



PT. SMART CAKRAWALA AVIATION

WORK ORDER

Form: SCA/MTC/030

Subject : Engine and Propeller Installation	No.	WO/012-SNB/XI/2022
	Date	8 Nov 2022
	A/C Reg.	PK-SNB MSN 1015
Reference: MP Pilatus PC-6 AMM Pilatus PC-6	Prepared By	TS
	Checked By	CI
	Approved By	TM
To : Engineer In Charge		
Description : 1. Perform Engine and Propeller Installation 2. Make an entry in Maintenance Log. 3. Return the Completed Work Order and Form to PPC. #If any finding, please close the routine card, and transferred to inspection card.		
Additional Work : - NIL -		
Compliance Statement CARRIED OUT ENGINE & PROPELLER INSTALLATION	Sign & Date Company Lic. No.: 02 JUNE 23 SCA 019 <i>(Signature)</i> (Engineer In Charge) AGUS SULAEMAN	Signature <i>(Signature)</i> (Technical Manager)

AIRCRAFT CHECK WORK SUMMARY
(Form: SCA/MTC/051)

DATE OF ISSUED	JO/WO #	TYPE OF MAINTENANCE	DATE OF ACCOMPLISHED		
8 Nov 2022	WOX12-SNB/XI/2022	Engine and Propeller Installation	02 JUNE 2023		
A/C Type		Mfg. Serial Number	A/C Registration		
PILATUS PORTER PC-6		1015	PK-SNB		
AIRCRAFT DATA					
Subject	Pos #	Serial Number (SN)	TTSN/TCSN		
Engine	#1	PCE-51153	TSN: 18599.8H CSN: 19230		
	#2	-	TSO: 0 CSO: 0		
Propeller/Rotor	#1	FY4093	TSN: 739.56H TSO: 0		
	#2	-			
Landing Gear	NLG				
	LH MLG				
	RH MLG				
PACKAGE COVERED					
No	Subject	Qty	Remark		
1	Non-Routine Card	-			
2	Inspection Card	-			
3	Work Order	1			
4	Summary Inspection List	1			
5	Material and Tool List	-			
6	Escalation form	-			
7	CRS (SMI / Unscheduled Maintenance)	1			
INSPECTION CARD (IC) LIST (Finding during maintenance)					
No	Taskcard Ref	Subject	Status		Name/ Sign & Stamp
			Open	Close	
<u>IC-001</u>					
<u>IC-002</u>					
<u>IC-003</u>					
<u>IC-004</u>					
<u>IC-005</u>					
<u>IC-006</u>					

<u>IC-007</u>					
<u>IC-008</u>					
<u>IC-009</u>					
<u>IC-010</u>					
<u>IC-011</u>					
<u>IC-012</u>					
<u>IC-013</u>					
<u>IC-014</u>					
<u>IC-015</u>					

Prepared by :
Technical Support



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Checked by :
Chief Maintenance



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Verified by :
Chief Inspector



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Yanuar

Approved by :
Technical Manager



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SUMMARY INSPECTION ITEMS
(Form: SCA/MTC/050)

WO Ref: WO/012-SNB/XI/2022

NO.	TASK CARD NO.	DESCRIPTION	DATE	EST MHR	NAME	STAMP
1	APPENDIX	PROPELLER INSTALLATION	02 JUNE 2023	48 MHR	AGUS SULAIMAN	
2	APPENDIX	ENGINE INSTALLATION	02 JUNE 2023	141 MHR	AGUS SULAIMAN	



SUMMARY INSPECTION ITEMS
(Form: SCA/MTC/050)

WO Ref: WO/012-SNB/XI/2022

NO.	TASK CARD NO.	DESCRIPTION	DATE	EST MHR	NAME	STAMP
1	APPENDIX	PROPELLER INSTALLATION	02 JUNE 2023	144 M	AGUS SULAEMAN	
2	APPENDIX	ENGINE INSTALLATION	02 JUNE 2023	144 MHR	AGUS SULAEMAN	



PT. SMART CAKRAWALA AVIATION

CERTIFICATE RETURN TO SERVICE
SCHEDULED MAINTENANCE INSPECTION
(CRS-SMI)

A/C TYPE : PILATUS PORTER PC-6
A/C REG : PK-SNB
MSN : 1015
TTSN : 476 : 14
TCSN : 580
DATE : 29 MAY 23

TYPE OF INSPECTION : ENGINE AND PROPELLER INSTALLATION
DUE AT : N/A
REF : MP PILATUS PC-6

EXCEPTION

-NIL-

AUTHORIZED PERSON

I hereby certify that this aircraft has been maintained accordance with CASR and Maintenance Program.
Aircraft safe and airworthy for flight

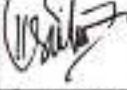
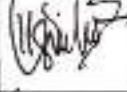
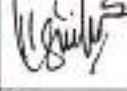
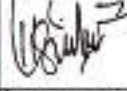
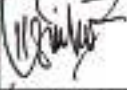
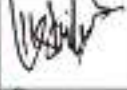
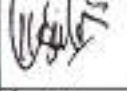
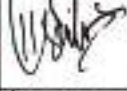
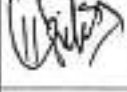
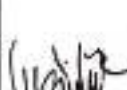
NAME	CAT	AMEL/OTR NO	SIGN&STAMP	DATE
AGUS SULAEMAN	AIRFRAME & POWER PLANT	6444 / SCA 019	 	02 JUNE 2023
	EIRA			

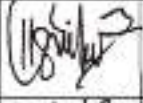
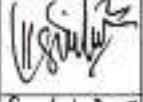
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DUE AT : 576 : 14 HOURS

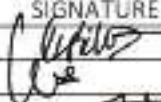
Form: SCA/MTC/049

		INSPECTION CARD (Form: SCA/MTC/ 048)			TECHNICAL DEPARTMENT	
1. CARD #	2. JO/WO #	3. ORIGINATOR	4. CARD REF	5. DATE		
6. A/C REG/MSN	7. A/C TYPE	8. TRADE	12. VENDOR ORDER #			
9. ZONE	10. STA	11. MTC TYPE				
13. DESCRIPTION/DEFECT-IF FINDING OF CPCP INSPECTION, PLEASE COMPLETE SET 20				14. PPC/ENG	15. DATE	
16. CORRECTIVE ACTION			17. MECH	18. ENG. L/C	19. DATE	
Performed at A/C TT : A/C TC /LDG :						
20. CORROSION INFORMATION						
LOCATION		CAUSE OF DAMAGE				
		☐ Environment				
		☐ Internal Leakage				
CORROSION ☐ Isolated ☐ Widespread		☐ Chemical Spill				
CORROSION LVL ☐ 1 ☐ 2 ☐ 3		☐ LAV/Galley Spill				
PROPOSED ACTION ☐ Doublers		☐ Blocked Drain				
☐ Others		☐ Wet Insulation Blanket				
		☐ Other				
21. If the defect is RII, Please Sign this card finally by RII Inspector				INSP	DATE	
NOTICE OF INSPECTOR						
22. PARTS REQUIRED						
PART DESCRIPTION	PART NO	QTY	SERIAL NO		STATUS	
			ON	OFF	CLOSE	OPEN
23. TOOLS REQUIRED						
DESCRIPTION	PART NO / MODEL	NEXT CALIBRATION DATE	STATUS			

Ref. AMM Pilatus Porter PC6 Chapter 61-10-00, PROPELLER INSTALLATION SHEET			
Date :	30 MAY 2023	Propeller P/N :	D4N00163SKD
Aircraft Registration :	PK-SNB	Propeller S/N :	FY4093
Aircraft Type :	PILATUS PC6, B2-H4	TSN :	739.56H
Aircraft Total Hours :	476:14	TSO :	0
Aircraft Total Cycle :	580	Work Order Number :	WO-012-SNB-XI-2022
NO	TASK	SIGNATURE	
		SIGN	STAMP
Job Set Up			
1	Do Engine Run Ground Sheet before Propeller installation, fill out Form No. XXXXXX	N/A	N/A
2	Ensure airplane electrical power is OFF.		
Installation (Ref. Fig. 401)			
1	(1) Attach the puller (16) and pull the slip ring fully forward to touch the rear face of the propeller bulkhead.	RII:  Sign & Stamp	
	(2) Install a new O-ring (15) on the propeller shaft.	RII:  Sign & Stamp	
	(3) Install the sling and put the propeller in position on the engine flange.	RII:  Sign & Stamp	
	(4) Lubricate the bolts (7) with grease (Material No. P04-009).	RII:  Sign & Stamp	
	CAUTION: MAKE SURE THAT THE CHAMFERED SIDE OF THE WASHER FACES THE BOLT HEAD.		
	(5) Install the attachment bolts (7) and washers (8), with the chamfered side of the washer next to the bolt head.	RII:  Sign & Stamp	
	(6) Torque the bolts in three steps as follows: a) To 54 Nm (480 lbf in.), (Ref. Fig. 402, Sequence A). b) To 108 Nm (960 lbf in.), (Ref. Fig. 402, Sequence A). c) To between 136 and 142 Nm (1200 and 1260 lbf in.), (Ref. Fig. 402, Sequence B).	RII:  Sign & Stamp	

