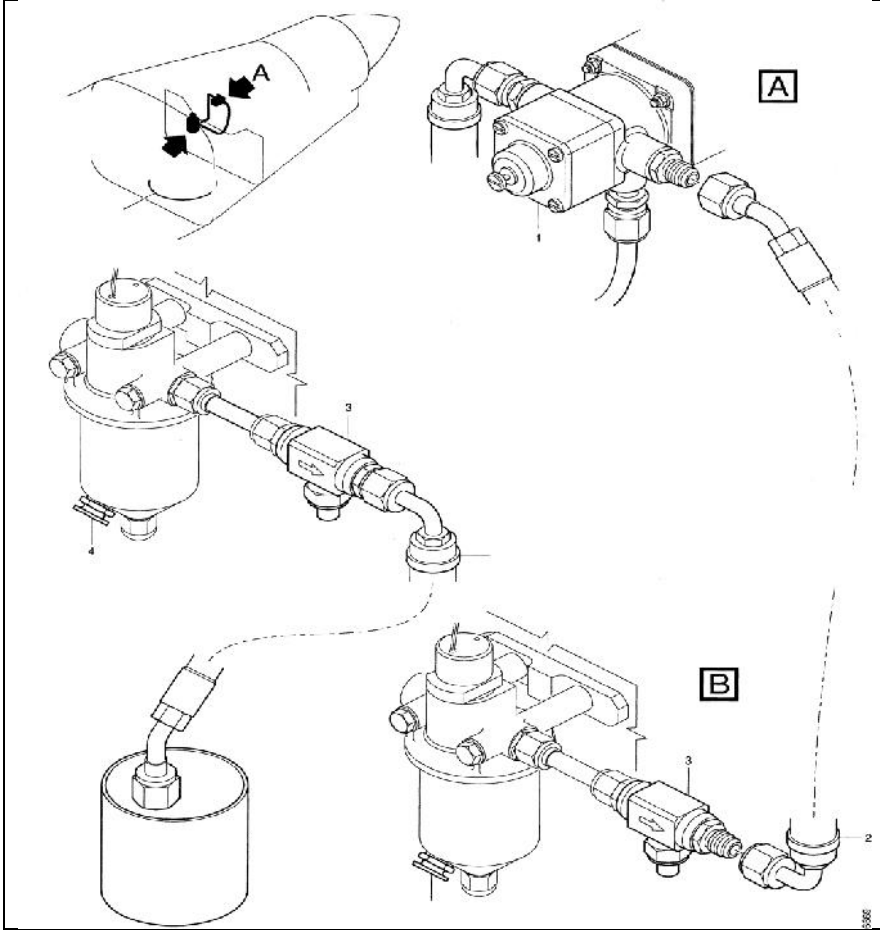
Reg. Mark	:	PK -	Date	:	
MSN	:		Station	:	
TSN / CSN	:		WO No.	:	

NO	TASK	SIGNATURE	
		SIGN	STAMP
Tools and Equipment Part No. — Description — Remarks — — Stopwatch — — Fuel container with measured graduation in liters or US gals — Minimum capacity 40 liters (10 US gals)			
Procedure **A. Job Set Up**			
1	Make sure that the aircraft is tall down ($10^{o} \pm 1^{o}$ nose up).		
2	Make sure that the aircraft is refueled to maximum (Ref. 12-11-28, page Block301).		
3	Set the fuel-system valve lever to CLOSED.		
4	Open the fuel-filter access panel PB3.		
5	Open the engine access panel PL1.		
B. Preparation			
1	Put the fuel container below the fuel filter.		
2	Open the filter drain valve (4) and let the fuel drain.		
3	When the flow of fuel stops, close the drain valve (4).		
4	Remove the outlet hose (2) from between the fuel flow transmitter (3) and theEngine Driven Pump (EDP) (1).		

NO	TASK	SIGNATURE	
		SIGN	STAMP
5	Turn the outlet hose through 180 degree, then install the outlet hose (2) to the fuelflow transmitter (3) with the other end through the access panel PB3.		
6	Put the fuel container below the disconnected end of the outlet hose (2). Do notextend the length of the hose for the test.		
7	Set the fuel-system valve-lever to OPEN until you get a constant flow of fuel in tothe container.		
8	Set the fuel-system valve-lever to CLOSED and empty the container.		
C. Gravity Flow System			
1	Put the fuel container below the disconnect end of the outlet hose.		
2	Set the fuel-system valve-lever to OPEN for 5 minutes.		
3	Set the fuel-system valve-lever to CLOSED.		
4	Make sure that there is not less than 6,95 liters (1,84 US gals) of fuel in thecontainer.		
5	Empty the container.		
D. Auxiliary Fuel Pump System			
1	Put the fuel container below the disconnected end of the outlet hose		
2	Energize the aircraft electrical system.		
3	Set the AUX F PUMP switch to ON and immediately set the fuel-system valvelever to OPEN.		
4	After 5 minutes, set the fuel-system valve lever to CLOSED and the AUX F PUMPswitch to OFF.		
5	De-energize the aircraft electrical system.		
6	Make sure that there is not less than 22,1 liters (5,84 US gals) of fuel in thecontainer.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
7	Empty the container.		
E. Close Up (Ref. Fig. 501)			
1	Remove the outlet hose (2) from the fuel flow transmitter (3).		
2	Install the outlet hose (2) between the fuel flow transmitter (3) and the EDP (1)		
3	Energize the aircraft electrical system.		
4	Set the fuel-system valve lever to OPEN.		
5	Set the AUX F PUMP switch to ON.		
6	Do leak checks at the outlet hose (2) and the tee adapter (3) connection. No leaksare permitted.		
7	Set the AUX F PUMP to OFF.		
8	Set the fuel-system valve lever to CLOSED.		
9	De-energize the aircraft electrical system.		
10	Discard the fuel as given in the local regulations.		
11	Make sure that the work area is clean and clear of tools and other items.		
12	Close the fuel-filter access panel PB3 and the engine access panel PL1.		

NO	TASK	SIGNATURE	
		SIGN	STAMP



PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

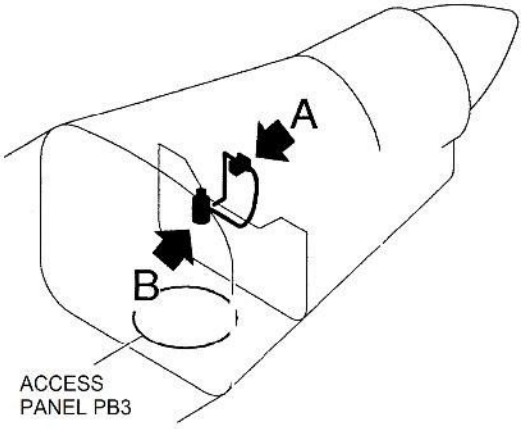
The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

Ref. AMM Pilatus Porter Chapter 28-40-00
FUEL INDICATING SYSTEM – ADJUSTMENT/TEST

Reg. Mark : PK -		Date :	
MSN :		Station :	
TSN / CSN :		WO No. :	

NO	TASK	SIGNATURE	
		SIGN	STAMP
Tools and Equipment Part No. — Description — Remarks — — Stopwatch — — Fuel container with measured graduation in liters or US gals — Minimum capacity 40 liters (10 US gals)			
Procedure **A. Job Set Up**			
1	Make sure that the aircraft is tail down ($10^{o} \pm 1^{o}$ nose up)		
2	Make sure that the aircraft is refuelled to maximum (Ref. 12-11-28, Page Block 301)		
3	Set the fuel-system valve-lever to CLOSED		
4	Open the fuel-filter access-panel PB3. 		
5	Open the engine acces panel PL1		
B. Preparation (Ref. Fig. 1)			
1	Put the fuel container below the fuel filter.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
2	Open the filter drain valve (4) and let the fuel drain.		
3	When the flow of fuel stops, close the drain valve (4).		
4	Remove the outlet hose (2) from between the fuel flow transmitter (3) and the Engin**eDriven Pump (EDP) (1).**		

