PT. SMART CAKRAWALA AVIATION

WORK ORDER

Form: SCA/MTC/030

Subject : **Engine and Propeller Installation**	No.	WO/012-SNB/XI/2022
	Date	8 Nov 2022
	A/C Reg.	PK-SNB MSN 1015
Reference : MP Pilatus PC-6 AMM Pilatus PC-6	Prepared By	TS
	Checked By	CI
	Approved By	TM

To : Engineer In Charge

Description :

1. Perform Engine and Propeller Installation
2. Make an entry in Maintenance Log.
3. Return the Completed Work Order and Form to PPC.

#If any finding, please close the routine card, and transferred to inspection card.

Additional Work :

- NIL -

Compliance Statement	Sign & Date	Signature
CARRIED OUT ENGINE & PROPELLER INSTALLATION	Company Lic. No.: SCA 019 02 JUNE 23 (Engineer In Charge) AGUS SULAEMAN	(Technical Manager)

AIRCRAFT CHECK WORK SUMMARY
(Form: SCA/MTC/051)

DATE OF ISSUED	JO/WO #	TYPE OF MAINTENANCE	DATE OF ACCOMPLISHED
8 Nov 2022	WO/012-SNB/XI/2022	Engine and Propeller Installation	02 JUNE 2023

A/C Type	Mfg. Serial Number	A/C Registration
PILATUS PORTER PC-6	1015	PK-SNB

AIRCRAFT DATA

Subject	Pos #	Serial Number (SN)	TTSN/TCSN
Engine	#1	PCE-51153	TSN: 18599:8H CSN: 19230
	#2	-	TSO: 0 CSO: 0
Propeller/Rotor	#1	FY4093	TSN: 739:56H TSO: 0
	#2	-	
Landing Gear	NLG		
	LH MLG		
	RH MLG		

PACKAGE COVERED

No	Subject	Qty	Remark
1	Non-Routine Card	-	
2	Inspection Card	-	
3	Work Order	1	
4	Summary Inspection List	1	
5	Material and Tool List	-	
6	Escalation form	-	
7	CRS (SMI / Unscheduled Maintenance)	1	

INSPECTION CARD (IC) LIST (Finding during maintenance)

No	Taskcard Ref	Subject	Status		Name/ Sign & Stamp
			Open	Close	
IC-001					
IC-002					
IC-003					
IC-004					
IC-005					
IC-006					

IC-007					
IC-008					
IC-009					
IC-010					
IC-011					
IC-012					
IC-013					
IC-014					
IC-015					

Prepared by : Technical Support	Checked by : Chief Maintenance	Verified by : Chief Inspector	Approved by : Technical Manager
			
.....	.....	.....	.....
Hani	Dodit	Yanuar	Istiono

smartaviation

SUMMARY INSPECTION ITEMS
(Form: SCA/MTC/050)

WO Ref: WO/012-SNB/XI/2022

NO.	TASK CARD NO.	DESCRIPTION	DATE	EST MHR	NAME	STAMP
1	APPENDIX	PROPELLER INSTALLATION	02 JUNE 2023	48 MHR	AGUS SULAEMAN	
2	APPENDIX	ENGINE INSTALLATION	02 JUNE 2023	144 MHR	AGUS SULAEMAN	

SUMMARY INSPECTION ITEMS
(Form: SCA/MTC/050)

WO Ref: WO/012-SNB/XI/2022

NO.	TASK CARD NO.	DESCRIPTION	DATE	EST MHR	NAME	STAMP
1	APPENDIX	PROPELLER INSTALLATION	02 JUNE 2023	144 M	AGUS SULAEMAN	
2	APPENDIX	ENGINE INSTALLATION	02 JUNE 2023	144 MHR	AGUS SULAEMAN	

PT. SMART CAKRAWALA AVIATION

CERTIFICATE RETURN TO SERVICE

SCHEDULED MAINTENANCE INSPECTION
(CRS-SMI)

A/C TYPE	: PILATUS PORTER PC-6	TTSN	: 476 : 14
A/C REG	: PK-SNB	TCSN	: 580
MSN	: 1015	DATE	: 29 MAY 23

TYPE OF INSPECTION : ENGINE AND PROPELLER INSTALLATION
DUE AT : N/A
REFF : MP PILATUS PC-6

EXCEPTION

-NIL-

AUTHORIZED PERSON

I hereby certify that this aircraft has been maintained accordance with CASR and Maintenance Program. Aircraft safe and airworthy for flight

NAME	CAT	AMEL/OTR NO	SIGN&STAMP	DATE
AGUS SULAEMAN	AIRFRAME & POWER PLANT	6444/SCA 019		02 JUNE 2023
	EIRA			

THE NEXT DUE TYPE OF INSPECTION : 100 HRS INSP

DUE AT : 576 : 14 HOURS

Form: SCA/MTC/049

smartaviation	**INSPECTION CARD** **(Form: SCA/MTC/ 048)**	TECHNICAL DEPARTMENT

1. CARD #	2. JO/WO #	3. ORIGINATOR	4. CARD REF	5. DATE
6. A/C REG/MSN	7. A/C TYPE	8. TRADE	12. VENDOR ORDER #	
9. ZONE	10. STA	11. MTC TYPE		

13. DESCRIPTION/DEFECT-IF FINDING OF CPCP INSPECTION, PLEASE COMPLETE SET 20	14 PPC/ENG	15 DATE

16. CORRECTIVE ACTION	17 MECH	18 ENG. LIC	19 DATE
Performed at A/C TT : ..... A/C TC /LDG :			
20. CORROSION INFORMATION			
LOCATION CORROSION ☐ Isolated ☐ Widespread CORROSION LVL ☐ 1 ☐ 2 ☐ 3 PROPOSED ACTION ☐ Doublers ☐ Others	CAUSE OF DAMAGE ☐ Environment ☐ Internal Leakage ☐ Chemical Spill ☐ LAV/Galley Spill ☐ Blocked Drain ☐ Wet Insulation Blanket ☐ Other			
21. If the defect is RII, Please Sign this card finally by RII Inspector		INSP	DATE
NOTICE OF INSPECTOR			

22. PARTS REQUIRED

PART DESCRIPTION	PART NO	QTY	SERIAL NO		STATUS	
			ON	OFF	CLOSE	OPEN

23. TOOLS REQUIRED

DESCRIPTION	PART NO. / MODEL	NEXT CALIBRATION DATE	STATUS

Ref. AMM Pilatus Porter PC6 Chapter 61-10-00,

PROPELLER INSTALLATION SHEET

Date	:	30 MAY 2023	Propeller P/N	:	D4N00163SKD
Aircraft Registration	:	PK-SNB	Propeller S/N	:	FY4093
Aircraft Type	:	PILATUS PC6, B2-H4	TSN	:	739:56H
Aircraft Total Hours	:	476:14	TSO	:	0
Aircraft Total Cycle	:	580	Work Order Number	:	WO-012-SNB-XI-2022

NO	TASK		SIGNATURE	
			SIGN	STAMP
Job Set Up				
1	Do Engine Run Ground Sheet before Propeller installation, fill out Form No. XXXXXX		N/A	N/A
2	Ensure airplane electrical power is OFF.			
Installation (Ref. Fig. 401)				
	(1) Attach the puller (16) and pull the slip ring fully forward to touch the rear face of the propeller bulkhead.	RII: Sign & Stamp		
	(2) Install a new O-ring (15) on the propeller shaft.	RII: Sign & Stamp		
	(3) Install the sling and put the propeller in position on the engine flange.	RII: Sign & Stamp		
	(4) Lubricate the bolts (7) with grease (Material No. P04-009).	RII: Sign & Stamp		
1	**CAUTION:** MAKE SURE THAT THE CHAMFERED SIDE OF THE WASHER FACES THE BOLT HEAD.			
	(5) Install the attachment bolts (7) and washers (8), with the chamfered side of the washer next to the bolt head.	RII: Sign & Stamp		
	(6) Torque the bolts in three steps as follows: a) To 54 Nm (480 lbf in.), (Ref. Fig. 402, Sequence A). b) To 108 Nm (960 lbf in.), (Ref Fig. 402, Sequence A). c) To between 136 and 142 Nm (1200 and 1260 lbf in.), (Ref. Fig402, Sequence B).	RII: Sign & Stamp		