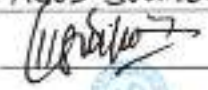
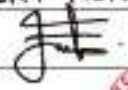
NO	TASK	SIGNATURE	SIGN	STAMP
	(7) Safety the bolts with lockwire (Material No. P02-001).	Ril:  Sign & Stamp 		
	(8) Remove the sling and puller.			
	(9) Put the carbon block (9) in position.	Ril:  Sign & Stamp 		
	(10) Put the reversing lever (12) in position.	Ril:  Sign & Stamp 		
	(11) Put the low pitch switch control rod in position.	Ril:  Sign & Stamp 		
	(12) Install the nut (5) and washer (6).	Ril:  Sign & Stamp 		
	(13) Safety the nut with a new cotter pin (4).	Ril:  Sign & Stamp 		
	(14) Install the pin (3) and washer (11).	Ril:  Sign & Stamp 		
	(15) Safety the pin with a new cotter pin (10).	Ril:  Sign & Stamp 		
	(16) Install the bolt (2), bush (15), washer (14) and nut (13).	Ril:  Sign & Stamp 		
	(17) Install the bolt (2), bush (15), washer (14) and nut (13).	Ril:  Sign & Stamp 		
	(18) Safety the nut with a new cotter pin (1).	Ril:  Sign & Stamp 		
	(19) Make sure the side clearance between the beta-ring and the carbon block (9) is between 0,03 and 0,25 mm (0.001 and 0.01 in). If necessary, remove material from, or replace, the carbon block (9) to get the correct clearance.	Ril:  Sign & Stamp 		
Job Close Up				
1	(1) Install the spinner dome to the bulkhead assembly. Align the marks on the spinner with the marks on the bulkhead.	Ril:  Sign & Stamp 		

NO	TASK	SIGNATURE	
		SIGN	STAMP
	NOTE: The spinner assembly (plate, bulkhead and dome) is balanced as a unit. The components are not interchangeable.		
	(2) Do a balance check.		
	(3) Do a ground run.		
	(4) Check for oil leaks from the propeller, - No leaks are acceptable.		
	(5) Make sure that the work area is clean and clear of tools and other items.		
	(6) Install the engine cowls PT1 and PB1.		
2	Make an appropriate entry in Work Order or Aircraft Flight Maintenance Log (AFML)		

PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER
AGUS SULAEMAN	ENGINEER		6444
Wahyono	Engineer		5702
FEBRI HERMANAN	ENGINEER		6445
PETER A.	MEKANIK		

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name	: AGUS SULAEMAN	Name of RII	: FEBRI HERMANAN
Signature	: 	Signature	: 
Stamp	: 	Stamp	: 
Place/Date	: BAYABIRU / 02 JUNE 23	Place/Date	: BAYABIRU / 02-JUNE-2023

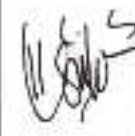
Ref. AMM Pilatus Porter PC6 Chapter 71-00-00, P&WC Maintenance Manual Model PT6A-27 Manual Part No. 3013242 Chapter 72-00-00 ENGINE INSTALLATION SHEET			
Date	: 29 MAY 2023	Engine P/N	: 3015000
Aircraft Registration	: PK-SNB	Engine S/N	: PCE-51153
Aircraft Type	: PILATUS PC6, B2-H4	TSN	: 18599.8H
Aircraft Total Hours	: 476.14H	TSO	: 19230
Aircraft Total Cycle	: 580	CSN	: 0
Work Order Number	: WO-012-SNB-XI-2022	CSO	: 0
NO	TASK	SIGNATURE	
		SIGN	STAMP
Job Set Up			
A	Make inventory records P/N and S/N of the engine and it's accessories from the engine that removed. Fill out into the form no ASI/MD-50 Engine Change- Major Component Inventory Records.		
Install the following item of Engine Accessories on the Engine before proceeding to install on the airplane:			
CAUTION: MAKE SURE THAT ALL BLANKS AND COVERS REMOVED FROM THE REPLACEMENT ENGINE ARE INSTALLED TO ALL OPENINGS ON THE REMOVED ENGINE			
Oil Drain Valve (Ref. Fig. 409)			
A	1. Remove the blank from the opening. 2. Install a new O-ring (4) to the oil drain valve (5). 3. Install the drain valve (5) with the lock pin (2). 4. Safety the lockpin (2) with a new cotter pin. 5. Install the bolt (1).		
Oil Temperature Sensor (Ref. Fig. 409)			
B	1. Remove the blank from the opening. 2. Install a new O-ring (11) to the oil temperature sensor. 3. Install the oil temperature sensor (10).		
Engine Breather Elbow (Ref. Fig. 409)			
C	1. Remove the blank from the opening. 2. Install a new O-ring (8) to the elbow (9). 3. Install the elbow (9) with the bolts (6) and washers (7). 4. Torque the bolts (6) to between 3.4 and 3.9 Nm (30 and 35 lbf in.). 5. Safety the bolts (6) with lockwire (Material No. PO2-001).		

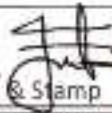
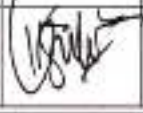
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D	Control Levers (Ref. Fig. 405)		
	- Idle Control Lever: - Put the lever (9) in position on the idle control unit. - Install the washer (10) and nut (11), then tighten the nut (11) with your hand - Install the bolt (6), control rod end (5), spacers (7) and (4), washer (3) and nut (2). - Torque the nut (2) to between 1.3 and 2.0 Nm (12 and 18 lbf in.). - Safety the nut (2) with a new cotter pin (1). - Torque the nut (11) to between 1.3 and 2.0 Nm (12 and 18 lbf in.). - Safety the nut (11) with a new cotter pin (8). Power Control Input Lever CAUTION: WHEN YOU INSTALL THE LEVER (18) IT MUST BE IN LINE WITH THE FOLLOWER LEVER ON THE SAME SHAFT OF THE CAM BOX. - Install the lever (18) to the splined shaft. - Install the bolt (17), the washer (19) and the nut (20). Safety the nut (20) with a new cotter pin (21).		
E	Engine Driven Fuel Pump (Ref. Fig. 410)		
	- Remove the blank from the opening. - Install a new gasket (8). - Install the fuel pump (7), washers (5) and nuts (6). - Torque the nuts (6) to between 8,5 and 9,6 Nm (75 and 85 lbf in.).		
F	Gas Generator and Propeller Tachometer Generators (Ref. Fig. 410)		
	- Remove the blank from the opening. - Install a new gasket (2). - Install the generator (1), washers (3) and nuts (4). - Torque the nuts (4) to between 8,5 and 9,6 Nm (75 and 85 lbf in.).		
G	Torque Indicating System (Ref. Fig. 411, Sheets 1 and 2)		
	- Install a new O-ring (1). - Install the adapter (8). - Remove the bottom RH nut (6) from the propeller tachometer generator. - Install the P-clip (4) and torque transducer (3). - Install the nut (6). - Torque the nut (6) to between 8,5 and 9,6 Nm (75 and 85 lbf in.). - Install the torque pressure pipe (2). - Install the case reference pressure hose (7).		
H	Propeller Overspeed Governor (Ref. Fig. 412)		
	- Install a new gasket (4) on the replacement engine. - Install the governor with the washers (1) and nuts (2). - Torque the nuts (2) to between 19,2 and 21,5 Nm (170 and 190 lbf in.).		

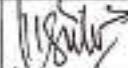
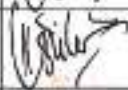
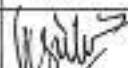
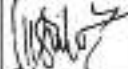
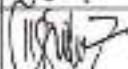
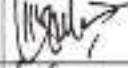
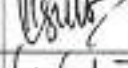
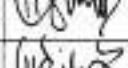
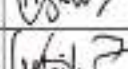
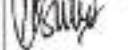
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		SIGN	STAMP
Drain Line Adapter, QAD Adapter (if applicable) and Starter-Generator (Ref. Fig. 413)			
I	1. Install the drain adapter (7). 2. Install a new gasket (1). 3. If applicable, install the QAD adapter (2) with the washers (3) and nuts (4). 4. Install the starter-generator (Ref. 24-31-11) but connect the electrical cables after you install the engine (Ref. Para. 7.H.(1))		
Harness (Ref. Fig. 414)			
J	1. If applicable, install the propeller de-icer unit (Ref. 30-60-01, Page Block 401). 2. If necessary, replace the heat insulating tape installed on the harness between the engine flanges A and C. 3. Put the harness assembly in position on the engine. 4. Install the P-clips to secure the harness. 5. Remove the identity marks and connect the ITT indication leads to the T5 terminal block. 6. If applicable, remove the identity marks and connect the propeller de-icer leads to the de-icer unit terminal block. 7. Connect the electrical plugs at the: i. chip detector. ii. propeller tachometer generator. iii. torque pressure transducer. 8. Remove the two thrust bearing cover attaching nuts. 9. Install the low pitch warning switch mounting bracket and warning switch on to the mounting studs. 10. Install the warning switch linkage block on to mounting stud on the propeller reversing lever. 11. Install the washer and nut to secure the warning switch linkage block, then tighten the nut with your hand. Make sure that the block is free to turn on the stud. 12. Safety the nut with a new cotter pin.		
Propeller Governor Maximum Stop Screw (Ref. Fig. 405)			
K	1. Check the maximum stop screw (35) on the propeller governor. If the screw is shorter than the stop screw on the removed engine, move the screws as follows: NOTE: The long stop screw is used to give the propeller speed of 2000 rpm used on B2H4 aircraft. a. On the removed engine: i. Loosen the locknut. ii. Remove the stop screw, at the same time count the number of turns necessary to remove the stop screw. iii. Make a record of the number of turns. b. Move the short stop screw from the replacement engine to the removed engine. c. Install the long stop screw on the propeller governor of the replacement engine with the same number of turns as recorded at (a) (iii). d. Tighten the stop screw locknut.		