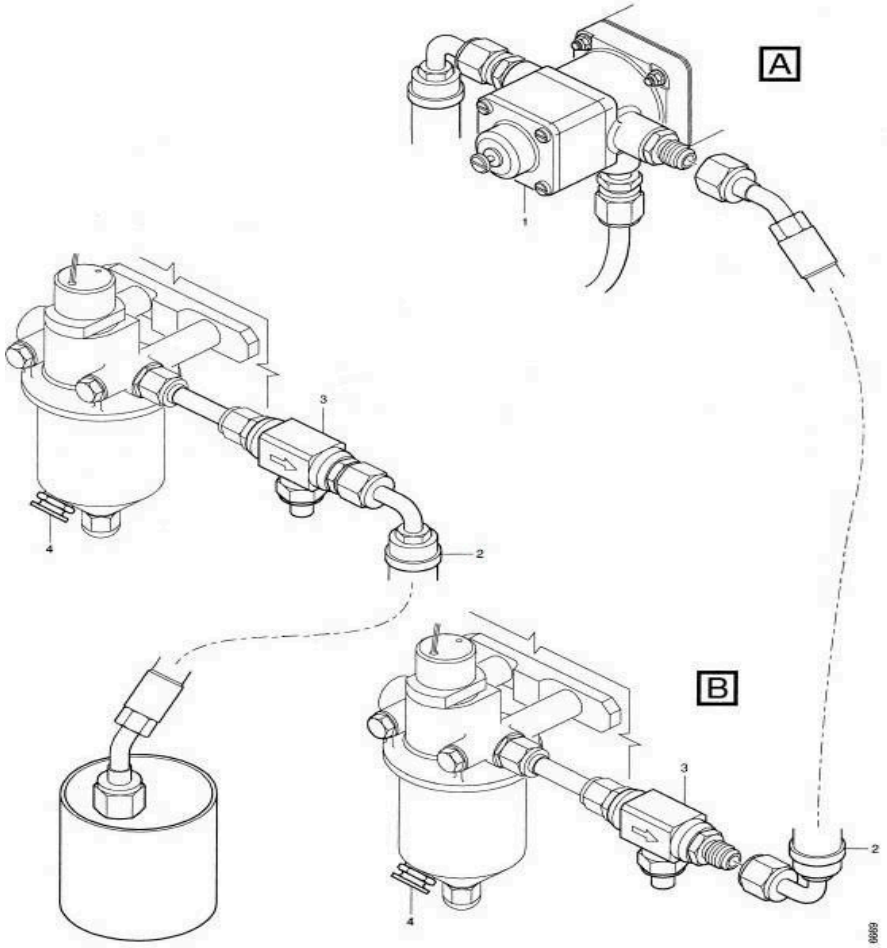


Figure 1 fuel Disrtibution – Adjusment/Test

5	Turn the outlet hose through 180 degrees, then install the outlet hose (2) to the fuelflow transmitter (3) with the other end through the access panel PB3.		
6	Put the container below the disconnect end of the outlet hose (2). Do not extend thelength of the hose for the test.		
7	Energize the aircraft electrical system.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
8	Set the fuel-system valve-lever to OPEN until you get a constant flow of fuel into thecontainer.		
9	Set the fuel-system valve-lever to CLOSED and empty the container.		
10	Reset the fuel used totalizer to zero.		
C. Low Fuel Flow Indication Check (Ref. Fig. 1)			
1	Set the fuel-system valve-lever to OPEN.		
2	When the fuel flow has stabilized, record the fuel flow indication. Fuel flow: _____		
3	After 5 minutes set the fuel-system valve-lever to CLOSED		
4	Record the fuel used from the fuel used totalizer. Fuel used: _____		
5	Measure the quantity of the fuel in the container. Quantity: _____		
6	Make sure the difference between the fuel used totalizer indication and the quantity offuel in the container is not more than ± 1 liter (± 0.26 US gals).		
7	Calculate the fuel rate for 1 hour as follows: - Quantity of fuel in container after 5 minutes x 12 = Flow rate/hour.		
8	Compare the calculated fuel flow with the actual fuel flow recorded at step (2): - The difference between the calculated fuel flow and the actual fuel flow must bethe same ± 12,5 liter/hour (± 3,3 gals/hour)		
9	If the difference between the actual and calculated fuel flow is out of limits, replacethe indicators, signal conditioner or fuel flow transmitter. **NOTE: IT IS RECOMMENDED THAT THE FUEL FLOW TRANSMITTER IS REPLACED FIRST.**		
10	Empty the container.		
D. High Fuel Flow Indication Check Using the Auxiliary Fuel Pump System (Ref. Fig. 1)			
1	Set the AUX F PUMP switch to ON.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
2	Move the fuel-system valve-lever towards the CLOSED position to get an indicated fuel flow of between 189 and 285 liters/hour (50 and 75 US gals/hour). Record the indicated fuel flow. Fuel flow: _____ **NOTE: TO MAKE THE TEST EASIER, YOU CAN MAKE AN ORIFICE TO GET THECORRECT FUEL FLOW. USE A METAL BLANK DRILLED WITH A 2,4 mm (3,32 in) HOLE INSTALLED IN THE END OF THE DISCONNECTED OUTLET HOSE (2). THIS WILL GIVE THE REQUIRED FUEL FLOW WITH THE FUEL-SYSTEM VALVE-LEVER FULLY "OPEN".**		
3	When the fuel flow is correct and constant, get a second person and do these steps atthe same time. Reset the fuel totalizer. Record the quantity of fuel in the container. Start the stopwatch.		
4	After 5 minutes, set the fuel-system valve-lever to CLOSED and the AUX F PUMPswitch to OFF.		
5	Record the fuel used from the fuel used totalizer. Fuel flow: _____		
6	Measure the quantity of the fuel in the container and subtract the quantity recorded inStep (3). Quantity: _____		
7	Make sure the difference between the fuel used totalizer indication and the quantity offuel in the container is not more than ± 1 liter (±0,26 US gals).		
8	Calculate and record the flow rate for 1 hour as follows: Quantity of fuel measured at step (6) x 12 = Fuel flow rate /hour.		
9	Make sure that the flow rate recorded at step (8) is not more than ± 12,5 liters/hour(±3.3 US gals/hour) different to the fuel flow indication recorded in Step (2).		
10	If the difference between the actual and calculated fuel flow is out of limits, replacethe indicators, signal conditioner or fuel flow transmitter. NOTE: IT IS RECOMMENDED THAT THE FUEL FLOW TRANSMITTER ISREPLACED FIRST.		
E. Close Up (Ref. Fig. 1)			
1	If necessary, remove the orifice from the outlet hose (2).		
2	Remove the outlet hose (2) from the fuel flow transmitter (3).		

NO	TASK	SIGNATURE	
		SIGN	STAMP
3	Install the outlet hose (2) between the fuel flow transmitter (3) and the EDP (1).		
4	Energize the aircraft electrical system.		
5	Set the fuel-system valve lever to OPEN.		
6	Set the AUX F PUMP switch ON.		
7	Do a leak check at the outlet hose (2) and the tee adapter (3) connection.No leaks are permitted.		
8	Set the AUX F PUMP switch to OFF.		
9	De-energized the aircraft electrical system.		
10	Discard the fuel in accordance with local regulation.		
11	Make sure that the work area is clean and clear of tools and other items.		
12	Close the fuel-filter access panel PB3.		
13	Close the engine access panel PL1.		

PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

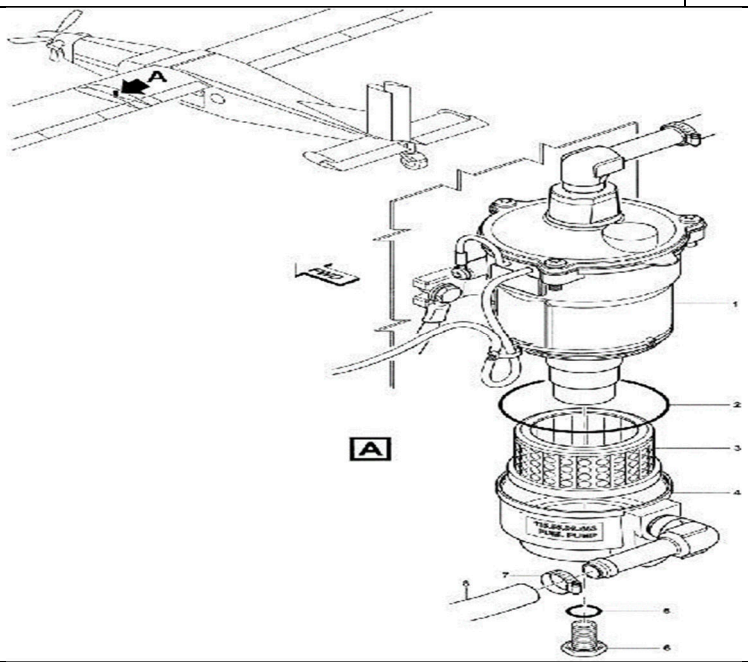
Ref. AMM Pilatus Porter Chapter 28-15-00

FUEL SYSTEM UNDERWING TANK INSPECTION TRANSFER PUMP FILTERS

Reg. Mark	:	PK -	Date	:	
MSN	:		Station	:	
TSN / CSN	:		WO No.	:	

NO	TASK	SIGNATURE	
		SIGN	STAMP
Tools Equipment for Aircraft with Underwing Fuel system and Fuel Transfer Pump. P/N: 115.55.06.443 Regulated Air supply			
Expendable Parts **Part no:** / **Description** / **Fig. Item no.** 968.84.30.305 / Filter (if required) / Fig 701, item 3 946.91.27.355 / O-ring / Fig 701, item 5 968.84.30.309 / O-ring / Fig 701, item 2			
WARNING: OBEY THE SAFETY PRECAUTIONS GIVEN IN 28-00-00, PAGE BLOCK 201, WHEN YOU DO WORK ON THE FUEL SYSTEM.			
1	Open and install the safety clip to the circuit breaker **EXT FUEL**.		
2	Remove the Access panel LB7, LB9, RB7 and RB8.		
3	Loosen the clamp (item 7) and disconnect the tube (item 8) from the elbow of thepump inlet.		
4	Remove the bolt (item 6), the bowl (item 4), the O-Ring (item 2) and filter (item 3)from the pump (item 1). **WARNING:** MAKE SURE YOUR HANDS ARE CLEAN BEFORE YOU CLEAN THE FILTER (3)DO NOT USE COTTON OR CLOTH TO CLEAN THE FILTER COTTON CAN CONTAMINATE THE FILTER.		
5	Use a regulated air supply to blow through from the inside of the filter (item 3) toremove unwanted material.		
6	If the filter (item 3) is damaged or cannot be cleaned, discard the filter and install anew one.		
7	Remove and discard the O-Ring (item 5) from the bolt (item 6) and install a new O-ring (item 5).		

NO	TASK	SIGNATURE	
		SIGN	STAMP
8	Put the filter (item 3), new O-ring (item 2) and bowl (item 4) in position and installthe bolt (item 6).		
9	Connect the tube (item 8) to the elbow of the pump outlet and tighten the clamp(item 7).		
10	Close the circuit breaker EXT FUEL.		
11	Energizer the aircraft electrical system (Ref. AMM 24-40-00, Page Block 1).		
12	Set the NORMAL-EMERG switch to EMERG and check that the pump operates.		
13	Check the pump for leaks. No leaks are permitted.		
14	Set the NORMAL-EMERG switch to NORMAL.		
15	Remove the electrical power from the aircraft (Ref. AMM 24-40-00, Page Block 1).		
16	Make sure that the work area is clean and clear of tools and other item.		
17	Install the access panel LB7, LB9, RB7 and RB8.		



PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

WHEEL AND BRAKES INSPECTION SHEET OF PILATUS PORTER PC6

Reg. Mark : PK - Date :

MSN : Station :

TSN / CSN : WO No. :

NO	TASK	SIGNATURE	
		SIGN	STAMP
1	Perform Detail Visual Inspection with Flash Light, Mirror, and Magnifying Glass ofthe Brake Pedals and System for Cracks, Corrosion, and Security of Installation.		
2	Inspect Wheel and Brakes IAW ATA 32-40-00 and BERINGER Time Limits /Maintenance Checks MC-STC-002.		
3	Inspect hydraulic brake fluid reservoir, check brake fluid level, apply brakes,examine system for leaks, and service with MIL-PRF-5606 (ROYCO 756) hydraulic fluid as required.		
4	Inspect tire condition IAW ATA 12-14-32 and Michelin Aircraft Tire Care and Service Manual (Michelin Service Manual can be used as a guide line for all approved main tires but will not supersede manufacture inspection recommendations)		
5	Check Brake Disc Thickness Record_____mm/inch. Minimum brake disc thickness **0.252 inch / 6.4 mm.**		
6	Examine brake disc condition for Coning, Groove and Bumps. See figure 2 asattached.		
7	Inspect Brake Pad for wear. Brake Pad must be changed before grooves are invisible. See figure 3 as attached. Friction material on Brake Pad minimumthickness **0.100 inch / 2.5 mm.**		
8	Check play between disc and key disc drive. Max play **0.024 inch / 0.6 mm**. Seefigure 3 as attached.		
9	Check Main wheels Tire. Examine and check inflation pressure 3,3 bar (**49 psi**).		
10	Check Tail wheel. Examine and check for installation and inflation pressure 2,2bar (**47 psi**).		
11	Check and examine brake master cylinders for leaks and connections.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
12	After complete installation, check disc safety wire. Safety wire (**0.041**) must be inplace to prevent disc from sliding out the slots. See figure 1 as attached.		
13	Record both Main Wheel Tire: S/N LH_____and Hub S/N_____ S/N RH_____and Hub S/N_____		
14	Record Tail Wheel Tire S/N _____Hub S/N_____		

PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

BERINGER
Aeropole, 05130 TALLARD - FRANCE
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e-mail : contact@beringer-aero.com

TIME LIMITS / MAINTENANCE CHECKS

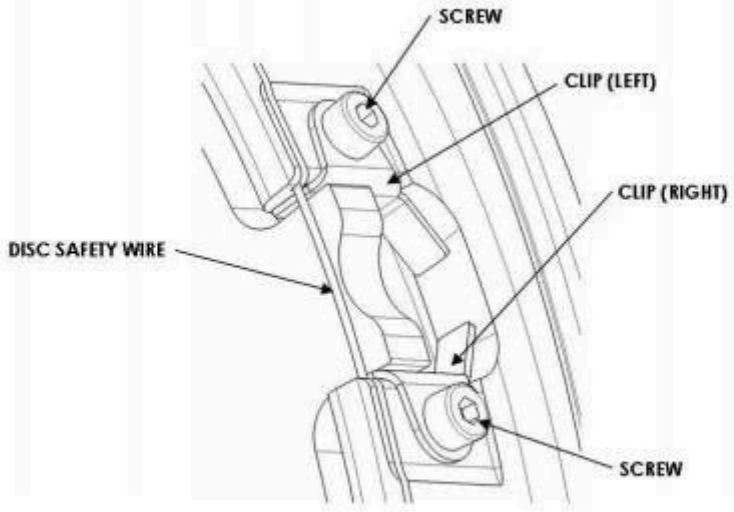
Manuel référence : BRG-ALTP-02
Référence document : MC-STC-002

2. Scheduled maintenance checks

2.1. Flight maintenance checks

Next flight maintenance checks are in addition to PC-6 maintenance manual.

Additional flight maintenance checks	Preflight inspection
Component	**Operation**
Safety wire of brake disc	Visual inspection
Brake pads	Inspect for wear and damage



CAUTION: Disc safety wire must be in place, it prevents disc from sliding out the slots.