NO	TASK		SIGNATURE	
			SIGN	STAMP
	(7) Safety the bolts with lockwire (Material No. P02-001).	RII: Sign & Stamp		
	(8) Remove the sling and puller.			
	(9) Put the carbon block (9) in position.	RII: Sign & Stamp		
	(10) Put the reversing lever (12) in position.	RII: Sign & Stamp		
	(11) Put the low pitch switch control rod in position.	RII: Sign & Stamp		
	(12) Install the nut (5) and washer (6).	RII: Sign & Stamp		
	(13) Safety the nut with a new cotter pin (4).	RII: Sign & Stamp		
	(14) Install the pin (3) and washer (11).	RII: Sign & Stamp		
	(15) Safety the pin with a new cotter pin (10).	RII: Sign & Stamp		
	(16) Install the bolt (2), bush (15), washer (14) and nut (13).	RII: Sign & Stamp		
	(17) Install the bolt (2), bush (15), washer (14) and nut (13).	RII: Sign & Stamp		
	(18) Safety the nut with a new cotter pin (1).	RII: Sign & Stamp		
	(19) Make sure the side clearance between the beta-ring and the carbon block (9) is between 0,03 and 0,25 mm (0.001 and 0.01 in). If necessary, remove material from, or replace, the carbon block (9) to get the correct clearance.	RII: Sign & Stamp		
Job Close Up				
1	(1) Install the spinner dome to the bulkhead assembly. Align the marks on the spinner with the marks on the bulkhead.	RII: Sign & Stamp		

NO	TASK	SIGNATURE	
		SIGN	STAMP
	NOTE: The spinner assembly (plate, bulkhead and dome) is balanced as a unit. The components are notinterchangeable.		
	(2) Do a balance check.		
	(3) Do a ground run.		
	(4) Check for oil leaks from the propeller, - No leaks are acceptable.		
	(5) Make sure that the work area is clean and clear of tools and other items.		
	(6) Install the engine cowls PT1 and PB1.		
2	Make an appropriate entry in Work Order or Aircraft Flight Maintenance Log (AFML)		

PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER
AGUS SULAEMAN	ENGINEER		6444
Wahyono	Engineer		5702.
FEBRI HERMAWAN	ENGINEER		6445
PITER - A .	MEKANIK .		

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name	: AGUS SULAEMAN		Name of RII	: FEBRI HERMAWAN	
Signature	:		Signature	:	
Stamp	:		Stamp	:	
Place/Date	: BAYA BIRU / 02 JUNE 23		Place/Date	: BAYA BIRU / 02-JUNE-2023	

Ref. AMM Pilatus Porter PC6 Chapter 71-00-00,
P&WC Maintenance Manual Model PT6A-27 Manual Part No. 3013242 Chapter 72-00-00

ENGINE INSTALLATION SHEET

Date	:	29 MAY 2023	Engine P/N	:	3015000
Aircraft Registration	:	PK-SNB	Engine S/N	:	PCE-51153
Aircraft Type	:	PILATUS PC6, B2-H4	TSN	:	18599:8H
Aircraft Total Hours	:	476:14H	TSO	:	19230
Aircraft Total Cycle	:	580	CSN	:	0
Work Order Number	:	WO-012-SNB-XI-2022	CSO	:	0

NO	TASK	SIGNATURE	
		SIGN	STAMP
Job Set Up			
A	Make inventory records P/N and S/N of the engine and it's accessories from the engine that removed. Fill out into the form no ASI/MD-50 Engine Change- Major Component Inventory Records.		
Install the following item of Engine Accessories on the Engine before proceeding to install on the airplane:			
	CAUTION: MAKE SURE THAT ALL BLANKS AND COVERS REMOVED FROM THE REPLACEMENT ENGINE ARE INSTALLED TO ALL OPENINGS ON THE REMOVED ENGINE		
	Oil Drain Valve (Ref. Fig. 409)		
A	1. Remove the blank from the opening. 2. Install a new O-ring (4) to the oil drain valve (5). 3. Install the drain valve (5) with the lock pin (2). 4. Safety the lockpin (2) with a new cotter pin. 5. Install the bolt (1).		
	Oil Temperature Sensor (Ref. Fig. 409)		
B	1. Remove the blank from the opening. 2. Install a new O-ring (11) to the oil temperature sensor. 3. Install the oil temperature sensor (10).		
	Engine Breather Elbow (Ref. Fig. 409)		
C	1. Remove the blank from the opening. 2. Install a new O-ring (8) to the elbow (9). 3. Install the elbow (9) with the bolts (6) and washers (7). 4. Torque the bolts (6) to between 3.4 and 3.9 Nm (30 and 35 lbf in.). 5. Safety the bolts (6) with lockwire (Material No. P02-001).		

NO	TASK	SIGNATURE	
		SIGN	STAMP
	Control Levers (Ref. Fig. 405)		
D	1. Idle Control Lever: a) Put the lever (9) in position on the idle control unit. b) Install the washer (10) and nut (11), then tighten the nut (11) with your hand c) Install the bolt (6), control rod end (5), spacers (7) and (4), washer (3) and nut (2). d) Torque the nut (2) to between 1.3 and 2.0 Nm (12 and 18 lbf in.). e) Safety the nut (2) with a new cotter pin (1). f) Torque the nut (11) to between 1.3 and 2.0 Nm (12 and 18 lbf in.). g) Safety the nut (11) with a new cotter pin (8). **2. Power Control Input Lever** **CAUTION:** WHEN YOU INSTALL THE LEVER (18) IT MUST BE IN LINE WITH THE FOLLOWER LEVER ON THE SAME SHAFT OF THE CAM BOX. a) Install the lever (18) to the splined shaft. b) Install the bolt (17), the washer (19) and the nut (20). Safety the nut (20) with a new cotter pin (21).		
E	**Engine Driven Fuel Pump (Ref. Fig. 410)** 1. Remove the blank from the opening. 2. Install a new gasket (8). 3. Install the fuel pump (7), washers (5) and nuts (6). 4. Torque the nuts (6) to between 8,5 and 9,6 Nm (75 and 85 lbf in.).		
F	**Gas Generator and Propeller Tachometer Generators (Ref. Fig. 410)** 1. Remove the blank from the opening. 2. Install a new gasket (2). 3. Install the generator (1), washers (3) and nuts (4). 4. Torque the nuts (4) to between 8,5 and 9,6 Nm (75 and 85 lbf in.).		
G	**Torque Indicating System (Ref. Fig. 411, Sheets 1 and 2)** 1. Install a new O-ring (1). 2. Install the adapter (8). 3. Remove the bottom RH nut (6) from the propeller tachometer generator. 4. Install the P-clip (4) and torque transducer (3). 5. Install the nut (6). 6. Torque the nut (6) to between 8,5 and 9,6 Nm (75 and 85 lbf in.). 7. Install the torque pressure pipe (2). 8. Install the case reference pressure hose (7).		
H	**Propeller Overspeed Governor (Ref. Fig. 412)** 1. Install a new gasket (4) on the replacement engine. 2. Install the governor with the washers (1) and nuts (2). 3. Torque the nuts (2) to between 19,2 and 21,5 Nm (170 and 190 lbf in.).		