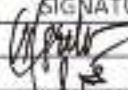
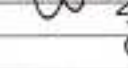
NO	TASK	SIGNATURE	
		SIGN	STAMP
L	Inner Bulkheads (Ref. Fig. 415)		
	1. Put the forward bulkheads (1) and (13) in position.		
	2. Install the bolts (2), washers (3) and nuts (4).		
	3. Install the bolts (5), washers (10) and nuts (11).		
	4. Install the harness slot filler plate on to the engine forward bulkhead.		
	5. Use caulking putty (Material No. P08-001) to seal the gaps between the harness and the bulkhead.		
	6. Put the rear bulkheads (6) and (12) in position.		
A	Engine Mounting Ring (Ref. Fig. 408)		
	1. Position the mounting ring on the engine to the rear of the attachment points. NOTE: Make sure that the support strut attachment lugs on the mounting ring are to the rear.		
	2. Install the engine lifting sling (Ref. P&WC EMM 72-00-00, SERVICING). CAUTION: THE BUSHES THAT ATTACH THE SHOCKMOUNT TO THE MOUNTING FRAME BOLT MUST BE INSTALLED. THE BUSH MUST HAVE THE THICK FLANGE TO THE REAR. IF THE BUSH IS INCORRECTLY INSTALLED, THE ENGINE WILL NOT ALIGN CORRECTLY.		
	3. Install the gaskets (7), shockmounts (10), washers (6) and bolts (5).		
	4. Torque the bolts (5) to between 31.0 and 34.0 Nm (275 and 300 lbf in.).		
	5. Safety the bolts in pairs with lockwire (Material No. P02-001).		
	6. Lubricate the shanks of the mounting ring attachment bolts (8).		
B	Install the Engine (Ref. Fig. 408)		
	1. Lift the engine in position.		
	2. Install the bolts (16) and washers (13).		
	3. Torque the bolts (16) to between 46.0 and 51.0 Nm (407 and 450 lbf in.).		
	4. Safety the bolts (16) with locking wire (Material No. P02-001).		

NO	TASK	SIGNATURE	
		SIGN	STAMP
C	Support Struts (Ref. Fig. 408)		
	1. Put the rear bulkhead over the engine.		
	2. Loosely install 10 bolts (five each side) to attach the upper and lower half of the bulkhead together.		
	3. Put the support struts in position through the holes in the engine rear bulkhead.		
	4. Install the bolt (1), washer (2) and nut (3).		
	5. Torque the nuts (3) to between 10,5 and 12,5 Nm (95 and 110 lb in.).		
	6. Safety the nuts with new cotter pins (4).		
	7. Make sure the bonded shim (18) is securely attached to the structure.		
	8. If the nuts (15) are self-locking nuts:		
	i. Install the bolt (19) washer (17) and nut (15). When you do this measure the rundown torque of the nut (15).		
D	Connect the Engine Bulkheads (Ref. Fig. 407)		
	1. Install the bolts (6) and washers (7) to attach the upper half of the engine rear bulkhead.		
	2. Tighten the bolts (4) and washers (3) that attach the support strut seal plates (5) and seals (2).		
E	Case Drains and P3 Air Bleed (Ref. Fig. 402)		
	1. Connect the gas generator case rear drain hose (8) at the rear drain valve.		
	2. Connect the gas generator case forward drain hose (1) at the engine forward bulkhead.		
	3. Connect the gas generator case forward drain pipe (7).		
	4. Install the gasket (6) to the mounting boss on the top of the gas generator case.		
	5. Put the adapter in position and install the four bolts (4).		
	6. Connect the P3 bleed air hose (2) to the adapter (5).		
	7. Install the P-clip (3) on the P3 bleed air hose (2) and engine forward bulkhead.		
	8. Safety the hose to the lower LH side of the engine mounting ring with a cable tie.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
F	Fuel Tubes (Ref. Fig. 404, Sheets 1 and 2)		
	1. Install a new O-ring (11) on the adapter (12).		
	2. Install the adapter to the starting flow control unit.		
	3. Set the adapter to the position noted at 4. E. (10).		
	4. Hold the adapter and then tighten the adapter locknut.		
	5. Connect the fuel dump hose (14) to the adapter (12).		
	6. Install and tighten the worm-drive clamp (13).		
	7. Connect fuel bleed return hose (10) to starting flow control unit.		
	8. Install a new O-ring (1) to the adapter (2).		
	9. Install the adapter (2) to the oil fuel heater.		
	10. (10) Connect the fuel supply hose (3) to the adapter (2).		
	11. Connect the drain tube (8) to the LP fuel pump.		
	12. Install a new O-ring (5) to the adapter (6).		
	13. Install the adapter (6) to the HP fuel pump.		
	14. Connect the drain tube (7) to the adapter (6).		
	15. Connect the inlet hose (4) to the LP fuel pump.		
	16. Connect outlet hose (9) to the LP fuel pump.		
G	Oil Tubes (Ref. Fig. 403)		
	1. Connect the starter generator drain hose (Ref. Fig. 413):		
	i. Connect the drain hose (5) to the drain adapter (7) below the starter generator drive.		
	ii. Install and tighten the clamp (6).		
	2. Connect the engine breather hose (3) to the breather elbow.		
	Install and tighten the clamp (2).		
	3. Connect the oil cooler return hose (1) to the adapter on top of the accessory gearbox.		
	4. Connect the engine oil cooler supply hose (7) to the connector on the scavenge pump.		
H	Electrical Connectors (Ref. Fig. 401)		
	1. Connect the electrical cables (1) to the starter-generator.		
	2. Connect the electrical plug (8) to the oil temperature sensor.		
	3. Connect the electrical plug (7) to the FCU sense line heater.		
	4. Connect the electrical plug (5) to the gas generator tachometer generator.		
	5. Connect the electrical plug (2) to the ignition unit.		
	6. Connect the electrical plug (9) at the RH of the front face of the engine rear bulkhead.		
	7. If the propeller de-icing system is installed:		
	i. Install electrical plug (10) at the RH front face of the engine rear bulkhead.		
	8. Install the earth leads (6) to engine bulkhead.		
	Connect bonding lead (3) to engine data plate securing bolt (4).		

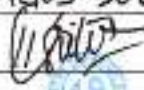
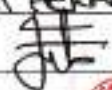
NO	TASK	SIGNATURE	
		SIGN	STAMP
I	Connect the Propeller Control Cable (Ref. Fig. 406)		
	1. Insert the cable through the hole in the engine rear bulkhead. 2. Install the two grommets to the cable. 3. Insert the cable through the hole in the forward bulkhead. 4. Position the cable on the engine. 5. Press the grommets in the holes of the bulkhead. 6. Put the support bracket (9) in position. 7. Install the bolts (12), washers (10) and nuts (11). 8. Put the cable in position on the support bracket (9). 9. Install the top plate (15) with the bolts (16), spring washers (14) and nuts (13).		
J	Connect the Controls (Ref. Fig. 405)		
	1. Insert the cable through the hole in the engine rear bulkhead. 2. Install the two grommets to the cable. 3. Insert the cable through the hole in the forward bulkhead. 4. Position the cable on the engine. 5. Press the grommets in the holes of the bulkhead. 6. Put the support bracket (9) in position. 7. Install the bolts (12), washers (10) and nuts (11). 8. Put the cable in position on the support bracket (9). 9. Install the top plate (15) with the bolts (16), spring washers (14) and nuts (13).		
K	Connect the Controls (Ref. Fig. 405)		
	1. Idle Control: i. Put the rod (15) in position and then install the bolt (16), washer (14) and nut (13). ii. Torque the nut (13) to between 1,3 and 2,0 Nm (12 and 18 lbf.in.). iii. Safety the nut with a new cotter pin. 2. Power Control: i. Put the rod end (25) in position and then install the bolt (26), washer (24) and nut (23). ii. Torque the nut (23) to between 1,3 and 2,0 Nm (12 and 18 lbf.in.). iii. Safety the nut with a new cotter pin (22). 3. Propeller Control: i. Put the control rod (31) in position and then install the bolt (34), washer (33), spacer (32), washer (29) and nut (26). ii. Torque the nut (26) to between 1,3 and 2,0 Nm (12 and 18 lbf.in.). iii. Safety the nut (26) with a new cotter pin (22).		
	R11 :  Sign & Stamp 		

NO	TASK		SIGNATURE	
			SIGN	STAMP
Job Close Up				
A	1. Adjust the engine controls.	RII:		 
	2. Install the exhaust stubs.			 
	3. Install the propeller.	RII:		 
	4. If applicable, install the oil cooler.			N/A N/A
	5. Fill the engine oil system.			 
	6. If installed, remove the aircraft tail stand and lower aircraft tail.			 
	7. Make sure that the work area is clean and clear of tools and other items.			 
	8. Do a fuel system de-preservation.			 
	9. Remove the engine compressor intake cover.			 
	10. Install the engine cowlings.			 
	11. Connect the aircraft battery.			 
	12. Do an engine ground run and fill out Engine Ground Run Sheet.			 
B	Make an appropriate entry in Work Order or Aircraft Flight Maintenance Log (AFML)			 

PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER
AGUS SULAEMAN	ENGINEER		6494
Wahyono	ENGINEER		5792
FERRI HERMAWAN	ENGINEER		6445

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name	:	<u>AGUS SULAEMAN</u>	Name of RII	:	<u>FERRI HERMANAN</u>
Signature	:		Signature	:	
Stamp	:		Stamp	:	
Place/Date	:	<u>BAYABIRU / 02 JUNE 23</u>	Place/Date	:	<u>BAYABIRU / 02-JUNE-2023</u>



TECHNICAL SUPPORT
TECHNICAL DEPARTMENT
ENGINEERING ORDER

Rev. No	Original
Rev. Date	

ENGINE CHANGE - Major Component Inventory Record

Registration	: PK-SNB	Work Order Number	: WO/012-SNB/X1/2022
Airframe Time	: 476 :1A	Airframe Landings	: 580
Engine Time	: 18599:8 H	Engine Cycle	: 19230

Description	Engine OFF			Engine ON		
	Part Number	Serial Number	Time Remaining	Part Number	Serial Number	Time Remaining
Engine Assembly	3015000	PCE-PG0565	SEND TO SHOP	3015000	PCE-51153	NEW OVERHAUL
Propeller Assembly	HC-DAN-3P/D9511F	FYA513		DAN00163SKD	FYA093	
Compressor Bleed Valve	3075033-01	AHX1501A2A		3100829-03	60820	
Fuel Control Unit	30A9635-02	C22A77		324A786-8-7	C22007	
Oil Fuel Heater	3032710	WA52186		10552E	565	
Igniter Exciter	3035889A	NNA15340030		10-381550-A	NNA08455077	
Flow Divider / CONTROL STARTING	3019109	20105		3019109	13228	
Oil Cooler	15060730	4500209514		15060730	4500209514	
Starter Generator	230A8-021	P1046		230A8-021	P1046	
Alternator	N/A	N/A		N/A	N/A	
Fuel Pump	303A79A	00451		025323-101-03	53A9	
Propeller Governor	3045798-02	201355A1		66503-8210-412-01	16980915	
Propeller Overspeed Governor	210615	20000956		210615	202307A9	
Fuel Nozzle	3011151B			3011151B		

NOTE: ANY OTHER COMPONENT CHANGES MUST BE FILL ON INSPECTION CARD (SCA/MTC/048)

	ENGINE GROUND RUN CHECK SHEET - PT6A-27 ENGINE WITH FOUR BLADE PROPELLER (HARTZELL STC SA377CH)
	

WORK ORDER NO.		: WO / 012-SNB / X1 / 2022	
Aircraft Registration	PK-SNB	Aircraft Total Hours	476:1A
Aircraft Serial No.	1015	Aircraft Total Landings	580
Engine Serial No.	PCE-51153	Engine TSN / TSO	18599:84/0
Propeller Serial No	FY4093	Propeller TSN / TSO	739:56 H / 0
Ambient Temp	33 °C	FBP (Field Barometric Pressure)	29.91 In.Hg
Date	02 JUNE 23	Time	15:50 LT
Mechanic / Engineer		Authorized Engineer	AGUS SULAIMAN
Reason For Ground Run		TEST PERFORMANCE AFTER ENGINE INSTALLED.	

Checks to be carried out, No:	1 2 4 5 7 8 9 10 11 12 13 14 15
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Engine Ground Run Check Frequency

Check Number	1	2	3	4	5	6	7	8	9	10	11	12	13	14	15
Each 100 / Yearly	x	x		x			x	x			x	x	x	x	x
Each 200									x						
Pre-Complete Overhaul	x	x	x	x		x	x	x	x	x	x	x	x	x	x
After Short Term Storage															x
After Long Term Storage	x	x	x	x		x	x	x	x	x	x	x	x	x	x

In addition the following check must be carried out after Installation, Repair and Adjustment of any of the following components.

Check Number	1	2	3	4	5	6	7	8	9	10	11	12	13	14	15
Engine Installation ✓	x	x	x	x	x	x	x	x	x	x	x	x	x	x	x
Propeller Installation		x	x	x	x			x							
Fuel Control Unit	x				x	x	x	x		x	x				
HP Fuel Pump						x	x								
Fuel Nozzle						x	x								
Starting Flow Control	x				x		x	x							
Emer Fuel Control Actuator											x				
Prop Governor	x		x	x	x		x	x							
Prop Overspeed Governor									x						
Compressor Bleed Valve						x	x								
Engine Controls	x			x	x			x	x						
Low Pitch Warning Switch				x											
Suction Components														x	

Use this sheet's to record engine run result, use in conjunction with task cards.

NO.	CHECK	TARGET	ACTUAL
ENGINE START			
	ITT (Troubleshoot If More Than 925°C)	Max. 1090 °C	623 °C
	Cabin Heat	OFF	✓ OK?
1	Low Idle (Minimum Governing) Speed	51 - 53 % Ng	53 % Ng
	Fuel Pressure / Boost Pump OFF	Light out or 25 ± 5 psi	✓ OK?
	ITT		536 °C
	Oil Pressure		93 psi
	Oil Temperature		68 °C
2	Propeller Governor		
	Maximum Np	1980 - 2000 rpm (90.0 - 90.9 %)	1980 rpm
	Py Disconnected		90.3 % Ng
	Py Connected		90.3 % Ng
	Difference	Maximum 0.3% Ng	0 %
	Airbleed Link at Minimum	1900 - 1950 rpm (86.4 - 88.6 %)	1950 rpm
	Aircraft with SB 161:		
	Propeller Control Lever at Minimum	1880 - 1900 rpm (85.5 - 86.4 %)	rpm
3	Propeller Fine Pitch Setting (High Idle)		
	Target Torque	psi	7.0 psi
	Power Lever to Give Np	1694 rpm (77 %)	1670 rpm
	Basic High Idle	68 - 72% Ng	68.7 %Ng
4	Propeller Low Pitch Warning		
	PCL from Reverse to Detent	Light OFF 1 to 2 mm before Detent	2 mm
5	Minimum Pitch in Flight		
	Ng	67 - 73 %	70 % Ng
	Np	1800 - 1950 rpm (81.8 - 88.6 %)	1900 rpm
	Torque	4 - 7 psi	7 psi
6	FCU Maximum Governing Speed (Ng) (Trim stop deployed)	97.1 % Ng	97.6 % Ng

NO.	CHECK	TARGET	ACTUAL
7	Engine Performance Target Torque Pressure Fuel flow (Actual minus 23 lb / hr or 3.4 gal / hr) Target Ng Maximum ITT	Ref: AMM 71-00-00 41 psi 55 gal/hr 342 lb/hr 95.4% Ng 697 °C	41 psi 44.6 / 262 lb/hr 96.0 % Ng 625 °C
8	Reverse Power Setting Np Torque	1880 - 1925 rpm (85.5 - 87.5 %) psi	1900 rpm 18. psi
9	Propeller Overspeed Governor Test Lever Selected to: TEST NORMAL	1880 - 1920 rpm (85.5 - 87.3 %) 1980 - 2000 rpm (90.0 - 90.9 %)	1920 rpm 1980 rpm
10	Acceleration 64 % – 90 % Ng Deceleration 85% to 60% Ng or low idle speed(Whichever comes first)	2.5 – 4 secs Maximum 6-12 sec (Dependent upon altitude)	3 secs 7 secs altitude (kft)
	Manual Override (MOR) (Aircraft with SB 164) Use Toggle Switch In Small Increment (REF. to WARNINGS and CAUTIONS in Check 11)	Increase to 15% above Idle (Max Increase less than 4 % per Second) Decrease To Idle (Max Decrease less Than 4% per Second)	✓ OK? ✓ OK?
12	Oil Pressure	80 -100 psi	92 psi
13	Generator (Ref. 24-30-00)	Online by 60% Ng	60 % Ng
14	Suction (High Idle)	4.5 – 5.2 in. Hg	N/A in. Hg
15	Engine Rundown Time After Stop	MIN 30 secs	33 secs
Additional			
	Generator Check (High Idle Under Load)	27.75 – 28.25 VDC	28.2 VDC
	After Engine Run		
	Check Eng. For Signs of Fuel/Oil/Air Leaks	NO LEAKS FOUND	✓ OK?
	Safety All Screws, Bolts, Locknuts as Req.		✓ OK?

	ENGINE GROUND RUN CHECK SHEET - PT6A-27 ENGINE WITH FOUR BLADE PROPELLER (HARTZELL STC SA377CH)
	

WORK ORDER NO.	: WO/012-SNB/X1/2022		
Aircraft Registration	PK-SNB	Aircraft Total Hours	476:14
Aircraft Serial No.	1015	Aircraft Total Landings	580
Engine Serial No.	PCE-51153	Engine TSN / TSO	18599:8 H / 0
Propeller Serial No	FY4093	Propeller TSN / TSO	739:56 H / 0
Ambient Temp	33 °C	FBP (Field Barometric Pressure)	29.91 In.Hg
Date	02 JUNE 23	Time	15:50 LT.
Mechanic / Engineer		Authorized Engineer	AGUS SULAEMAN
Reason For Ground Run	TEST PERFORMANCE AFTER PROP INSTALLED.		