FIGURE 1

BERINGER
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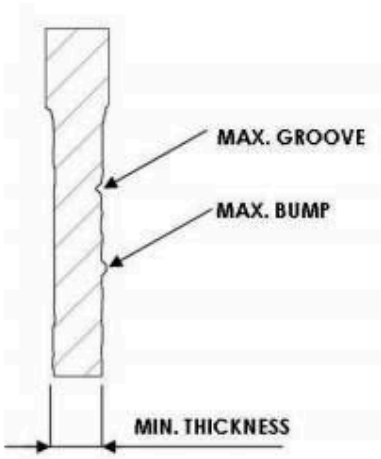
e-mail : contact@beringer-aero.com

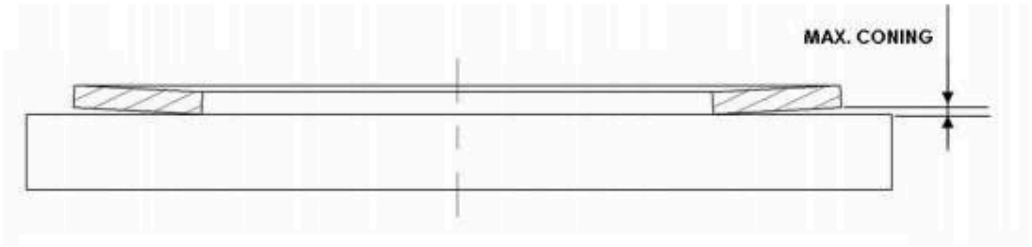
TIME LIMITS / MAINTENANCE CHECKS

Manuel référence : BRG-ALTP-02
Référence document : MC-STC-002



DISC WEAR LIMITS:

Min. Thickness DSC-011	6.4mm	0.252 in
Min. Thickness DSC-011.2	~~7.0mm~~	~~0.275 in~~
	6.4mm	0.252 in
Max. Coning	0.3mm	0.012 in
Max. Groove	0.2mm	0.008 in
Max. Bump	0.2mm	0.008 in



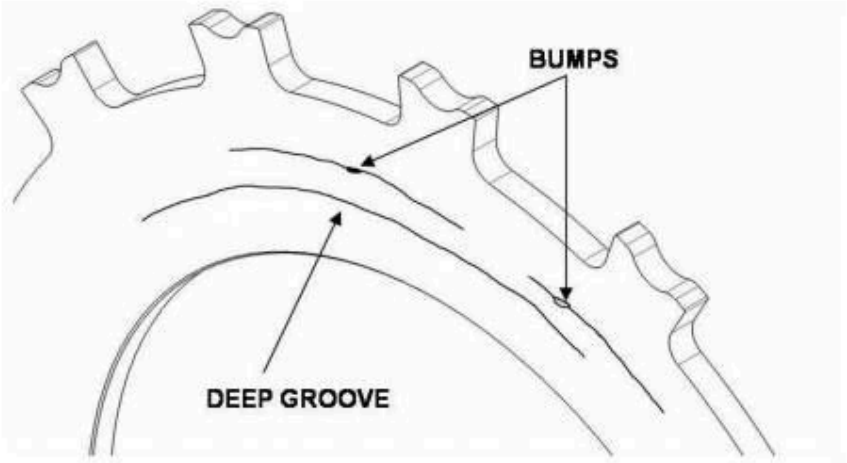


FIGURE 2

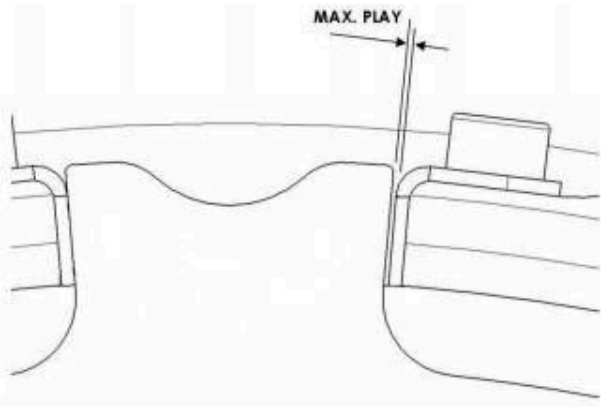
BERINGER
Aeropole, 05130 TALLARD - FRANCE
Tel:+33 (0)4 92 20 16 19 Fax:+33 (0)4 92 52 69 66
e-mail : contact@beringer-aero.com

TIME LIMITS / MAINTENANCE CHECKS

Manuel référence : BRG-ALTP-02
Référence document : MC-STC-002

CLIP WEAR LIMITS:

Max. Play	0.6mm	0.024 in



PAD WEAR LIMITS:

Min. Thickness	groove nearly invisible Friction material min. thickness 2.5mm (0.100 in)

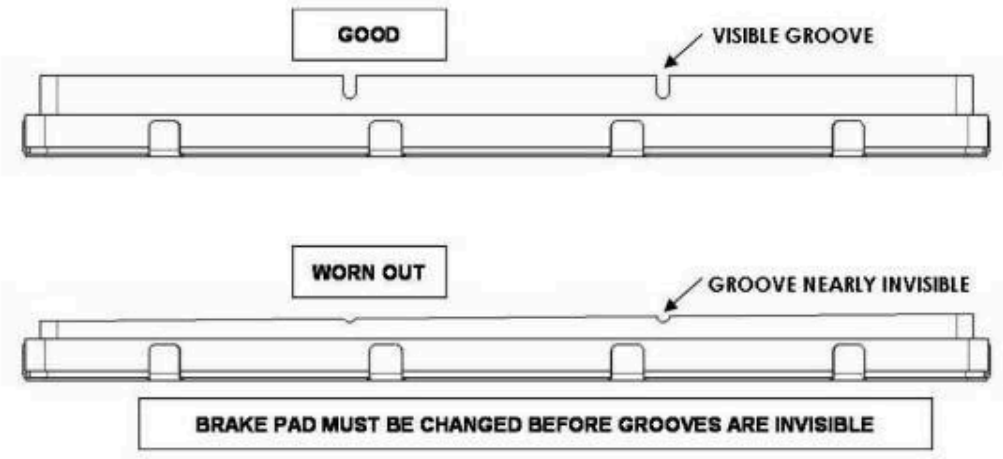


FIGURE 3

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/006-SNB/XII/2021	22-Dec-21	REPLACEMENT	PK-SNB/1015
5. CARD #	6. ATA SPEC	7. TRADE	8. STA
001	73		
9. ZONE	10. PANEL	-	

11. DESCRIPTION

PERFORM FUEL NOZZLE REPLACEMENT

REFERENCE	☒ AMM Ch. 73-10-05	☐	☐ OTHER
RII (*)	☐ Y	☒ N	MHR :

12. RESULT				MECH	ENG	INSP (*)
Performed at A/C TT : ..... A/C TC /LDG :						
FINDING	☐ Y	☐ N	ACT MHR :	DATE/TIME (DD/MM/YY)		
INSPECTION CARD (IC) #						

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS

Instructions (Note Post SB 1372 Engines Sheet).
Each Listed Inspection Item is to be performed in accordance with the P & W Maintenance Manual 3013242 Chapter 73-10-05 Latest Revision and any other applicable publications

FUEL NOZZLE CHANGE WORK SHEET PT6A-27

Date Performed :		**Engine S/N :**	
Removed from Aircraft :		**Engine TSN :**	
Aircraft Total Hours :		**Engine TSO :**	
Aircraft Total Cycle :		**Work Order Number :**	