NO	TASK	SIGNATURE	
		SIGN	STAMP
	Drain Line Adapter, QAD Adapter (if applicable) and Starter-Generator (Ref. Fig. 413)		
I	1. Install the drain adapter (7). 2. Install a new gasket (1). 3. If applicable, install the QAD adapter (2) with the washers (3) and nuts (4). 4. Install the starter-generator (Ref. 24-31-11) but connect the electrical cables after you install the engine (Ref. Para. 7.H.(1))		
	Harness (Ref. Fig. 414)		
J	1. If applicable, install the propeller de-icer unit (Ref. 30-60-01, Page Block 401). 2. If necessary, replace the heat insulating tape installed on the harness between the engine flanges A and C. 3. Put the harness assembly in position on the engine. 4. Install the P-clips to secure the harness. 5. Remove the identity marks and connect the ITT indication leads to the T5 terminal block. 6. If applicable, remove the identity marks and connect the propeller de- icer leads to the de-icer unit terminal block. 7. Connect the electrical plugs at the: i. chip detector. ii. propeller tachometer generator. iii. torque pressure transducer. 8. Remove the two thrust bearing cover attaching nuts. 9. Install the low pitch warning switch mounting bracket and warning switch on to the mounting studs. 10. Install the warning switch linkage block on to mounting stud on the propeller reversing lever. 11. Install the washer and nut to secure the warning switch linkage block, then tighten the nut with your hand. Make sure that the block is free to turn on the stud. 12. Safety the nut with a new cotter pin.		
	Propeller Governor Maximum Stop Screw (Ref. Fig. 405)		
K	1. Check the maximum stop screw (35) on the propeller governor. If the screw is shorter than the stop screw on the removed engine, move the screws as follows: **NOTE:** The long stop screw is used to give the propeller speed of 2000 rpm used on B2H4 aircraft. a. On the removed engine: i. Loosen the locknut. ii. Remove the stop screw, at the same time count the number of turns necessary to remove the stop screw. iii. Make a record of the number of turns. b. Move the short stop screw from the replacement engine to the removed engine. c. Install the long stop screw on the propeller governor of the replacement engine with the same number of turns as recorded at (a) (iii). d. Tighten the stop screw locknut.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
	Inner Bulkheads (Ref. Fig. 415)		
L	1. Put the forward bulkheads (1) and (13) in position. 2. Install the bolts (2), washers (3) and nuts (4). 3. Install the bolts (5), washers (10) and nuts (11). 4. Install the harness slot filler plate on to the engine forward bulkhead. 5. Use caulking putty (Material No. P08-001) to seal the gaps between the harness and the bulkhead. 6. Put the rear bulkheads (6) and (12) in position. 7. Install the bolts (7), washers (8) and nuts (9).		
Installation			
	Engine Mounting Ring (Ref. Fig. 408)		
A	1. Position the mounting ring on the engine to the rear of the attachment points. **NOTE:** Make sure that the support strut attachment lugs on the mounting ring are to the rear. 2. Install the engine lifting sling (Ref. P&WC EMM 72-00-00, SERVICING). **CAUTION:** THE BUSHES THAT ATTACH THE SHOCKMOUNT TO THE MOUNTING FRAME BOLT MUST BE INSTALLED. THE BUSH MUST HAVE THE THICK FLANGE TO THE REAR. IF THE BUSH IS INCORRECTLY INSTALLED, THE ENGINE WILL NOT ALIGN CORRECTLY. 3. Install the gaskets (7), shockmounts (10), washers (6) and bolts (5). 4. Torque the bolts (5) to between 31.0 and 34.0 Nm (275 and 300 lbf in.). 5. Safety the bolts in pairs with lockwire (Material No. P02-001). 6. Lubricate the shanks of the mounting ring attachment bolts (8). 7. Put the mounting ring in position. 8. Install the bolts (8), washers (9) and (12) and nuts (11). 9. Torque the nuts (11) to between 55.0 and 60.0 Nm (486 and 531 lb in.).		
	Install the Engine (Ref. Fig. 408)		
B	1. Lift the engine in position. 2. Install the bolts (16) and washers (13). 3. Torque the bolts (16) to between 46,0 and 51,0 Nm (407 and 450 lb in.). 4. Safety the bolts (16) with locking wire (Material No. P02-001).		

NO	TASK	SIGNATURE	
		SIGN	STAMP
C	**Support Struts (Ref. Fig. 408)** 1. Put the rear bulkhead over the engine. 2. Loosely install 10 bolts (five each side) to attach the upper and lower half of the bulkhead together. 3. Put the support struts in position through the holes in the engine rear bulkhead. 4. Install the bolt (1), washer (2) and nut (3). 5. Torque the nuts (3) to between 10,5 and 12,5 Nm (95 and 110 lb in.). 6. Safety the nuts with new cotter pins (4). 7. Make sure the bonded shim (18) is securely attached to the structure. 8. If the nuts (15) are self-locking nuts: i. Install the bolt (19) washer (17) and nut (15). When you do this measure the rundown torque of the nut (15). ii. Torque the nuts (15) to between 51 and 56 Nm (450 and 500 lb in.) plus the rundown torque. 9. If the nuts (15) are castellated nuts: i. Install the bolt (19), washer (17) and nut (15). ii. Torque the nuts (15) to between 51 and 56 Nm (450 and 500 lb in.). iii. Safety the nuts (15) with new cotter pins (14). 10. Remove the lifting sling from the engine		
D	**Connect the Engine Bulkheads (Ref. Fig. 407)** 1. Install the bolts (6) and washers (7) to attach the upper half of the engine rear bulkhead. 2. Tighten the bolts (4) and washers (3) that attach the support strut seal plates (5) and seals (2).		
E	**Case Drains and P3 Air Bleed (Ref. Fig. 402)** 1. Connect the gas generator case rear drain hose (8) at the rear drain valve. 2. Connect the gas generator case forward drain hose (1) at the engine forward bulkhead. 3. Connect the gas generator case forward drain pipe (7). 4. Install the gasket (6) to the mounting boss on the top of the gas generator case. 5. Put the adapter in position and install the four bolts (4). 6. Connect the P3 bleed air hose (2) to the adapter (5). 7. Install the P-clip (3) on the P3 bleed air hose (2) and engine forward bulkhead. 8. Safety the hose to the lower LH side of the engine mounting ring with a cable tie.		

NO	TASK	SIGNATURE	
		SIGN	STAMP
	Fuel Tubes (Ref. Fig. 404, Sheets 1 and 2)		
F	1. Install a new O-ring (11) on the adapter (12). 2. Install the adapter to the starting flow control unit. 3. Set the adapter to the position noted at 4. E. (10). 4. Hold the adapter and then tighten the adapter locknut. 5. Connect the fuel dump hose (14) to the adapter (12). 6. Install and tighten the worm-drive clamp (13). 7. Connect fuel bleed return hose (10) to starting flow control unit. 8. Install a new O-ring (1) to the adapter (2). 9. Install the adapter (2) to the oil fuel heater. 10. (10) Connect the fuel supply hose (3) to the adapter (2). 11. Connect the drain tube (8) to the LP fuel pump. 12. Install a new O-ring (5) to the adapter (6). 13. Install the adapter (6) to the HP fuel pump. 14. Connect the drain tube (7) to the adapter (6). 15. Connect the inlet hose (4) to the LP fuel pump. 16. Connect outlet hose (9) to the LP fuel pump.		
	Oil Tubes (Ref. Fig. 403)		
G	1. Connect the starter generator drain hose (Ref. Fig. 413): i. Connect the drain hose (5) to the drain adapter (7)below the starter generator drive. ii. Install and tighten the clamp (6). 2. Connect the engine breather hose (3) to the breather elbow. Install and tighten the clamp (2). 3. Connect the oil cooler return hose (1) to the adapter on top of theaccessory gearbox. 4. Connect the engine oil cooler supply hose (7) to the connector on thescavenge pump. 5. Install a new O-ring (4) to the adapter (5) at the RH of the accessorygearbox. 6. Install the adapter (5). 7. Connect the oil pressure transducer hose (6) to the adapter (5).		
	Electrical Connectors (Ref. Fig. 401)		
H	1. Connect the electrical cables (1) to the starter-generator. 2. Connect the electrical plug (8) to the oil temperature sensor. 3. Connect the electrical plug (7) to the FCU sense line heater. 4. Connect the electrical plug (5) to the gas generator tachometergenerator. 5. Connect the electrical plug (2) to the ignition unit. 6. Connect the electrical plug (9) at the RH of the front face of theengine rear bulkhead. 7. If the propeller de-icing system is installed: i. Install electrical plug (10) at the RH front face of the enginerear bulkhead. 8. Install the earth leads (6) to engine bulkhead. Connect bonding lead (3) to engine data plate securing bolt (4).		