Checks to be carried out. No:	1 2 4 5 7 8 9 10 11 12 13 14 15
-------------------------------	---------------------------------

Engine Ground Run Check Frequency

Check Number	1	2	3	4	5	6	7	8	9	10	11	12	13	14	15
Each 100 / Yearly	x	x		x			x	x			x	x	x	x	x
Each 200									x						
Pre-Complete Overhaul	x	x	x	x		x	x	x	x	x	x	x	x	x	x
After Short Term Storage															x
After Long Term Storage	x	x	x	x		x	x	x	x	x	x	x	x	x	x

In addition the following check must be carried out after Installation, Repair and Adjustment of any of the following components.

Check Number	1	2	3	4	5	6	7	8	9	10	11	12	13	14	15
Engine Installation	x	x	x	x	x	x	x	x	x	x	x	x	x	x	x
Propeller Installation ✓		x	x	x	x			x							
Fuel Control Unit	x				x	x	x	x		x	x				
HP Fuel Pump						x	x								
Fuel Nozzle						x	x								
Starting Flow Control	x				x		x	x							
Emer Fuel Control Actuator											x				
Prop Governor	x		x	x	x		x	x							
Prop Overspeed Governor									x						
Compressor Bleed Valve						x	x								
Engine Controls	x			x	x			x	x						
Low Pitch Warning Switch				x											
Suction Components														x	

Use this sheet's to record engine run result, use in conjunction with task cards.

NO.	CHECK	TARGET	ACTUAL
ENGINE START			
	ITT (Troubleshoot If More Than 925°C)	Max. 1090°C	°C
	Cabin Heat	OFF	OK?
1	Low Idle (Minimum Governing) Speed	51 - 53 % Ng	% Ng
	Fuel Pressure / Boost Pump OFF	Light out or 25 ± 5 psi	OK?
	ITT		°C
	Oil Pressure		psi
	Oil Temperature		°C
2	Propeller Governor		
	Maximum Np	1980 - 2000 rpm (90.0 - 90.9 %)	1980 rpm
	Py Disconnected		90.3 % Ng
	Py Connected		90.3 % Ng
	Difference	Maximum 0.3% Ng	0 %
	Airbleed Link at Minimum	1900 - 1950 rpm (86.4 - 88.6 %)	1950 rpm
	Aircraft with SB 161:		
	Propeller Control Lever at Minimum	1880 - 1900 rpm (85.5 - 86.4 %)	rpm
3	Propeller Fine Pitch Setting (High Idle)		
	Target Torque	psi	7.0 psi
	Power Lever to Give Np	1694 rpm (77 %)	1670 rpm
	Basic High Idle	68 - 72% Ng	68.7 % Ng
4	Propeller Low Pitch Warning		
	PCL from Reverse to Detent	Light OFF	
		1 to 2 mm before Detent	2 mm
5	Minimum Pitch in Flight		
	Ng	67 - 73 %	70 % Ng
	Np	1800 - 1950 rpm (81.8 - 88.6 %)	1900 rpm
	Torque	4 - 7 psi	7 psi
6	FCU Maximum Governing Speed (Ng)	97.1 % Ng	% Ng
	(Trim stop deployed)		

NO.	CHECK	TARGET	ACTUAL
7	Engine Performance Target Torque Pressure Fuel flow (Actual minus 23 lb / hr or 3.4 gal / hr) Target Ng Maximum ITT	Ref: AMM 71-00-00 psi lb / hr % Ng °C	 psi lb / hr % Ng °C
8	Reverse Power Setting Np Torque	1880 - 1925 rpm (85.5 - 87.5 %) psi	1900 rpm 18 psi
9	Propeller Overspeed Governor Test Lever Selected to: TEST NORMAL	1880 - 1920 rpm (85.5 - 87.3 %) 1980 - 2000 rpm (90.0 - 90.9 %)	 rpm rpm
10	Acceleration 64 % – 90 % Ng Deceleration 85% to 60% Ng or low idle speed(Whichever comes first)	2.5 – 4 secs Maximum 6-12 sec (Dependent upon altitude)	secs secs altitude (kFt)
	Manual Override (MOR) (Aircraft with SB 164) Use Toggle Switch In Small Increment (REF. to WARNINGS and CAUTIONS in Check 11)	Increase to 15% above Idle (Max Increase less than 4 % per Second) Decrease To Idle (Max Decrease less Than 4% per Second)	OK? OK?
12	Oil Pressure	80 -100 psi	psi
13	Generator (Ref. 24-30-00)	Online by 60% Ng	% Ng
14	Suction (High Idle)	4.5 – 5.2 in. Hg	in. Hg
15	Engine Rundown Time After Stop	MIN 30 secs	secs
Additional			
	Generator Check (High Idle Under Load)	27.75 – 28.25 VDC	VDC
	After Engine Run		
	Check Eng. For Signs of Fuel/Oil/Air Leaks	NO LEAKS FOUND	OK?
	Safety All Screws, Bolts, Locknuts as Req.		OK?

1. Approving Civil Aviation Authority/Country: FAA/United States		2. AUTHORIZED RELEASE CERTIFICATE FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG			3. Form Tracking Number: T6943-1	
4. Organization Name and Address: Turbines, Inc. 7303 Maynard Wheeler Lane. Terre Haute International Airport. Terre Haute, IN 47803					5. Work Order/Contract/Invoice Number: T6943	
6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:	
1	PT6A-27 Turboprop Engine	3015000	1	PCE-51153	Repaired	
12. Remarks: Partially disassembled. Cleaned RGB, AGB and Oil Tank. Configured to BS 877 (Pilatus). Assembled and Functionally Tested Engine. Test results satisfactory. Reference Overhaul Manual PN 3013243 Rev. 45. Total Time: 18599.8 Hrs. Total Cycles: 19230 TSO: 0.0 Hrs. CSO: 0						
13a. Certifies the items identified above were manufactured in conformity to: ☐ Approved design data and are in condition for safe operation ☐ Non-approved design data specified in Block 12.			14a. ☒ 14 CFR 43.9 Return to Service ☐ Other regulation specified in Block 12 Certifies that unless otherwise specified in block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.			
13b. Authorized Signature:		13c. Approval/Authorization No.:		14b. Authorized Signature:		14c. Approval/Certificate No.:
						UTBR513L
13d. Name (Typed or Printed):		13e. Date (dd/mm/yyyy):		14d. Name (Typed or Printed): Jay Streb		14e. Date (dd/mm/yyyy): 10 JUN 2022
User/Installer Responsibilities						
It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.						
Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.						
Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.						

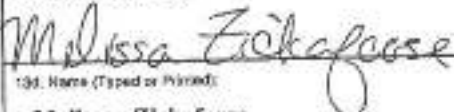
1. Approving Civil Aviation Authority/Country: FAA/UNITED STATES		AUTHORIZED RELEASE CERTIFICATE FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG			3. Form Tracking Number: 20220304AT	
4. Organization Name and Address: Hartzell Propeller Inc.		One Propeller Place Piqua, Ohio 45356 (PC-10)			5. Work Order/Construction Number: 295269	

6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:
1	O - RING	C-3317-230	200	N/A	NEW
2	O - RING	C-3317-251	50	N/A	NEW
3	DE-ICE BOOT	4H2271-10	60	N/A	NEW
4	BEARING, NEEDLE, CLOSED END	B-6680	100	N/A	NEW
5	DOWEL PIN, 3/8"	B-6260	50	N/A	NEW
6	FITTING, LUBRICATION, 65"	109032	50	N/A	NEW
END					

12. Remarks:
 Additional Information: Eligibility information for Hartzell Propeller system components is contained in the appropriate Hartzell maintenance manual. This information is also available for Hartzell ice protection system components at www.hartzellprop.com

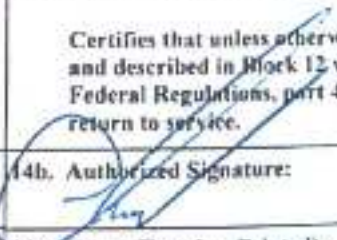
Packaging Methods used to protect against corrosion and damage: **QC-05-003** Duration of Methods: **6 months**

11a. Certifies the items identified above were manufactured in conformity to: ☒ Approved design data and are in condition for safe operation ☐ Non-approved design data specified in Block 12		14a. ☒ 14 CFR 43.9 Return to Service ☐ Other regulations specified in Block 12 Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.	
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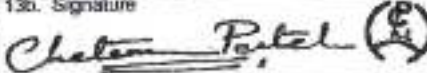
13a. Authorized Signature: 	13c. Approval/Authorization No.: ODA-100082-CE	14b. Authorized Signature:	14c. Approval/Certificate No.:
13d. Name (Typed or Printed): Melissa . Zickafosse	13e. Date (dd/mm/yyyy): 04/Mar/2022	14d. Name (Typed or Printed):	14e. Date (dd/mm/yyyy):

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the engine/propeller/article. Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness accepts engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1. Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