NO	TASK	SIGNATURE	
		SIGN	STAMP
Procedure Before Removal			
1	To ease accessibility to the transfer tubes and manifold adapters adjacent to the spark igniters, disconnect the ignition leads (ref EMM 74-20-00/74-20-01). Release the ignition lead loop clamps from the support brackets at the center fire seal lower attachment brackets and move the leads clear. Install blanking caps on the spark igniters and lead connectors.		
2	Disconnect both lines from the fuel inlet adapter or flow divider and install blanking caps.		
Removal of Fuel Manifold Adapters			
	Note: The following procedure itemizes a removal sequence commencing with the No 8 fuel manifold inlet adapter which is best achieved by consideration of the Nos. 7, 8 and 9 adapters as a group. The procedural sequence may be modified by the operator as convenient for adapters at other locations.		
1	Using a suitable dye maker (PWC05-027) or (PWC05-046), number the position of each manifold adapter to identify its original position. (Refer to Figure 201) and toaid detecting hot section damage		
2	Remove bolts securing transfer tube locking plate (6, Fig 202) and inlet manifoldadapter (item 2) to gas generator case. Remove locking plate (item 6).		
3	Remove bolts securing locking plates (item 6) to the primary and secondary manifold adapters (item 5) adjacent to the inlet manifold adapter (item 2). Remove locking plates (item 6).		
4	Hold all the three adapters, then move the interconnecting fuel transfer tubes (item 1) into adapter bores (item 5). Use the puller (PWC54246) or pusher (PWC32366)to move the fuel transfer tubes in a clockwise direction, away from the inlet manifold adapter bores.		
5	Remove the inlet manifold adapter (item 2) (with flow divider and dump valve (item 14) installed). Use the pusher (PWC32366) or puller (PWC54246) to remove thefuel transfer tubes (item 1) from the adjacent adapters (item 5). Remove and discardpreformed packings (item 13) from the transfer tubes.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
6	Remove the gasket (item 4) from the sheath (item 3) on the inlet manifold adapter(item 2).		
7	Remove the remaining manifold adapters (5) progressively from the gas generator case by removing bolts and locking plates (6). As each adapter is removed, withdraw interconnecting fuel transfer tubes (1). Remove and discard preformed packing's (13). Remove metal gaskets (4) from sheaths (3) on the adapters. **CAUTION: DO NOT PRY THE SHEATHS OFF WITH A SCREWDRIVER.**		
8	Remove the sheaths (item 3) from the manifold adapters using puller (PWC30416) ifsheath is tight fitting on adapter boss.		
9	Straighten the lugs on the key washers (item 7) and remove the nozzle assemblies(item 8) from the adapters (item 2) and (item 5). Discard the keywashers.		
10	Note the relative angle of each elbow to the inlet manifold adapter (item 2) (starting control installation only) and remove the elbows (Item 9) and locknuts (item 10). Discard preformed packing's (item 11) and back up rings (item 12)		
11	To prevent exposure to dust and dirt, place manifold components in clean, and putin special boxes provided, with the removal positions identified in the box.		
12	Send rejected assemblies for repair or overhaul in original packaging to preventparts contacting each other during shipment.		

Removed Parts Record. (Post SB1372 worksheet)

Part Number	Description	Serial Number/s	Qty	Remarks/ Findings
	Secondary Inlet Adapter Manifold		1	
	Primary Inlet Adapters		10	
	Secondary Inlet Adapters		3	
	Fuel Nozzle Sheaths		14	

Installation of Fuel Manifold Adapters

	CAUTION: EXTREME CARE MUST BE EXERCISED WHEN HANDLING THE FULE NOZZLE ASSEMBLIES SINCE EVEN FINGERPRINTS ON THE ORIFICE MAY PRODUCE POOR SPRAY PATTERN, CLEAN LINT-FREE COTTON GLOVES OR SURGICAL GLOVES SHOULD BE WORN AT ALL TIMES WHEN HANDLING THESE PARTS.
	CAUTION: MAKE SURE TO CORRECTLY ENGAGE THE WRENCH SOCKET ON THE NOZZLE DURING INSTALLATION. HOLD THE STEM WHEN INSTALL THE NOZZLE TOPREVENT BENDING

NO	TASK	SIGNATURE	
		SIGN	STAMP
1	With a 10X magnifying glass verify that each manifold adapter assembly carries thecorrect detail fuel nozzle assembly tip part number		
2	Install the elbows (Item 9) on the inlet adapter (starting control installation only) inthe same positions as noted on removal		
3	Install the fuel nozzle (Item 8, Fig 202) in the fuel manifold adapters (Item 2 and 5)with new keywasher (item 7) at each location		
	Use engine oil (PWC03-001) and torque the nozzle assemblies 45 to lbf. In.		
	NOTE: Leak test and function test of each Nozzle and Adapter Assembly may perform OFF WING in House or approvedoverhaul/repair vendor as required. A company serviceable tag with a copy of the Approved Parts Tag FAA Form 8130-3, EASA formone or equivalent for new parts.		
4	**CAUTION:** DO NOT USE SHARP EDGE TOOLS TO BEND OR SET KEYWASHER TABS.		
	On completion of installation lock each key washer on respective nozzle assembly. Do not exceed specified torque to align flat on tip with key washer.		
5	Install the manifold adapters and fuel transfer tubes as follows.		
	NOTE: Primary fuel manifold adapters are identified by a single weld blob on the larger mounting flange. Other weld blobs appearing on the knuckle section of the adapters should be ignored. (SeeFig 201)		
	a) Assemble Sheaths (Item 3) on all adapters (Items 2 & Item 5). Make sure eachlocating pin engages hole in each sheath.		
	b) With Nozzle adapter and sheath pressed together by hand, check clearance between adapter and sheath flanges. Maximum gap allowed is 0.003 inch (Ref. Fig 203 upper drawing) Larger gap suggests either or both parts are distorted. Iffound Send distorted parts to parts P &W for repair.		
	c) Carefully check gap between fuel nozzle tip and side hole in sheath; clearance of 0.020 inch is required (Ref. Fig 203). If clearance is less at any point, either orboth parts are distorted. If found Send distorted parts to parts Pratt and Whitney for repair.		
	d) Lubricate and install preformed packings (13, Fig. 202) on all fuel transfer tubes (1) using a thin layer of engine oil (PWC03-001). Position fuel transfer tubes intoports on one side of manifold adapters (2 and 5). Fully insert fuel transfer tubes in their respective ports until the bottom of manifold adapter is reached.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
	CAUTION: TO AVOID POSSIBLE OF THE STAINLESS STEEL GASKETS, IT IS ADVISABLE TO SLIDE THE GASKET OVER THE SHEATH, CAREFULLY ALIGN THE HOLES AND INSERT BOTH BOLTS, THE WHOLE ASSEMBLY CAN THEN BE MATED WITH THE PADS ON THE GAS GENERATOR CASE TUS AVOIDING ANY ATTEMPT TO LEVER THE GASKETS INTO ALIGNMENT WITH THE BOLTS.		
	e) Position the Pre SB 1276/ Post SB 1276 gasket (Item 4) over the sheath (Item3) on the inlet manifold adapter (Item 2) and align the bolt holes or mounting studs. The sheath flange must be flat. NOTE: The gasket may be put on either side. For consistency, all the gasketsshould be installed with the flat side against the gas generator. NOTE: **Post-SB1167**: For engines with the conversion coated gas generatorcase only, lightly coat both faces of the gasket (4) with corrosion- preventive compound (PWC09-003).		
	f) Position the gaskets (4) over the sheaths (3) on the primary and secondarymanifold adapters (5) and align on the bolt holes.		
	g) Start with each side of the inlet manifold adapter (2). Install the remaining manifold adapters (5) and sheaths (3) on the gas generator case at the locations specified before. Use the pusher (PWC32366) or puller (PWC54246)to engage fuel transfer tubes (1) to interconnect with the adjacent adapter at each location.		
	h) Install locking plate (Item 6) and bolts to the gas generator case. Torque boltsfinger tight.		
6	When all remaining manifold adapters (5) are positioned, remove bolts or locknuts, as applicable, from inlet manifold adapter (2). Assemble the locking plate (6) and reinstall the bolts or locknuts.		
	CAUTION: MAKE SURE ALL 14 MANIFOLD ADAPTER LOCKING PLATES ARE CORRECTLY INSTALLED.		
	CAUTION: PRIOR TO TIGHTENING BOLTS OR LOCKNUTS, CHECK SEATING OF METAL GASKET RELATIVE TO SHEATH AND BOSS. TIGHTENING SHOULD BE DONE EVENLY ON EACH ADAPTER TO PROVIDE FULL SEATING POTENTIAL OF METAL GASKET.		
7	Tighten all adapter mounting bolts, in a sequence, 15 to 20 lb.in. Retighten 32 to 36lb. in. in the same sequence. **Secure bolts with lock wire.**		
	NOTE: After torqueing, a 0.001 in. gap is allowed between the adapter and sheath flanges (Ref. Fig. 203 lower drawing).		
8	Remove the blanking caps from the fuel delivery lines and connect the lines to theelbows. Tighten the coupling nuts 90 to 100 lb.in. and fasten with lockwire.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
9	Remove the blanking caps from the spark igniters and from harness leads and connect the leads to the igniters. Tighten the connections finger tight, plus 45 degrees and fasten with lock wire. Secure the ignition lead loop clamps to the support brackets at the center fire seal lower mounts and tighten the nuts 32 to 36 lb.in.		
10	Check function of fuel manifold installation (Ref. Adjustment/Test).		