NO	TASK	SIGNATURE	
		SIGN	STAMP
	Connect the Propeller Control Cable (Ref. Fig. 406)		
I	1. Insert the cable through the hole in the engine rear bulkhead. 2. Install the two grommets to the cable. 3. Insert the cable through the hole in the forward bulkhead. 4. Position the cable on the engine. 5. Press the grommets in the holes of the bulkhead. 6. Put the support bracket (9) in position. 7. Install the bolts (12), washers (10) and nuts (11). 8. Put the cable in position on the support bracket (9). 9. Install the top plate (15) with the bolts (16), spring washers (14) andnuts (13).		
J	**Connect the Controls (Ref. Fig. 405)** 1. Insert the cable through the hole in the engine rear bulkhead. 2. Install the two grommets to the cable. 3. Insert the cable through the hole in the forward bulkhead. 4. Position the cable on the engine. 5. Press the grommets in the holes of the bulkhead. 6. Put the support bracket (9) in position. 7. Install the bolts (12), washers (10) and nuts (11). 8. Put the cable in position on the support bracket (9). 9. Install the top plate (15) with the bolts (16), spring washers (14) andnuts (13). RII : Sign & Stamp		
K	**Connect the Controls (Ref. Fig. 405)** **1. Idle Control:** i. Put the rod (15) in position and then install the bolt (16),washer (14) and nut (13). ii. Torque the nut (13) to between 1,3 and 2,0 Nm (12 and 18 lbfin.). iii. Safety the nut with a new cotter pin. **2. Power Control:** i. Put the rod end (25) in position and then install the bolt (26),washer (24) and nut (23). ii. Torque the nut (23) to between 1,3 and 2,0 Nm (12 and 18 lbfin.). iii. Safety the nut with a new cotter pin (22). **3. Propeller Control:** i. Put the control rod (31) in position and then install the bolt(34), washer (33), spacer (32), washer (29) and nut (26). ii. Torque the nut (26) to between 1,3 and 2,0 Nm (12 and 18 lbfin.). iii. Safety the nut (26) with a new cotter pin (22). RII : Sign & Stamp		

NO	TASK		SIGNATURE	
			SIGN	STAMP
Job Close Up				
A	1. Adjust the engine controls.	RII:		
	2. Install the exhaust stubs.			
	3. Install the propeller.	RII:		
	4. If applicable, install the oil cooler.		N/A	N/A
	5. Fill the engine oil system.			
	6. If installed, remove the aircraft tail stand and lower aircraft tail.			
	7. Make sure that the work area is clean and clear of tools and otheritems.			
	8. Do a fuel system de-preservation.			
	9. Remove the engine compressor intake cover.			
	10. Install the engine cowlings.			
	11. Connect the aircraft battery.			
	12. Do an engine ground run and fill out Engine Ground Run Sheet.			
B	Make an appropriate entry in Work Order or Aircraft Flight Maintenance Log (AFML)			

PERSONNEL PARTICIPATING IN THIS INSPECTION			
NAME	POSITION	SIGNATURE	LICENSE NUMBER
AGUS SULAEMAN	ENGINEER		6999
Wahyono	Engineer		5792.
FERRI HERMAWAN	ENGINEER		6445.

RETURN TO SERVICE

The work recorded above has been carried out in accordance with the requirements of the Civil Aviation Safety Regulation for the time being in force and in that respect the aircraft is consider fit for Release to Service.

Name	:	AGUS SULAEMAN	Name of RII	:	FEBRI HERMAWAN
Signature	:		Signature	:	
Stamp	:		Stamp	:	
Place/Date	:	BAYABIRU / 02 JUNE 23	Place/Date	:	BAYABIRU / 02-JUNE-2023

smartaviation	TECHNICAL SUPPORT TECHNICAL DEPARTMENT **ENGINEERING ORDER**	Rev. No	Original
		Rev. Date	

ENGINE CHANGE - Major Component Inventory Record			
Registration	: PK-SNB	**Work Order Number**	: WO/012-SNB/X1/2022
Airframe Time	: A76 :1A	**Airframe Landings**	: 580
Engine Time	: 18599:8 H	**Engine Cycle**	: 19230

	Engine OFF			**Engine ON**		
Description	**Part Number**	**Serial Number**	**Time Remaining**	**Part Number**	**Serial Number**	**Time Remaining**
Engine Assembly	3015000	PCE-PG0565	SEND TO SHOP	3015000	PCE-51153	NEW OVERHAUL
Propeller Assembly	HC-DAN-3P/09511F	FY4513		DAN00163SKD	FY4093	
Compressor Bleed Valve	3075033-01	AHx1501424		3100829-03	6C820	
Fuel Control Unit	3049635-02	C22477		3244786-8-7	C22007	
Oil Fuel Heater	3032710	WA52186		10552E	565	
Igniter Exciter	3035889G	NNA15340030		10-381550-A	NNA08455077	
Flow Divider /CONTROL STARTING	3019109	20105		3019109	13228	
Oil Cooler	15060730	45002095 14		15060730	45002095 14	
Starter Generator	23048-021	P1046		23048-021	P1046	
Alternator	N/A	N/A		N/A	N/A	
Fuel Pump	3034794	00451		025323-101-03	5349	
Propeller Governor	3045798-02	20135541		66503-8210-4120	16980915	
Propeller Overspeed Governor	210615	20000956		210615	20230749	
Fuel Nozzle	3011151B			3011151B		

NOTE: ANY OTHER COMPONENT CHANGES MUST BE FILL ON INSPECTION CARD (SCA/MTC/048)

ENGINE GROUND RUN CHECK SHEET - PT6A-27 ENGINE
WITH FOUR BLADE PROPELLER (HARTZELL STC SA377CH)

PILATUS

HARTZELL

WORK ORDER NO.		: WO/012-SNB/XI/2022	
Aircraft Registration	PK-SNB	Aircraft Total Hours	476:14
Aircraft Serial No.	1015	Aircraft Total Landings	580
Engine Serial No.	PCE-51153	Engine TSN / TSO	18599:8 H/O
Propeller Serial No	FY4093	Propeller TSN / TSO	739:56 H /O
Ambient Temp	33 °C	FBP (Field Barometric Pressure)	29.91 In.Hg
Date	02 JUNE 23	Time	15:50 LT
Mechanic / Engineer		Authorized Engineer	AGUS SULAEMAN
Reason For Ground Run		TEST PERFORMANCE AFTER ENGINE INSTALLED.	

Checks to be carried out. No:	1 2 4 5 7 8 9 10 11 12 13 14 15

Engine Ground Run Check Frequency

Check Number	1	2	3	4	5	6	7	8	9	10	11	12	13	14	15
Each 100 / Yearly	x	x		x			x	x			x	x	x	x	x
Each 200									x						
Pre-Complete Overhaul	x	x	x	x		x	x	x	x	x	x	x	x	x	x
After Short Term Storage															x
After Long Term Storage	x	x	x	x		x	x	x	x	x	x	x	x	x	x

In additional the following check must be carried out after Installation, Repair and Adjustment of any ofthe following components.