1. Approving Civil Aviation Authority/Country: United States		2. AUTHORIZED RELEASE CERTIFICATE FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG			3. Form Tracking Number: 22-1262																						
4. Organization Name and Address:		Air Prop Specialist 3671 Industrial Park Drive Marianna, FL 32446			5. Work Order/Contract/Invoice Number: 22-1262																						
6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:																						
1	Hartzell Propeller	DMN00163SKD	1	FY4093	Overhauled																						
12. Remarks:																											
Overhauled propeller in accordance with Hartzell Manuals: 143A, 133C, 159, 202A. All other current service information and Airworthiness Directives complied with as of this date. NOTICE: An Airworthiness Directive may apply to the installation of the propeller described hereon. The installer is responsible for ensuring complete compliance with any applicable installation Airworthiness Directive. Knockdown propeller in accordance with Hartzell Manuals. Shipped to customer disassembled. The prop must be assembled by trained personnel.					<table style="width: 100%; border-collapse: collapse;"> <thead> <tr> <th></th> <th style="text-align: right;">TSN</th> <th style="text-align: right;">TSO</th> </tr> </thead> <tbody> <tr> <td>Propeller</td> <td style="text-align: right;">739.56</td> <td style="text-align: right;">0.00</td> </tr> <tr> <td>Hub S/N - FY4093</td> <td style="text-align: right;">739.56</td> <td style="text-align: right;">0.00</td> </tr> <tr> <td>Blade # 1 - L47095</td> <td style="text-align: right;">739.56</td> <td style="text-align: right;">0.00</td> </tr> <tr> <td>Blade # 2 - L47104</td> <td style="text-align: right;">739.56</td> <td style="text-align: right;">0.00</td> </tr> <tr> <td>Blade # 3 - L47096</td> <td style="text-align: right;">739.56</td> <td style="text-align: right;">0.00</td> </tr> <tr> <td>Blade # 4 - L47105</td> <td style="text-align: right;">739.56</td> <td style="text-align: right;">0.00</td> </tr> </tbody> </table>			TSN	TSO	Propeller	739.56	0.00	Hub S/N - FY4093	739.56	0.00	Blade # 1 - L47095	739.56	0.00	Blade # 2 - L47104	739.56	0.00	Blade # 3 - L47096	739.56	0.00	Blade # 4 - L47105	739.56	0.00
	TSN	TSO																									
Propeller	739.56	0.00																									
Hub S/N - FY4093	739.56	0.00																									
Blade # 1 - L47095	739.56	0.00																									
Blade # 2 - L47104	739.56	0.00																									
Blade # 3 - L47096	739.56	0.00																									
Blade # 4 - L47105	739.56	0.00																									
13a. Certifies the items identified above were manufactured in conformity to:			14a. ☒ 14 CFR 43.9 Return to Service ☐ Other regulation specified in Block 12																								
☐ Approved design data and are in a condition for safe operation. ☐ Non-approved design data specified in Block 12			Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title, 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.																								
13b. Authorized Signature:		13c. Approval/Authorization No.:		14b. Authorized Signature:		14c. Approval/Certificate No.:																					
						1X4R274M																					
13d. Name (Typed or Printed):		13e. Date (dd/mm/yyyy):		14d. Name (Typed or Printed):		14e. Date (dd/mm/yyyy):																					
				Christopher Giersberg		19 May 2022																					
User/Installer Responsibilities																											
It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article. Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1. Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.																											

DOCUMENT OFFICIEL - OFFICIAL RECORD

1. Autorité de l'aviation civile/Pays qui approuve le bon de sortie Approving Civil Aviation Authority/Country		2. BON DE SORTIE AUTORISÉE - AUTHORIZED RELEASE CERTIFICATE		3. Numéro de suivi du formulaire - Form Tracking No.	
Transport Canada		FORM ONE		800653792-000001 3986596	
4. Nom et adresse de l'organisme - Organization Name and Address		Pratt & Whitney Canada Cie. / Pratt & Whitney Canada Corp. 1000 boul. Marie-Victorin Longueuil, Québec Canada J4G 1A1 (450) 677-9411		5. Bon de travail/Contrat/Facture - Work Order/Contract/Invoice	
				3488865 4500355275	
6. Article - Item	7. Description	8. Numéro de pièce - Part No.	9. Qtes. - Qty.	10. Numéro de série/lot - Serial/Batch No.	11. Situation/Travail - Status/Work
000001	GASKET	ST3277-01	100 EA	N/A	NEW
12. Remarques - Remarks					
NOTE: Les numéros de série figurant sur les étiquettes ou sur les pièces servent à la fabrication et ne doivent pas être inscrits sur le présent certificat. NOTE: Any serial numbers that may be present on labels or parts are for manufacturing purposes and need not be referenced on this certificate.					
Le cas échéant, la durée de stockage est précisée sur l'emballage de l'article - When applicable, shelf life is indicated on the item packaging					
"Le numéro de pièce identifié sur la pièce peut inclure un suffixe alphabétique. Ce suffixe alphabétique désigne le niveau de révision de la pièce. Tous les niveaux de révision pour une pièce donnée sont parfaitement interchangeables. Le niveau de révision d'une pièce n'est pas requis pour ce certificat." "The part number marked on a part may include an alpha suffix. This alpha suffix denotes the part's revision level. All revision levels for a given part number are fully interchangeable. The part's revision level is not required for this certification."					
13a. Le présent bon de sortie certifie que les articles indiqués ci-dessus ont été construits conformément à : Certifies that the items identified above were manufactured in conformity to:		14a. ☐ R.A.C. 671.10 (certification après maintenance) - CAR 671.10 Maintenance Release ☐ Autre réglementation précisée à la case 12 - Other regulations specified in block 12			
☒ des données de conception approuvées et qu'ils peuvent être utilisés en toute sécurité approved design data and are in condition for safe operation. ☐ des données de conception non approuvées indiquées à la case 12. non approved design data specified in block 12.		Le présent bon de sortie certifie que, sauf indication contraire à la case 12, le travail indiqué à la case 11 et décrit à la case 12, a été effectué conformément au Règlement de l'aviation canadien. Certifies that unless otherwise specified in block 12, the work identified in block 11 and described in block 12 has been performed in compliance with the Canadian Aviation Regulations.			
13b. Signature	13c. Numéro de l'organisme agréé Approved Organization Number	14b. Signature	14c. Numéro de l'organisme agréé Approved Organization Number		
 T.C. 4-58 NO. 087	4-58	S/O - N/A	S/O - N/A		
13d. Nom - Name	13e. Date (dd/mm/yyyy)	14d. Nom - Name	14e. Date (dd/mm/yyyy)		
C. Patel	19/MAY/2022	S/O - N/A	S/O - N/A		

* Le monteur doit contre-vérifier l'admissibilité avec les données approuvées - Installer must cross-check eligibility with approved data

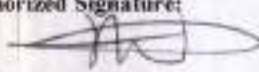
RESPONSABILITÉS DU MONTEUR - INSTALLER RESPONSIBILITIES

1. Le présent bon de sortie ne constitue pas une autorisation de montage. - 1. This certificate does not constitute authority to install.

2. Le monteur qui travaille conformément à la réglementation d'un pays autre que celui spécifié à la case 1 doit s'assurer que la réglementation en question reconnaît la certification du pays ainsi spécifié. - 2. Installers working in accordance with the national regulations of a country other than that specified in block 1 must ensure that their regulations recognize certifications from country specified.

3. Les déclarations des cases 13a et 14a ne constituent pas une certification du montage. Dans tous les cas, le dossier technique de l'aéronef doit inclure une certification de montage délivrée conformément à la réglementation nationale qui s'applique, avant que l'aéronef puisse voler. - 3. Statements in block 13a and 14a do not constitute installation certification. In all cases, the technical record for the aircraft must contain an installation certification issued in accordance with the applicable national regulations before the aircraft may be flown.

1. Approving Competent Authority / Country FOCA/Switzerland		2. AUTHORISED RELEASE CERTIFICATE EASA FORM 1				3. Form Tracking Number AST-23-112 Page 1 of 1	
4. Approved Organisation Name and Address PILATUS		PILATUS AIRCRAFT LTD CH 6371 STANS SWITZERLAND		Telephone +41 41 619 61 11 Telefax +41 41 619 66 76 Website www.pilatus-aircraft.com		5. Work Order/Contract/Invoice 5650451	
6. Item	7. Description	8. Part No.	9. Quantity	10. Serial No.	11. Status/Work		
1	GOVERNOR ASSEMBLY, TURBOPROP	8210-412-01	1	16980915	INSPECTED/TESTED		
12. Remarks GOVERNOR, PROPELLER INSPECTED AND TESTED I.A.W. CMM WG60044 REV. 3 1997-09-03 THIS ITEM IS FOR 2000 PROPELLER RPM CONFIGURATION (PC 6 B2/H4) ONLY TSN: 0.00 HRS. TSO: N/A ITEM FROM STORE Canadian Approval No. TCCA 898-37 The work identified in Block 11 and described herein has been accomplished in accordance with 14 CFR part 43 and in respect to that work, the items are approved for return to service under Certificate no. 8UKY006D							
13a. Certifies that the items identified above were manufactured in accordance in conformity to: ☐ approved design data and are in condition for safe operation ☐ non-approved design data specified in block 12				14a. ☒ Part-145.A.50 Release to Service ☒ Other regulation specified in block 12 Certifies that unless otherwise specified in block 12, the work identified in block 11 and described in block 12 was accomplished in accordance with Part-145 and in respect to that work the items are considered ready for release to service.			
13b. Authorised Signature		13c. Approval/Authorisation Number		14b. Authorised Signature  Meyer Ch. CH-2591		14c. Certificate/Approval Ref. No. CH.145.0209	
13d. Name		13e. Date		14d. Name Ch.Meyer CH-2591		14e. Date 24/Feb/2023	
USER / INSTALLER RESPONSIBILITIES NOTE: 1 This certificate does not automatically constitute authority to install the item(s). 2 Where the user/installer perform work in accordance with regulations of an airworthiness authority different than the airworthiness authority specified in block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts items from the airworthiness authority specified in block 1. 3 Statements in blocks 13a and 14a do not constitute installation certification. In all cases aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.							