Installed Parts Record. (Post SB1372 worksheet)

Part Number	Description	Serial Number/s	Qty	Remarks
	Secondary Inlet Adapter Manifold		1	
	Primary Inlet Adapters		10	
	Secondary Inlet Adapters		3	
	Fuel Nozzle Sheaths		14	

PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name : _____ Stamp : _____

Signature : _____ Place/Date : _____

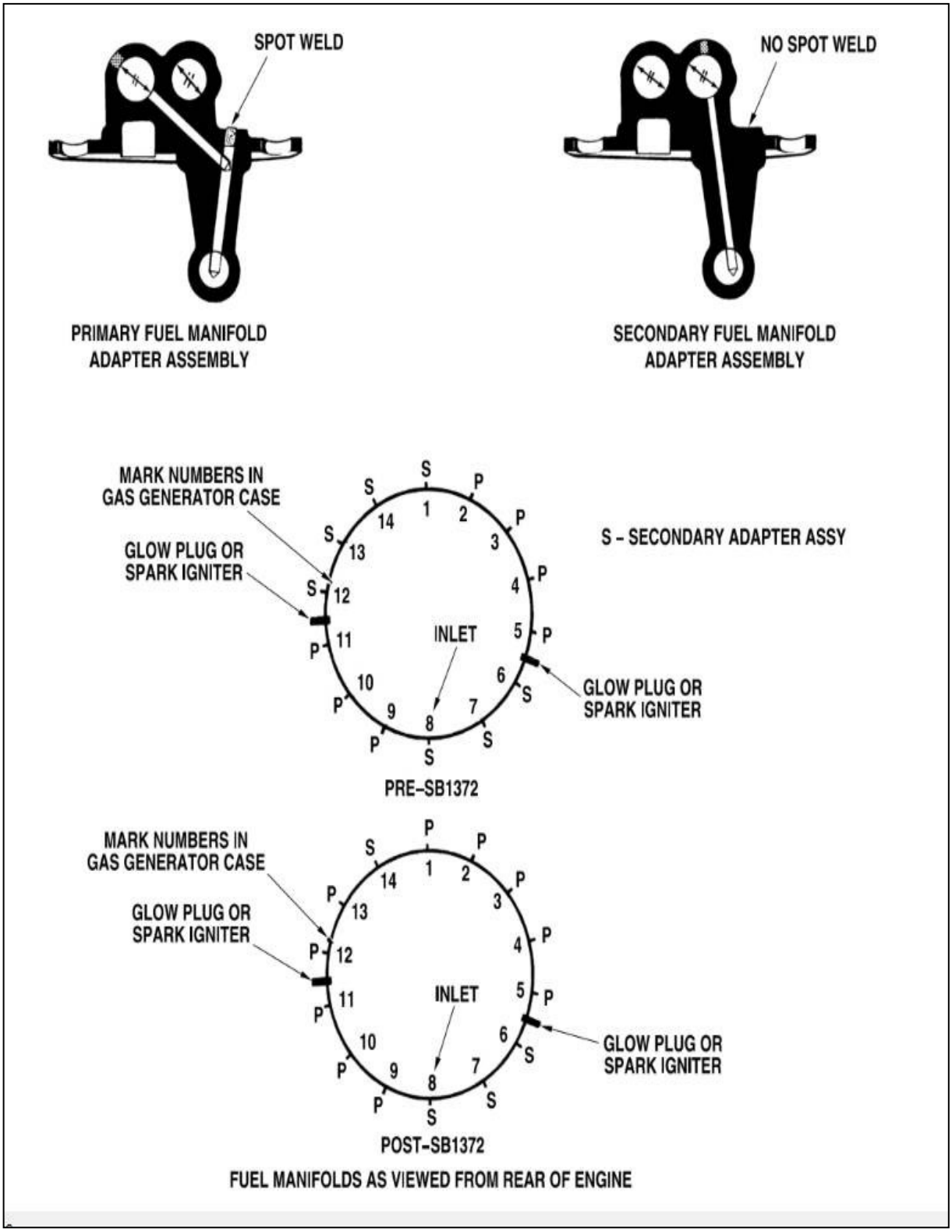


Figure 201 - Identification and Location of Fuel Manifold Adapters

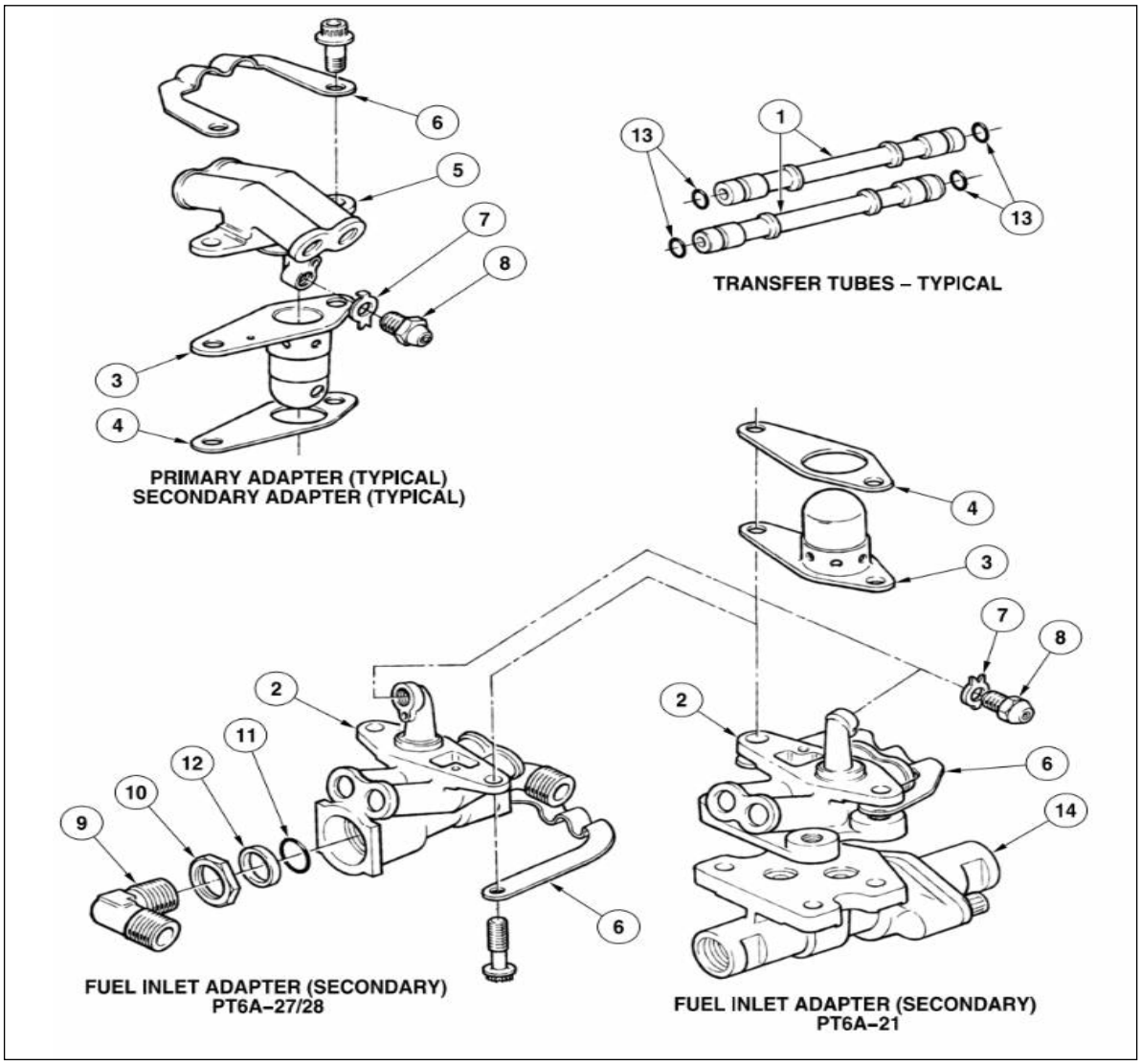


Figure 202 - Removal/ Installation of Fuel Manifold Adapter

Key to Figure 202

1. Fuel Transfer Tube
2. Fuel Manifold Inlet Adapter
3. Sheath
4. Gasket
5. Fuel Manifold Adapter
6. Lockplate
7. Keywasher
8. Fuel Nozzle
9. Elbow
10. Locknut
11. Preformed Packing
12. Back-up ring
13. Preformed Packing
14. Flow Divider and Dump or Purge Valve