Check Number	1	2	3	4	5	6	7	8	9	10	11	12	13	14	15
Engine Installation ✓	x	x	x	x	x	x	x	x	x	x	x	x	x	x	x
Propeller Installation		x	x	x	x			x							
Fuel Control Unit	x				x	x	x	x		x	x				
HP Fuel Pump						x	x								
Fuel Nozzle						x	x								
Starting Flow Control	x				x		x	x							
Emer Fuel Control Actuator											x				
Prop Governor	x		x	x	x		x	x							
Prop Overspeed Governor									x						
Compressor Bleed Valve						x	x								
Engine Controls	x			x	x			x	x						
Low Pitch Warning Switch				x											
Suction Components														x	

Use this sheet's to record engine run result, use in conjunction with task cards.

NO.	CHECK	TARGET	ACTUAL
ENGINE START	ITT (Troubleshoot If More Than 925□C)	Max. 1090 □C	623 □C
	Cabin Heat	OFF	✓ OK?
1	Low Idle (Minimum Governing) Speed	51 - 53 % Ng	53 % Ng
	Fuel Pressure / Boost Pump OFF	Light out or 25 ± 5 psi	✓ OK?
	ITT		536 □C
	Oil Pressure		93 psi
	Oil Temperature		68 □C
2	Propeller Governor		
	Maximum Np	1980 - 2000 rpm (90.0 - 90.9 %)	1980 rpm
	Py Disconnected		90.3 % Ng
	Py Connected		90.3 % Ng
	Difference	Maximum 0.3% Ng	0 %
	Airbleed Link at Minimum	1900 - 1950 rpm (86.4 - 88.6 %)	1950 rpm
	Aircraft with SB 161:		
	Propeller Control Lever at Minimum	1880 - 1900 rpm (85.5 - 86.4 %)	rpm
3	Propeller Fine Pitch Setting (High Idle)		
	Target Torque	psi	7.0 psi
	Power Lever to Give Np	1694 rpm (77 %)	1670 rpm
	Basic High Idle	68 - 72% Ng	68.7 %Ng
4	Propeller Low Pitch Warning		
	PCL from Reverse to Detent	Light OFF	
		1 to 2 mm before Detent	2 mm
5	Minimum Pitch in Flight		
	Ng	67 - 73 %	70 % Ng
	Np	1800 - 1950 rpm (81.8 - 88.6 %)	1900 rpm
	Torque	4 - 7 psi	7 psi
6	FCU Maximum Governing Speed (Ng)	97.1 % Ng	97.6 % Ng
	(Trim stop deployed)		

NO.	CHECK	TARGET	ACTUAL
7	Engine Performance	Ref: AMM 71-00-00	
	Target Torque Pressure	41 psi	41 psi
	Fuel flow (Actual minus 23 lb / hr or 3.4 gal / hr)	55 gal/hr 342 lb / hr	44.6 /262 lb / hr
	Target Ng	95.4 % Ng	96.0 % Ng
	Maximum ITT	697 °C	625 °C
8	Reverse Power Setting		
	Np	1880 - 1925 rpm (85.5 - 87.5 %)	1900 rpm
	Torque	psi	18. psi
9	Propeller Overspeed Governor		
	Test Lever Selected to:		
	TEST	1880 - 1920 rpm (85.5 - 87.3 %)	1920 rpm
	NORMAL	1980 - 2000 rpm (90.0 - 90.9 %)	1980 rpm
10	Acceleration		
	64 % – 90 % Ng	2.5 – 4 secs	3 secs
	Deceleration		
	85% to 60% Ng or low idle speed(Whichever comes first)	Maximum 6-12 sec (Dependent upon altitude)	7 secs altitude (kFt)
	Manual Override (MOR) (Aircraft with SB 164)		
	Use Toggle Switch In Small Increment (REF. to WARNINGS and CAUTIONS in Check 11)	Increase to 15% above Idle (Max Increase less than 4 % per Second)	✓ OK?
		Decrease To Idle (Max Decrease less Than 4% per Second)	✓ OK?
12	Oil Pressure	80 -100 psi	92 psi
13	Generator (Ref. 24-30-00)	Online by 60% Ng	60 % Ng
14	Suction (High Idle)	4.5 – 5.2 in. Hg	N/A in. Hg
15	Engine Rundown Time After Stop	MIN 30 secs	33 secs
Additional			
Generator Check (High Idle Under Load)		27.75 – 28.25 VDC	28.2 VDC
After Engine Run			
Check Eng. For Signs of Fuel/Oil/Air Leaks		NO LEAKS FOUND	✓ OK?
Safety All Screws, Bolts, Locknuts as Req.			✓ OK?

ENGINE GROUND RUN CHECK SHEET - PT6A-27 ENGINE
WITH FOUR BLADE PROPELLER (HARTZELL STC SA377CH)

PILATUS

WORK ORDER NO.		: WO/012-SNB/XI/2022	
Aircraft Registration	PK-SNB	Aircraft Total Hours	476:14
Aircraft Serial No.	1015	Aircraft Total Landings	580
Engine Serial No.	PCE-51153	Engine TSN / TSO	18599:8 H / 0
Propeller Serial No	FY4093	Propeller TSN / TSO	739:56 H / 0
Ambient Temp	33 °C	FBP (Field Barometric Pressure)	29.91 In.Hg
Date	02 JUNE 23	Time	15:50 LT.
Mechanic / Engineer		Authorized Engineer	AGUS SULAEMAN
Reason For Ground Run		TEST PERFORMANCE AFTER PROP INSTALLED.	

Checks to be carried out. No:	1 2 4 5 7 8 9 10 11 12 13 14 15

Engine Ground Run Check Frequency

Check Number	1	2	3	4	5	6	7	8	9	10	11	12	13	14	15
Each 100 / Yearly	x	x		x			x	x			x	x	x	x	x
Each 200									x						
Pre-Complete Overhaul	x	x	x	x		x	x	x	x	x	x	x	x	x	x
After Short Term Storage															x
After Long Term Storage	x	x	x	x		x	x	x	x	x	x	x	x	x	x

In additional the following check must be carried out after Installation, Repair and Adjustment of any ofthe following components.

Check Number	1	2	3	4	5	6	7	8	9	10	11	12	13	14	15
Engine Installation	x	x	x	x	x	x	x	x	x	x	x	x	x	x	x
Propeller Installation ✓		x	x	x	x			x							
Fuel Control Unit	x				x	x	x	x		x	x				
HP Fuel Pump						x	x								
Fuel Nozzle						x	x								
Starting Flow Control	x				x		x	x							
Emer Fuel Control Actuator											x				
Prop Governor	x		x	x	x		x	x							
Prop Overspeed Governor									x						
Compressor Bleed Valve						x	x								
Engine Controls	x			x	x			x	x						
Low Pitch Warning Switch				x											
Suction Components														x	

Use this sheet's to record engine run result, use in conjunction with task cards.