1. Approving Civil Aviation Authority/Country: FAA/United States		2. AUTHORIZED RELEASE CERTIFICATE FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG			3. Form Tracking Number: T6110-90	
4. Organization Name and Address: Turbines, Inc. 7303 Maynard Wheeler Lane, Terre Haute International Airport, Terre Haute, IN 47803					5. Work Order/Contract/Invoice Number: T6110	
6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:	
1	Cam block	NPN	1	NSN	Assembled	
12. Remarks: Assembled Cam block reference to PWC Overhaul Manual 3021243.						
Turbines, Inc. certifies that the work specified in block 11/12 was carried out in accordance with EASA Part-145 and in respect to that work the article is considered ready for release to service under EASA Part-145 Approval Number: EASA.145.6780.						
13a. Certifies the items identified above were manufactured in conformity to: ☐ Approved design data and are in condition for safe operation ☐ Non-approved design data specified in Block 12.			14a. ☒ 14 CFR 43.9 Return to Service ☒ Other regulation specified in Block 12 Certifies that unless otherwise specified in block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.			
13b. Authorized Signature:		13c. Approval/Authorization No.:		14b. Authorized Signature: 		14c. Approval/Certificate No.: UTBR513L
13d. Name (Typed or Printed):		13e. Date (dd/mm/yyyy):		14d. Name (Typed or Printed): Dustin Wincinger		14e. Date (dd/mm/yyyy): 28/FEB/2023
User/Installer Responsibilities						
It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.						
Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.						
Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.						

CERTIFICATE OF CONFORMANCE

Certificate Ref. No.

80370407

Page 1 of 1



Address

Certified to EN 9100/ISO 14001 standards

PILATUS AIRCRAFT LTD
CH-6371 STANS
SWITZERLAND

Telephone

+41 41 619 61 11

Telefax

+41 41 619 64 93

Your Work Order / Contract / Pilatus Reference

PO-433 /

249021

Item	Description Vendor Part No. Date / Expiry Date	Part No.	Unit of Supply	Qty.	Serial / Batch No.
00010	GOVERNOR, PROPELLER 3045798-02 Man. Date: 10.10.2012 Exp. Date: 09.09.2018 ----- END -----	9682011544	ST	1	16980915
REMARKS					

Certified that the whole of the supplies detailed on all pages of this certificate have been manufactured, inspected and unless otherwise stated above, conform in all respects to the specifications, drawings and contract/order relevant thereto.

Name F. Antonazzo

Date (d/m/y) 1.MAR.2023

Authorised Signature
(Electronic Signature on File)

Autorité de l'aviation civile/Pays qui approuve le bon de sortie Approving Civil Aviation Authority/Country Transport Canada		2. BON DE SORTIE AUTORISÉE - AUTHORIZED RELEASE CERTIFICATE FORM ONE			3. Numéro de suivi du formulaire - Form Tracking No. 800727997-000010 4038542	
Nom et adresse de l'organisme - Organization Name and Address  Pratt & Whitney Canada Cie. / Pratt & Whitney Canada Corp. 1000 boul. Marie-Victorin Longueuil, Québec Canada J4G 1A1 (450) 677-8411		5. Bon de travail/Contrat/Facture - Work Order/Contract/Invoice 3491940 4500357246				
Article - Item	Description	6. Numéro de pièce - Part No.	9. Cité - City	10. Numéro de série/lot - Serial/Batch No.	11. Situation/Travail - Status/Work	
000001	GASKET, OVERSPEED GOVERNOR	3118953-01	20 EA	N/A	NEW	
12. Remarques - Remarks NOTE: Les numéros de série figurant sur les étiquettes ou sur les pièces servent à la fabrication et ne doivent pas être inscrits sur le présent certificat. NOTE: Any serial numbers that may be present on labels or parts are for manufacturing purposes and need not be referenced on this certificate.						
Le cas échéant, la durée de stockage est précisée sur l'emballage de l'article - When applicable, shelf life is indicated on the item packaging "Le numéro de pièce identifié sur la pièce peut inclure un suffixe alphabétique. Ce suffixe alphabétique dénote le niveau de révision de la pièce. Tous les niveaux de révision pour une pièce donnée sont parfaitement interchangeables. Le niveau de révision d'une pièce n'est pas requis pour ce certificat." "The part number marked on a part may include an alpha suffix. This alpha suffix denotes the part's revision level. All revision levels for a given part number are fully interchangeable. The part's revision level is not required for this certification."						
13a. Le présent bon de sortie certifie que les articles indiqués ci-dessus ont été construits conformément à : Certifies that the items identified above were manufactured in conformity to: ☒ des données de conception approuvées et qu'ils peuvent être utilisés en toute sécurité approved design data and are in condition for safe operation. ☐ des données de conception non approuvées indiquées à la case 12. non approved design data specified in block 12.			14a. ☐ RAC 571.10 (certification après maintenance) - CAR 571.10 Maintenance Release ☐ Autre réglementation précisée à la case 12 - Other regulations specified in block 12. Le présent bon de sortie certifie que, sauf indication contraire à la case 12, le travail indiqué à la case 11 et décrit à la case 12, a été effectué conformément au Règlement de l'aviation canadien. Certifies that unless otherwise specified in block 12, the work identified in block 11 and described in block 12 has been performed in compliance with the Canadian Aviation Regulations.			
13b. Signature   TC 448 NO. 131		13c. Numéro de l'organisme agréé Approved Organization Number 4-58		14b. Signature S/O - N/A		14c. Numéro de l'organisme agréé Approved Organization Number S/O - N/A
13d. Nom - Name I. Drukker		13e. Date (dd/mm/yyyy) 11/AUG/2022		14d. Nom - Name S/O - N/A		14e. Date (dd/mm/yyyy) S/O - N/A

Le monteur doit contre-vérifier l'admissibilité avec les données approuvées - Installer must cross-check eligibility with approved data

RESPONSABILITÉS DU MONTEUR - INSTALLER RESPONSIBILITIES

Le présent bon de sortie ne constitue pas une autorisation de montage. - 1. This certificate does not constitute authority to install.

Le monteur qui travaille conformément à la réglementation d'un pays autre que celui spécifié à la case 1 doit s'assurer que la réglementation en question reconnaît la certification du pays ainsi spécifié. - 2. Installers working in accordance with the national regulations of a country other than that specified in block 1 must ensure that their regulations recognize certifications from country specified.

Les déclarations des cases 13a et 14a ne constituent pas une certification du montage. Dans tous les cas, le dossier technique de l'aéronef doit inclure une certification de montage délivrée conformément à la réglementation nationale qui s'applique, avant que l'aéronef puisse voler. - 3. Statements in block 13a and 14a do not constitute installation certification. In all cases, the technical record for the aircraft must contain an installation certification issued in accordance with the applicable national regulations before the aircraft may be flown.

1. Approving Competent Authority / Country FEDERAL OFFICE OF CIVIL AVIATION SWITZERLAND		2. AUTHORISED RELEASE CERTIFICATE EASA FORM 1		3. Item Tracking Number 80357896 0010  Page 1 of 1	
4. Organisation Name and Address 		Address PILATUS AIRCRAFT LTD CH 6371 STANS SWITZERLAND +41 41 619 61 11		5. Work order / Contract / Invoice Your: PO#LJ-PO-050/SCA/X/2022 / Our : 243314	
6. Item	7. Description	8. Part No.	9. Quantity	10. Serial No.	11. Status / Work
1	BOLT, HEX HD	6201106401	4	N/A	New
12. Remarks FOR ADDITIONAL INFORMATION TO THE ITEM/S LISTED ABOVE, SEE CERTIFICATE OF CONFORMANCE 80357896 This document has been issued according to an approved computer generated signature procedure					
13a. Certifies that the items identified above were manufactured in conformity to: ☒ approved design data and are in condition for safe operation ☐ non approved design data specified in block 12			14a. ☐ Part 145A.50 Release to Service ☐ Other regulation specified in block 12 Certifies that unless otherwise specified in block 12 the work identified in block 11 and described in block 12 was accomplished in accordance with Part 145 and in respect to that work the items are considered ready for release to service.		
13b. Authorized Signature 		13c. Approval / Authorisation Number CH.21G.0002		14b. Authorized Signature	
13d. Name F. Micaletto		13e. Date (dd mm yyyy) 13 Oct 2022		14c. Certificate / Approval Ref. No. 14d. Name 14e. Date (dd mm yyyy)	
USER / INSTALLER RESPONSIBILITIES This certificate does not automatically constitute authority to install the item(s). Where the user/installer performs work in accordance with regulations of an airworthiness authority different than the airworthiness authority specified in block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts items from the airworthiness authority specified in block 1. Statements in block 13a and 14a do not constitute installation certification. In all cases the aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.					