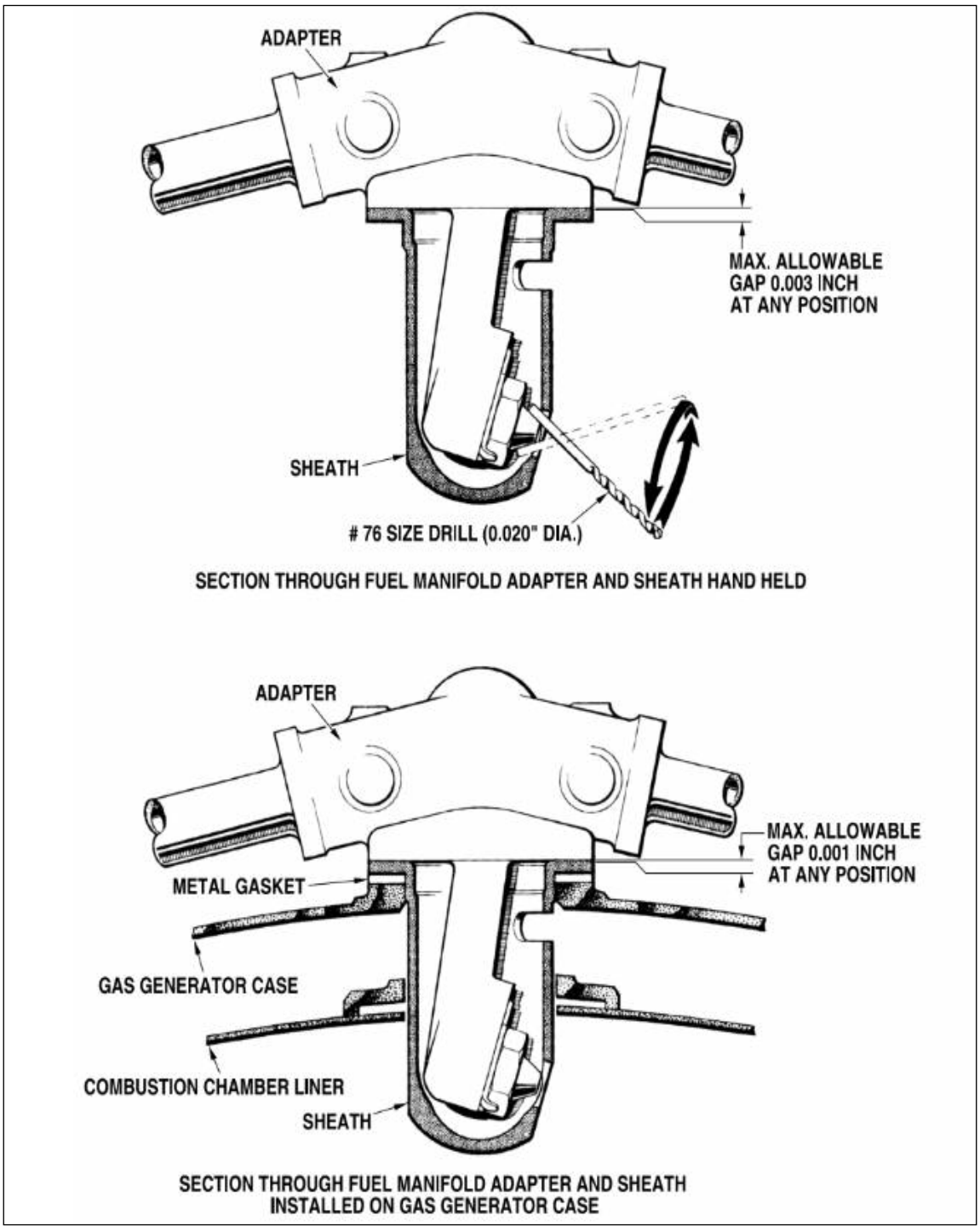


Figure 203 Fuel Manifold Installation - Clearance Checks

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/006-SNB/XII/2021	22-Dec-21	INSPECTION	PK-SNB/1015
5. CARD #	6. ATA SPEC	7. TRADE	8. STA
002	72		
9. ZONE	10. PANEL	-	

11. DESCRIPTION

PERFORM BORESCOPE HOT SECTION INSPECTION

REFERENCE	☒ EMM Ch. 72-00-00	☐	☐ OTHER
RII (*)	☐ Y	☒ N	MHR :

12. RESULT				MECH	ENG	INSP (*)
Performed at A/C TT : ..... A/C TC /LDG :						
FINDING	☐ Y	☐ N	ACT MHR :	DATE/TIME (DD/MM/YY)		
INSPECTION CARD (IC) #						

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS

Engine Borescope Hot Section Inspection

Work No: _____

Engine Serial Number	Date	Base / Location	Aircraft Registration
Aircraft Total Time	**Aircraft Total Cycle**	**Reason For Borescope**	

Note:
Record any discrepancies found during inspection, and/or take photographic evidence.
If None, then write No Findings. If you find defects, please quote EMM (Engine Maintenance Manual) Reference and Limitations.

Item	Work Description	SIGN	STAMP
1	Remove fuel manifold adapter as necessary (Ref. 73-10-05).		
2	Perform inspection of the First Stage Compressor. Defects: ……. If defects found, quote MM Limitation and References :		

Photo of First Stage Compressor 1st Quadrant	Photo of First Stage Compressor 2nd Quadrant
Photo of First Stage Compressor 3rd Quadrant	Photo of First Stage Compressor 4th Quadrant

		SIGN	STAMP
3	Perform inspection of Combustion Chamber Liner Assembly. Defects: …… If defects found, quote MM Limitation and References :		

Photo of Combustion Chamber 1st Quadrant	Photo of Combustion Chamber 2nd Quadrant
Photo of Combustion Chamber 3rd Quadrant	Photo of Combustion Chamber 4th Quadrant

		SIGN	STAMP
4	Perform Inspection of CT-Stator assembly. Defects: ……... If defects found, quote MM Limitation and References :		

Photo of CT Stator 1st Quadrant		Photo of CT Stator 2nd Quadrant	
Photo of CT Stator 3rd Quadrant		Photo of CT Stator 4th Quadrant	
5	Perform inspection of CT blades and shroud segments. Defects: …….. If defects found, quote MM Limitation and References :	SIGN	STAMP
Photo of Leading Edge CT-Blades 1st Quadrant		Photo of Leading Edge CT-Blades 2nd Q	

Photo of Leading Edge CT-Blades 3rd Quadrant		Photo of Leading Edge CT-Blades 4th Quadrant	
6	Perform inspection Trailing Edge CT-Blades Defects: …….. If defects found, quote MM Limitation and References :	SIGN	STAMP
Photo of Trailing Edge CT-Blades 1st Q		Photo of Trailing Edge CT-Blades 2nd Q	
Photo of Trailing Edge CT-Blades 3rd Q		Photo of Trailing Edge CT-B 4th Q	

7	Perform inspection of PT-Blades Defects: …….. If defects found, quote MM Limitation and References :	SIGN	STAMP

Photo of PT-Blades 1st Q	Photo of PT-Blades 2nd Q
Photo of PT-Blades 3rd Q	Photo of PT-Blades 4th Q

8	Install fuel manifold adapter(s) (Ref. 73-10-05).	SIGN	STAMP
9	Perform fuel leak check post fuel nozzle installation		

BORESCOPE PERFORMED BY

Name:_____ Signature : _____ Stamp :_____

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/006-SNB/XII/2021	22-DEC-2021	CHECK FOR WEAR	PK-SNB/1015
5. CARD #	6. ATA SPEC	7.TRADE	8. STA
003	24		
9. ZONE	10. PANEL	-	

11. DESCRIPTION

PERFORM CHECK FOR WEAR STARTER GENERATOR BRUSHES

REFERENCE	☒ AMM Ch. 24-31-11, Page block 601	☐	☐ OTHER
RII (*)	☐ Y	☒ N	MHR :