NO.	CHECK	TARGET	ACTUAL
	ENGINE START ITT (Troubleshoot If More Than 925□C)	Max. 1090 □C	□C
	Cabin Heat	OFF	OK?
1	Low Idle (Minimum Governing) Speed	51 - 53 % Ng	% Ng
	Fuel Pressure / Boost Pump OFF	Light out or 25 ± 5 psi	OK?
	ITT		□C
	Oil Pressure		psi
	Oil Temperature		□C
(2)	Propeller Governor		
	Maximum Np	1980 - 2000 rpm (90.0 - 90.9 %)	1980 rpm
	Py Disconnected		90.3 % Ng
	Py Connected		90.3 % Ng
	Difference	Maximum 0.3% Ng	0 %
	Airbleed Link at Minimum	1900 - 1950 rpm (86.4 - 88.6 %)	1950 rpm
	Aircraft with SB 161:		
	Propeller Control Lever at Minimum	1880 - 1900 rpm (85.5 - 86.4 %)	rpm
(3)	Propeller Fine Pitch Setting (High Idle)		
	Target Torque	psi	7.0 psi
	Power Lever to Give Np	1694 rpm (77 %)	1670 rpm
	Basic High Idle	68 - 72% Ng	68.7 %Ng
(4)	Propeller Low Pitch Warning		
	PCL from Reverse to Detent	Light OFF	
		1 to 2 mm before Detent	2 mm
(5)	Minimum Pitch in Flight		
	Ng	67 - 73 %	70 % Ng
	Np	1800 - 1950 rpm (81.8 - 88.6 %)	1900 rpm
	Torque	4 - 7 psi	7 psi
6	FCU Maximum Governing Speed (Ng) (Trim stop deployed)	97.1 % Ng	% Ng

NO.	CHECK	TARGET	ACTUAL
7	Engine Performance	Ref: AMM 71-00-00	
	Target Torque Pressure	psi	psi
	Fuel flow (Actual minus 23 lb / hr or 3.4 gal / hr)	lb / hr	lb / hr
	Target Ng	% Ng	% Ng
	Maximum ITT	^0C	^0C
8	Reverse Power Setting		
	Np	1880 - 1925 rpm (85.5 - 87.5 %)	1900 rpm
	Torque	psi	18 psi
9	Propeller Overspeed Governor		
	Test Lever Selected to:		
	TEST	1880 - 1920 rpm (85.5 - 87.3 %)	rpm
	NORMAL	1980 - 2000 rpm (90.0 - 90.9 %)	rpm
10	Acceleration		
	64 % – 90 % Ng	2.5 – 4 secs	secs
	Deceleration		
	85% to 60% Ng or low idle speed(Whichever comes first)	Maximum 6-12 sec (Dependent upon altitude)	secs altitude (kFt)
	Manual Override (MOR) (Aircraft with SB 164)		
	Use Toggle Switch In Small Increment (REF. to WARNINGS and CAUTIONS in Check 11)	Increase to 15% above Idle (Max Increase less than 4 % per Second)	OK?
		Decrease To Idle (Max Decrease less Than 4% per Second)	OK?
12	Oil Pressure	80 -100 psi	psi
13	Generator (Ref. 24-30-00)	Online by 60% Ng	% Ng
14	Suction (High Idle)	4.5 – 5.2 in. Hg	in. Hg
15	Engine Rundown Time After Stop	MIN 30 secs	secs
Additional			
Generator Check (High Idle Under Load)		27.75 – 28.25 VDC	VDC
After Engine Run			
Check Eng. For Signs of Fuel/Oil/Air Leaks		NO LEAKS FOUND	OK?
Safety All Screws, Bolts, Locknuts as Req.			OK?

AUTHORIZED RELEASE CERTIFICATE

FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG

1. Approving Civil Aviation Authority/Country:	2.	3. Form Tracking Number:
FAA/United States		T6943-1
4. Organization Name and Address:		5.Work Order/Contract/Invoice Number:
Turbines, Inc. 7303 Maynard Wheeler Lane. Terre Haute International Airport. Terre Haute, IN 47803		T6943

6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:
1	PT6A-27 Turboprop Engine	3015000	1	PCE-51153	Repaired

12. Remarks:
Partially disassembled. Cleaned RGB, AGB and Oil Tank. Configured to BS 877 (Pilatus). Assembled and Functionally Tested Engine. Test results satisfactory. Reference Overhaul Manual PN 3013243 Rev. 45.

Total Time: 18599.8 Hrs. **Total Cycles:** 19230 **TSO:** 0.0 Hrs. **CSO:** 0

13a. Certifies the items identified above were manufactured in conformity to:

☐ Approved design data and are in condition for safe operation.

☐ Non-approved design data specified in Block 12.

14a. ☒ 14 CFR 43.9 Return to Service ☐ Other regulation specified in Block 12

Certifies that unless otherwise specified in block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

13b. Authorized Signature:	13c. Approval/Authorization No.:	14b. Authorized Signature:	14c. Approval/Certificate No.:
			UTBR513L
13d. Name (Typed or Printed):	13e. Date (dd/mmm/yyyy):	14d. Name (Typed or Printed): Jay Streb	14e. Date (dd/mmm/yyyy): 10 JUN 2022

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.

Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

1. Approving Civil Aviation Authority/Country: FAA/UNITED STATES

2.

AUTHORIZED RELEASE CERTIFICATE

FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG

3. Form Tracking Number: 20220304AT

4. Organization Name and Address: Hartzell Propeller Inc. One Propeller Place Piqua, Ohio 45356 (PC-10)

5. Work Order/Contract/Invoice Number: 295269

6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:
1	O - RING	C-3317-230	200	N/A	NEW
2	O - RING	C-3317-251	50	N/A	NEW
3	DE-ICE BOOT	4H2271-10	60	N/A	NEW
4	BEARING,NEEDLE,CLOSED END	B-6680	100	N/A	NEW
5	DOWEL PIN, 3/8 "	B-6260	50	N/A	NEW
6	FITTING, LUBRICATION, 65°	109032	50	N/A	NEW
		END			

12. Remarks:

Additional Information: Eligibility information for Hartzell Propeller system components is contained in the appropriate Hartzell maintenance manual. This information is also available for Hartzell ice protection system components at www.hartzellprop.com

Packaging Methods used to protect against corrosion and damage: QC-05-003 **Duration of Methods:** 6 months

13a. Certifies the items identified above were manufactured in conformity to:

- [x] Approved design data and are in condition for safe operation
- [ ] Non-approved design data specified in Block 12

14a. [ ] 14 CFR 43.9 Return to Service [ ] Other regulations specified in Block 12

Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

13b. Authorized Signature: Melissa Zickafoose

13c. Approval/Authorization No.: ODA-100082-CE

13d. Name (Typed or Printed): Melissa . Zickafoose

13e. Date (dd/mmm/yyyy): 04/Mar/2022

14b. Authorized Signature:

14c. Approval/Certificate No.:

14d. Name (Typed or Printed):

14e. Date (dd/mmm/yyyy)

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the engine/propeller/article. Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness accepts engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1. Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

NSN: 0052-00-012-9059

FAA Form 8130-3 (02-14)

1. Approving Civil Aviation Authority/Country:	2. AUTHORIZED RELEASE CERTIFICATE FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG	3. Form Tracking Number:
United States		22-1262
4. Organization Name and Address:	Air Prop Specialist 3671 Industrial Park Drive Marianna, FL 32446	5. Work Order/Contract/Invoice Number: 22-1262

6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:
1	Hartzell Propeller	D4N00163SKD	1	FY4093	Overhauled

12. Remarks:

Overhauled propeller in accordance with Hartzell Manuals: 143A, 133C, 159, 202A. All other current service information and Airworthiness Directives complied with as of this date. NOTICE: An Airworthiness Directive may apply to the installation of the propeller described hereon. The installer is responsible for ensuring complete compliance with any applicable installation Airworthiness Directive.

Knockdown propeller in accordance with Hartzell Manuals. Shipped to customer disassembled. The prop must be assembled by trained personnel.

	TSN	TSO
Propeller	739.56	0.00
Hub S/N - FY4093	739.56	0.00
Blade # 1 - L47095	739.56	0.00
Blade # 2 - L47104	739.56	0.00
Blade # 3 - L47096	739.56	0.00
Blade # 4 - L47105	739.56	0.00

13a. Certifies the items identified above were manufactured in conformity to:

- ☐ Approved design data and are in a condition for safe operation.
- ☐ Non-approved design data specified in Block 12

14a. ☑ 14 CFR 43.9 Return to Service ☐ Other regulation specified in Block 12

Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title, 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

13b. Authorized Signature:	13c. Approval/Authorization No.:	14b. Authorized Signature:	14c. Approval/Certificate No.: 1X4R274M
13d. Name (Typed or Printed):	13e. Date (dd/mmm/yyyy):	14d. Name (Typed or Printed): Christopher Giersberg	14e. Date (dd/mmm/yyyy): 19 May 2022

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.

Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

1. Autorité de l'aviation civile/Pays qui approuve le bon de sortie Approving Civil Aviation Authority/Country	2. BON DE SORTIE AUTORISÉE - AUTHORIZED RELEASE CERTIFICATE	3. Numéro de suivi du formulaire - Form Tracking No.
Transport Canada	**FORM ONE**	800653792-000001 3986596

4. Nom et adresse de l'organisme - Organization Name and Address	5. Bon de travail/Contrat/Facture - Work Order/Contract/Invoice
Pratt & Whitney Canada Cie. / Pratt & Whitney Canada Corp. 1000 boul. Marie-Victorin Longueuil, Québec Canada J4G 1A1 (450) 677-9411	3488865 4500355275

6. Article - Item	7. Description	8. Numéro de pièce - Part No.	9. Qtés. - Qty.	10. Numéro de série/de lot - Serial/Batch No.	11. Situation/Travail - Status/Work
000001	GASKET	ST3277-01	100 EA	N/A	NEW

12. Remarques - Remarks

NOTE: Les numéros de série figurant sur les étiquettes ou sur les pièces servent a la fabrication et ne doivent pas être inscrits sur le présent certificat.
NOTE: Any serial numbers that may be present on labels or parts are for manufacturing purposes and need not be referenced on this certificate.

Le cas échéant, la durée de stockage est précisée sur l'emballage de l'article -When applicable,shelf life is indicated on the item packaging

"Le numéro de pièce identifié sur la pièce peut inclure un suffixe alphabétique. Ce suffixe alphabétique dénote le niveau de revision de la pièce. Tous les niveaux de revision pour une pièce donnée sont parfaitement interchangeables. Le niveau de revision d'une pièce n'est pas requis pour ce certificat."

"The part number marked on a part may include an alpha suffix.This alpha suffix denotes the parts's revision level.All revision levels for a given part number are fully interchangeable.The part's revision level is not required for this certification."

13a. Le présent bon de sortie certifie que les articles indiqués ci-dessus ont été construits conformément à :
Certifies that the items identified above were manufactured in conformity to :

- [X] des données de conception approuvées et qu'ils peuvent être utilisés en toute sécurité
  approved design data and are in condition for safe operation.
- [ ] des données de conception non approuvées indiquées à la case 12.
  non approved design data specified in block 12.

14a.
- [ ] RAC 571.10 (certification après maintenance) - CAR 571.10 Maintenance Release
- [ ] Autre réglementation précisée à la case 12 - Other regulations specified in block 12

Le présent bon de sortie certifie que, sauf indication contraire à la case 12, le travail indiqué à la case 11 et décrit à la case 12, a été effectué conformément au *Règlement de l'aviation canadien*.
Certifies that unless otherwise specified in block 12, the work identified in block 11 and described in block 12 has been performed in compliance with the *Canadian Aviation Regulations*.

13b. Signature	13c. Numéro de l'organisme agréé Approved Organization Number	14b. Signature	14c. Numéro de l'organisme agréé Approved Organization Number
Chetan Patel T.C. 4-58 NO. 087	4 - 58	S/O - N/A	S/O - N/A
13d. Nom - Name **C. Patel**	13e. Date (dd/mm/yyyy) 19/MAY/2022	14d. Nom - Name S/O - N/A	14e. Date (dd/mm/yyyy) S/O - N/A

* Le monteur doit contre-vérifier l'admissibilité avec les données approuvées - *Installer must cross-check eligibility with approved data*

RESPONSABILITÉS DU MONTEUR - INSTALLER RESPONSIBILITIES

1. Le présent bon de sortie ne constitue pas une autorisation de montage. - 1. This certificate does not constitute authority to install.

2. Le monteur qui travaille conformément à la réglementation d'un pays autre que celui spécifié à la case 1 doit s'assurer que la réglementation en question reconnaît la certification du pays ainsi spécifié. - 2. Installers working in accordance with the national regulations of a country other than that specified in block 1 must ensure that their regulations recognize certifications from country specified.

3. Les déclarations des cases 13a et 14a ne constituant pas une certification du montage. Dans tous les cas, le dossier technique de l'aéronef doit inclure une certification de montage délivrée conformément à la réglementation nationale qui s'applique, avant que l'aéronef puisse voler. - 3. Statements in block 13a and 14a do not constitute installation certification. In all cases, the technical record for the aircraft must contain an installation certification issued in accordance with the applicable national regulations before the aircraft may be flown.

P&WC JR3-0054-E (2021-03)

DOCUMENT OFFICIEL – OFFICIAL RECORD

1. Approving Competent Authority / Country	2.	3. Form Tracking Number
FOCA/Switzerland	**AUTHORISED RELEASE CERTIFICATE** **EASA FORM 1**	AST-23-112 Page 1 of 1

4. Approved Organisation Name and Address			5. Work Order/Contract/Invoice
PILATUS	PILATUS AIRCRAFT LTD CH-6371 STANS SWITZERLAND	Telephone +41 41 619 61 11 Telefax +41 41 619 65 76 Website www.pilatus-aircraft.com	5650451

6. Item	7. Description	8. Part No.	9. Quantity	10. Serial No.	11. Status/Work
1	GOVERNOR ASSEMBLY, TURBOPROP	8210-412-01	1	16980915	INSPECTED/TESTED

12. Remarks
GOVERNOR, PROPELLER INSPECTED AND TESTED I.A.W. CMM WG60044 REV. 3 1997-09-03
THIS ITEM IS FOR 2000 PROPELLER RPM CONFIGURATION (PC-6 B2/H4) ONLY

TSN: 0.00 HRS.
TSO: N/A

ITEM FROM STORE

Canadian Approval No. TCCA 898-37
The work identified in Block 11 and described herein has been accomplished in accordance with 14 CFR part 43 and in respect to that work, the items are approved for return to service under Certificate no. 8UKY006D

13a. Certifies that the items identified above were manufactured in accordance in conformity to:
- ☐ approved design data and are in condition for safe operation
- ☐ non-approved design data specified in block 12

14a. ☑ Part-145.A.50 Release to Service ☑ Other regulation specified in block 12

Certifies that unless otherwise specified in block 12, the work identified in block 11 and described in block 12 was accomplished in accordance with Part-145 and in respect to that work the items are considered ready for release to service.

13b. Authorised Signature	13c. Approval/Authorisation Number	14b. Authorised Signature Meyer Ch. CH-2591	14c. Certificate/Approval Ref. No. **CH.145.0209**
13d. Name	13e. Date	14d. Name Ch.Meyer CH-2591	14e. Date 24/Feb/2023

USER / INSTALLER RESPONSIBILITIES

NOTE: 1 This certificate does not automatically constitute authority to install the item(s).

2 Where the user/installer perform work in accordance with regulations of an airworthiness authority different than the airworthiness authority specified in block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts items from the airworthiness authority specified in block 1.

3 Statements in blocks 13a and 14a do not constitute installation certification. In all cases aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

1. Approving Civil Aviation Authority/Country:	2.	3. Form Tracking Number:
FAA/United States	**AUTHORIZED RELEASE CERTIFICATE** **FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG**	T6110-90

4. Organization Name and Address:	5.Work Order/Contract/Invoice Number:
Turbines, Inc. 7303 Maynard Wheeler Lane. Terre Haute International Airport. Terre Haute, IN 47803	T6110

6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:
1	Cam block	NPN	1	NSN	Assembled

12. Remarks:
Assembled Cam block reference to PWC Overhaul Manual 3021243.

Turbines, Inc. certifies that the work specified in block 11/12 was carried out in accordance with EASA Part-145 and in respect to that work the article is considered ready for release to service under EASA Part-145 Approval Number: EASA.145.6780.