DOCUMENT OFFICIEL - OFFICIAL RECORD

1. Autorité de l'aviation canadienne qui approuve le bon de sortie Approving Civil Aviation Authority/Country Transport Canada		2. BON DE SORTIE AUTORISÉE - AUTHORIZED RELEASE CERTIFICATE FORM ONE		3. Numéro de suivi du formulaire - Form Tracking No. 800732552-000002 4041795	
4. Nom et adresse de l'organisme - Organization Name and Address  Pratt & Whitney Canada Cie. / Pratt & Whitney Canada Corp. 1000 boul. Marie-Victorin Longueuil, Québec Canada J4G 1A1 (450) 677-8411		5. Bon de travail/Contrat/Facture - Work Order/Contract/Invoice 3519811 4500361346			
6. Article - Item	7. Description	8. Numéro de pièce - Part No.	9. Qte. - Qty.	10. Numéro de série/Lot - Serial/Batch No.	11. Situation/Travail - Status/Work
000001	GASKET	ST3276-01	300 EA	N/A	NEW
12. Remarques - Remarks NOTE: Les numéros de série figurant sur les étiquettes ou sur les pièces servent à la fabrication et ne doivent pas être inscrits sur le présent certificat. NOTE: Any serial numbers that may be present on labels or parts are for manufacturing purposes and need not be referenced on this certificate.					
Le cas échéant, la durée de stockage est précisée sur l'emballage de l'article - When applicable, shelf life is indicated on the item packaging "Le numéro de pièce identifié sur la pièce peut inclure un suffixe alphabétique. Ce suffixe alphabétique dénote le niveau de révision de la pièce. Tous les niveaux de révision pour une pièce donnée sont parfaitement interchangeables. Le niveau de révision d'une pièce n'est pas requis pour ce certificat." "The part number marked on a part may include an alpha suffix. This alpha suffix denotes the part's revision level. All revision levels for a given part number are fully interchangeable. The part's revision level is not required for this certification."					
13a. Le présent bon de sortie certifie que les articles indiqués ci-dessus ont été construits conformément à : Certifies that the items specified above were manufactured in conformity to : ☒ des données de conception approuvées et qu'ils peuvent être utilisés en toute sécurité approved design data and are in condition for safe operation. ☐ des données de conception non approuvées indiquées à la case 12. non approved design data specified in block 12.			14a. ☐ RAC 571.10 (certification après maintenance) - CAR 571.10 Maintenance Release ☐ Autre réglementation précisée à la case 12 - Other regulations specified in block 12. Le présent bon de sortie certifie que, sauf indication contraire à la case 12, le travail indiqué à la case 11 et décrit à la case 12, a été effectué conformément au Règlement de l'aviation canadien. Certifies that unless otherwise specified in block 12, the work identified in block 11 and described in block 12 has been performed in compliance with the Canadian Aviation Regulations.		
13b. Signature   TD 4-66 NO. 131		13c. Numéro de l'organisme agréé Approved Organization Number 4 - 58		14b. Signature S/O - N/A	
13d. Nom - Name I. Drukker		13e. Date (dd/mm/yyyy) 17/AUG/2022		14c. Date (dd/mm/yyyy) S/O - N/A	

* Le monteur doit contre-vérifier l'admissibilité avec les données approuvées - *Installer must cross-check eligibility with approved data

RESPONSABILITÉS DU MONTEUR - INSTALLER RESPONSIBILITIES

1. Le présent bon de sortie ne constitue pas une autorisation de montage. - 1. This certificate does not constitute authority to install.

2. Le monteur qui travaille conformément à la réglementation d'un pays autre que celui spécifié à la case 1 doit s'assurer que la réglementation en question reconnaît la certification du pays ainsi spécifié. - 2. Installers working in accordance with the national regulations of a country other than that specified in block 1 must ensure that their regulations recognise certifications from country specified.

3. Les déclarations des cases 13a et 14a ne constituent pas une certification du montage. Dans tous les cas, le dossier technique de l'aéronef doit inclure une certification de montage délivrée conformément à la réglementation nationale qui s'applique, avant que l'aéronef puisse voler. - 3. Statements in block 13a and 14a do not constitute installation certification. In all cases, the technical record for the aircraft must contain an installation certification issued in accordance with the applicable national regulations before the aircraft may be flown.

1. Approving Civil Aviation Authority/Country: FAA/United States		2. AUTHORIZED RELEASE CERTIFICATE FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG		3. Form Tracking Number: 1103309-1	
4. Organization Name and Address: WOODWARD, INC. 5001 North Second Street Rockford, IL 61125 PQ2020CE		5. Work Order/Contract/Invoice Number: Inv. #4727926 P.O. #4500206473 S.O. #3219183			
6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:
2	GOVERNOR - ASSY TURBO PROP	210615	1	20230749	NEW
12. Remarks: EXPORT AIRWORTHINESS APPROVAL. This article meets any special requirements of SWITZERLAND This/these PMA part(s) is/are not a critical component					
DRG.210615rev.D. For installation eligibility, login to "www.woodward.com/airsys/customer/pma_eligibility.cfm"					
13a. Certifies the items identified above were manufactured in conformity to: ☒ Approved design data and are in a condition for safe operation. ☐ Non-approved design data specified in block 12.			13b. ☐ 14 CFR 43.9 Return to Service ☐ Other regulations specified in block 12 Certifies that unless other than specified in block 12, the work specified in block 11 and described in block 12 was accomplished in accordance with the 14 CFR 43.9 of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.		
13b. Authorized Signature: <i>Linda M. Michaelis</i>		13c. Approval/Authorization No: 676699092		14a. Authorized Signature:	
13d. Name(Typed or Printed): Linda M. Michaelis		13e. Date(dd/mm/yyyy): 04/NOV/2015		14b. Approval/Authorization No:	
				14c. Name(Typed or Printed):	
				14d. Date(dd/mm/yyyy):	
User/Installer Responsibilities					
It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.					
Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in block 1.					
Statements in blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.					

1. Approving Competent Authority / Country FEDERAL OFFICE OF CIVIL AVIATION SWITZERLAND		2. AUTHORISED RELEASE CERTIFICATE EASA FORM 1		3. Form Tracking Number 80362001-0010 		Page 1 of 1
4. Organisation Name and Address 		Address PILATUS AIRCRAFT LTD CH-6371 STANS SWITZERLAND Telephone +41 41 619 61 11		5. Work order / Contract / Invoice Your: PO#LJ-PO-050/SCA/X/2022 / Our : 243314		
6. Item	7. Description	8. Part No.	9. Quantity	10. Serial No.	11. Status / Work	
1	BOLT, BEARING	6201114901	4	N/A	New	
12. Remarks FOR ADDITIONAL INFORMATION TO THE ITEM/S LISTED ABOVE, SEE CERTIFICATE OF CONFORMANCE 80362001 This document has been issued according to an approved computer generated signature procedure FROM ELECTRONIC SIGNATURE						
13a. Certifies that the items identified above were manufactured in conformity to: ☒ approved design data and are in condition for safe operation ☐ non-approved design data specified in block 12				14a. ☐ Rep-145A, 56 Release to Service ☐ Other regulation specified in block 12 Certifies that unless otherwise specified in block 12, the work identified in block 11 and described in block 12 was accomplished in accordance with Part 145 and in respect to that work the items are considered ready for release to service.		
13b. Authorised Signature (Electronic Signature on File) 		13c. Approval / Authorisation Number CH.21G.0002		14b. Authorised Signature		14c. Certificate / Approval Ref. No.
13d. Name A. Keller		13e. Date (dd mm yyyy) 25 Nov 2022		14d. Name		14e. Date (dd mm yyyy)
USER / INSTALLER RESPONSIBILITIES This certificate does not automatically constitute authority to install the item(s). Where the user/installer performs work in accordance with regulations of an airworthiness authority different than the airworthiness authority specified in block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts items from the airworthiness authority specified in block 1. Statements in block 13a and 14a do not constitute reculation certification. In all cases the aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.						

1. Approving Civil Aviation Authority/Country: FAA/UNITED STATES		2. AUTHORIZED RELEASE CERTIFICATE FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG			3. Form Tracking Number: 20211015F	
4. Organization Name and Address: Hartzell Propeller Inc. One Propeller Place Piqua, Ohio 45356 (PC-10)					5. Work Order/Contract/Invoice Number: 291336	
6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:	
1	BOLT, MOUNTING, 9/16-18, 12 POINT	B-3339	60	N/A	NEW	
-----END-----						
12. Remarks: Additional Information: Eligibility information for Hartzell Propeller system components is contained in the appropriate Hartzell maintenance manual. This information is also available for Hartzell ice protection system components at www.hartzellprop.com						
Packaging Methods used to protect against corrosion and damage: QC-05-003			Duration of Methods: 6 months			
13a. Certifies the items identified above were manufactured in conformity to: ☒ Approved design data and are in condition for safe operation ☐ Non-approved design data specified in Block 12			14a. ☒ 14 CFR 43.9 Return to Service ☐ Other regulations specified in Block 12 Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.			
13b. Authorized Signature: 		13c. Approval/Authorization No.: ODA-100082-CE	14b. Authorized Signature:		14c. Approval/Certificate No.:	
13d. Name (Typed or Printed): Melissa . Zickafosse		13e. Date (dd/mm/yyyy): 15/Oct/2021	14d. Name (Typed or Printed):		14e. Date (dd/mm/yyyy):	
User/Installer Responsibilities						
It is important to understand that the existence of this document alone does not automatically constitute authority to install the engine/propeller/article. Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness accepts engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1. Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.						