12. RESULT				MECH	ENG	INSP (*)
Performed at A/C TT : ..... A/C TC /LDG :						
FINDING	☐ Y	☐ N	ACT MHR :	DATE/TIME (DD/MM/YY)		
INSPECTION CARD (IC) #						

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/006-SNB/XII/2021	22-DEC-2021	CHECK OPERATION	PK-SNB/1015
5. CARD #	6. ATA SPEC	7. TRADE	8. STA
002	71		
9. ZONE	10. PANEL	-	

11. DESCRIPTION

PERFORM CHECK OPERATION OF PROPELLER OVERSPEED GOVERNOR

REFERENCE	☒ AMM Ch. 71-00-00	☐	☐ OTHER
RII (*)	☐ Y	☒ N	MHR :

12. RESULT				MECH	ENG	INSP (*)
Performed at A/C TT : ..... A/C TC /LDG :						
FINDING	☐ Y	☐ N	ACT MHR :	DATE/TIME (DD/MM/YY)		
INSPECTION CARD (IC) #						

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/006-SNB/XII/2021	22-DEC-2021	CHECK OPERATION	PK-SNB/1015
5. CARD #	6. ATA SPEC	7. TRADE	8. STA
005	24		
9. ZONE	10. PANEL	-	

11. DESCRIPTION

PERFORM CHECK OPERATION TEST OF EMERGENCY BATTERY

REFERENCE	☒ AMM Ch. 24	☐	☐ OTHER
RII (*)	☐ Y	☒ N	MHR :

12. RESULT				MECH	ENG	INSP (*)
Performed at A/C TT : ..... A/C TC /LDG :						
FINDING	☐ Y	☐ N	ACT MHR :	DATE/TIME (DD/MM/YY)		
INSPECTION CARD (IC) #						

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/006-SNB/XII/2021	22-DEC-2021	VISUAL INSPECTION	PK-SNB/1015
5. CARD #	6. ATA SPEC	7. TRADE	8. STA
006	57		
9. ZONE	10. PANEL	-	

11. DESCRIPTION

PERFORM VISUAL INSPECTION OF LEFT & RIGHT WING-STRUT FITTING

REFERENCE	☒ AMM Ch. 57-00-02	☐	☐ OTHER
RII (*)	☐ Y	☒ N	MHR :

12. RESULT				MECH	ENG	INSP (*)
Performed at A/C TT : ..... A/C TC /LDG :						
FINDING	☐ Y	☐ N	ACT MHR :	DATE/TIME (DD/MM/YY)		
INSPECTION CARD (IC) #						

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/006-SNB/XII/2021	22-DEC-2021	CHECK OPERATION	PK-SNB/1015
5. CARD #	6. ATA SPEC	7. TRADE	8. STA
007	28		
9. ZONE	10. PANEL	-	

11. DESCRIPTION

PERFORM CHECK OPERATION ENGINE DRIVEN FUEL PUMP - DRIVE SHAFT FOR BACKLASH

REFERENCE	☒ AMM Ch. 28-20-03	☐	☐ OTHER
RII (*)	☐ Y	☒ N	MHR :

12. RESULT				MECH	ENG	INSP (*)
Performed at A/C TT : ..... A/C TC /LDG :						
FINDING	☐ Y	☐ N	ACT MHR :	DATE/TIME (DD/MM/YY)		
INSPECTION CARD (IC) #						

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/006-SNB/XII/2021	22-DEC-2021	INSPECTION	PK-SNB/1015
5. CARD #	6. ATA SPEC	7. TRADE	8. STA
008	72		
9. ZONE	10. PANEL	-	

11. DESCRIPTION

PERFORM INSPECTION COMPRESSOR - EXAMINE FOR CORROSION AND EROSION

REFERENCE	☒ EMM Ch. 72-30-05	☐	☐ OTHER
RII (*)	☐ Y	☒ N	MHR :

12. RESULT				MECH	ENG	INSP (*)
Performed at A/C TT : ..... A/C TC /LDG :						
FINDING	☐ Y	☐ N	ACT MHR :	DATE/TIME (DD/MM/YY)		
INSPECTION CARD (IC) #						

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/006-SNB/XII/2021	22-DEC-2021	INSPECTION	PK-SNB/1015
5. CARD #	6. ATA SPEC	7.TRADE	8. STA
009	72		
9. ZONE	10. PANEL	-	

11. DESCRIPTION

PERFORM INSPECTION TURBINE - EXAMINE FOR CORROSION AND EROSION

REFERENCE	☒ EMM Ch. 72-30-05	☐	☐ OTHER
RII (*)	☐ Y	☒ N	MHR :

12. RESULT				MECH	ENG	INSP (*)
Performed at A/C TT : ..... A/C TC /LDG :						
FINDING	☐ Y	☐ N	ACT MHR :	DATE/TIME (DD/MM/YY)		
INSPECTION CARD (IC) #						

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/006-SNB/XII/2021	22-DEC-2021	FUNCTIONAL CHECK	PK-SNB/1015
5. CARD #	6. ATA SPEC	7.TRADE	8. STA
010	74		
9. ZONE	10. PANEL	-	

11. DESCRIPTION

PERFORM FUNCTIONAL CHECK OF SPARK IGNITORS OR GLOW PLUGS

REFERENCE	☒ AMM Ch. 74-20-00	☐	☐ OTHER
RII (*)	☐ Y	☒ N	MHR :

12. RESULT				MECH	ENG	INSP (*)
Performed at A/C TT : ..... A/C TC /LDG :						
FINDING	☐ Y	☐ N	ACT MHR :	DATE/TIME (DD/MM/YY)		
INSPECTION CARD (IC) #						

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS

NON ROUTINE CARD
(Form: SCA/MTC/047)

1. JO/WO #	2. DATE	3. MTC TYPE	4. A/C REG/MSN
WO/006-SNB/XII/2021	22-DEC-2021	LUBRICATE	PK-SNB/1015
5. CARD #	6. ATA SPEC	7. TRADE	8. STA
011	61		
9. ZONE	10. PANEL	-	

11. DESCRIPTION

PERFORM LUBRICATE OF PROPELLER HARTZELL

REFERENCE	☒ AMM Ch. 61-00-49	☐	☐ OTHER
RII (*)	☐ Y	☒ N	MHR :

12. RESULT				MECH	ENG	INSP (*)
Performed at A/C TT : ..... A/C TC /LDG :						
FINDING	☐ Y	☐ N	ACT MHR :	DATE/TIME (DD/MM/YY)		
INSPECTION CARD (IC) #						

13. PARTS REQUIRED

DESCRIPTION	PART NO	QTY	REMARK	
			STOCK	STATUS

14. TOOLS REQUIRED

DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS

	EMERGENCY EQUIPMENT LIST INSPECTION & MONITOR	**PT. SMART CAKRAWALA AVIATION** DEPARTMENT TEKNIK Form: **SCA/MTC/023**

DATE :	**A/C REG :**
A/C TYPE :	**CHECKER :** **SIGN:**

No.	Description	P/N	S/N	Next Insp.	Remarks
1	Pilot Life Vest				
2	Co-Pilot Life Vest				
3	Pax Life Vest				
4	Pax Life Vest				
5	Pax Life Vest				
6	Pax Life Vest				
7	Pax Life Vest				
8	Pax Life Vest				
9	Pax Life Vest				
10	Pax Life Vest				
11	Pax Life Vest				
12	Pax Life Vest				
13	Firt Aid Kit				
14	Crash Axe Installed				
15	Fire Extinguisher				
16	Life Raft (If Installed)				
17	Survival Kit (If Installed				
OTHERS					

smartaviation

Additional Work Sheet
100 Hours / Annual Inspection

Aircraft Registration: **PK-SNB**

WO# Nr: **WO/006-SNB/XII/2021**

Parts Used Sheet

Special Tool Used

No.	Description	Due Date	Engineer

Additional Work Sheet
100 Hours / Annual Inspection

Aircraft Registration: **PK-SNB**

WO# Nr: **WO/006-SNB/XII/2021**

Parts Used Sheet

Part Used

Date	Part Nr.	Serial Nr.	Description	Quantity	Engineer