13a. Certifies the items identified above were manufactured in conformity to:		14a. ☒ 14 CFR 43.9 Return to Service ☒ Other regulation specified in Block 12	
☐ Approved design data and are in condition for safe operation. ☐ Non-approved design data specified in Block 12.		Certifies that unless otherwise specified in block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.	
13b. Authorized Signature:	13c. Approval/Authorization No.:	14b. Authorized Signature:	14c. Approval/Certificate No.: UTBR513L
13d. Name (Typed or Printed):	13e. Date (dd/mmm/yyyy):	14d. Name (Typed or Printed): Dustin Wincinger	14e. Date (dd/mmm/yyyy): 28/FEB/2023

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.

Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

CERTIFICATE OF CONFORMANCE

Certificate Ref. No.
80370407

PILATUS

Address Certified to EN 9100/ISO 14001 standards
PILATUS AIRCRAFT LTD
CH-6371 STANS
SWITZERLAND

Telephone +41 41 619 61 11
Telefax +41 41 619 64 93

Your Work Order / Contract / Pilatus Reference
PO-433 / 249021

Item	Description Vendor Part No. Cure / Expire Date	Part No.	Unit of Supply	Qty.	Serial / Batch No.
00010	GOVERNOR,PROPELLER 3045798-02 Man. Date: 10.10.2012 Exp. Date: 09.09.2018	9682011544	ST	1	16980915
	END				

REMARKS

Certified that the whole of the supplies detailed on all pages of this certificate have been manufactured, inspected and unless otherwise stated above, conform in all respects to the specifications, drawings and contract/order relevant thereto.

Name	Date (d/m/y)	Authorised Signature (Electronic Signature on File)
F. Antonazzo	1.MAR.2023	

1. Autorité de l'aviation civile/Pays qui approuve le bon de sortie Approving Civil Aviation Authority/Country	2. BON DE SORTIE AUTORISÉE - AUTHORIZED RELEASE CERTIFICATE	3. Numéro de suivi du formulaire - Form Tracking No.
Transport Canada	**FORM ONE**	800727997-000010 4038542

4. Nom et adresse de l'organisme - Organization Name and Address	5. Bon de travail/Contrat/Facture - Work Order/Contract/Invoice
Pratt & Whitney Canada Cie. / Pratt & Whitney Canada Corp. 1000 boul. Marie-Victorin Longueuil, Québec Canada J4G 1A1 (450) 677-9411	3491940 4500357246

6. Article - Item	7. Description	8. Numéro de pièce - Part No.	9. Qtés - Qty.	10. Numéro de série/de lot - Serial/Batch No.	11. Situation/Travail - Status/Work
000001	GASKET,OVERSPEED GOVERNOR	3118953-01	20 EA	N/A	NEW

12. Remarques - Remarks

NOTE: Les numéros de série figurant sur les étiquettes ou sur les pièces servent a la fabrication et ne doivent pas être inscrits sur le présent certificat.
NOTE: Any serial numbers that may be present on labels or parts are for manufacturing purposes and need not be referenced on this certificate.

Le cas échéant, la durée de stockage est précisée sur l'emballage de l'article -When applicable,shelf life is indicated on the item packaging

"Le numéro de pièce identifié sur la pièce peut inclure un suffixe alphabétique. Ce suffixe alphabétique dénote le niveau de revision de la pièce. Tous les niveaux de revision pour une pièce donnée sont parfaitement interchangeables. Le niveau de revision d'une pièce n'est pas requis pour ce certificat."

"The part number marked on a part may include an alpha suffix.This alpha suffix denotes the parts's revision level.All revision levels for a given part number are fully interchangeable.The part's revision level is not required for this certification."

13a. Le présent bon de sortie certifie que les articles indiqués ci-dessus ont été construits conformément à :
Certifies that the items identified above were manufactured in conformity to :

☒ des données de conception approuvées et qu'ils peuvent être utilisés en toute sécurité
approved design data and are in condition for safe operation

☐ des données de conception non approuvées indiquées à la case 12.
non approved design data specified in block 12.

14a. ☐ RAC 571.10 (certification après maintenance) - CAR 571.10 Maintenance Release

☐ Autre réglementation précisée à la case 12 - Other regulation specified in block 12.

Le présent bon de sortie certifie que, sauf indication contraire à la case 12, le travail indiqué à la case 11 et décrit à la case 12, a été effectué conformément au Règlement de l'aviation canadien.
Certifies that unless otherwise specified in block 12, the work identified in block 11 and described in block 12 has been performed in compliance with the *Canadian Aviation Regulations*.

13b. Signature	13c. Numéro de l'organisme agréé Approved Organization Number	14b. Signature	14c. Numéro de l'organisme agréé Approved Organization Number
T.C. 4-58 NO. 131	4 - 58	S/O - N/A	S/O - N/A
13d. Nom - Name **I. Drukker**	13e. Date (dd/mmm/yyyy) 11/AUG/2022	14d. Nom - Name S/O - N/A	14e. Date (dd/mmm/yyyy) S/O - N/A

Le monteur doit contre-vérifier l'admissibilité avec les données approuvées - *Installer must cross-check eligibility with approved data*

RESPONSABILITÉS DU MONTEUR - INSTALLER RESPONSIBILITIES

1. Le présent bon de sortie ne constitue pas une autorisation de montage. - 1. This certificate does not constitute authority to install.

2. Le monteur qui travaille conformément à la réglementation d'un pays autre que celui spécifié à la case 1 doit s'assurer que la réglementation en question reconnaît la certification du pays ainsi spécifié. - 2. Installers working in accordance with the national regulations of a country other than that specified in block 1 must ensure that their regulations recognize certifications from country specified

3. Les déclarations des cases 13a et 14a ne constituent pas une certification du montage. Dans tous les cas, le dossier technique de l'aéronef doit inclure une certification de montage délivrée conformément à la réglementation nationale qui s'applique, avant que l'aéronef puisse voler. - 3. Statements in block 13a and 14a do not constitute installation certification. In all cases, the technical record for the aircraft must contain an installation certification issued in accordance with the applicable national regulations before the aircraft may be flown.

WC JR3-0154-E (2021-03)

1. Approving Competent Authority / Country	2.	3. Form Tracking Number
FEDERAL OFFICE OF CIVIL AVIATION SWITZERLAND	**AUTHORISED RELEASE CERTIFICATE** **EASA FORM 1**	80357896-0010 Page 1 of 1

4. Organisation Name and Address		5. Work order / Contract / Invoice
PILATUS	Address: PILATUS AIRCRAFT LTD CH-6371 STANS SWITZERLAND Telephone: +41 41 619 61 11	Your: PO#LJ-PO-050/SCA/X/2022 / Our : 243314

6. Item	7. Description	8. Part No.	9. Quantity	10. Serial No.	11. Status / Work
1	BOLT,HEX HD	6201106401	4	N/A	New

12. Remarks

FOR ADDITIONAL INFORMATION TO THE ITEM/S LISTED ABOVE, SEE CERTIFICATE OF CONFORMANCE 80357896

This document has been issued according to an approved computer generated signature procedure

13a. Certifies that the items identified above were manufactured in conformity to:

[X] approved design data and are in condition for safe operation

[ ] non-approved design data specified in block 12

14a. [ ] Part-145A.50 Release to Service [ ] Other regulation specified in block 12

Certifies that unless otherwise specified in block 12 the work identified in block 11 and described in block 12 was accomplished in accordance with Part-145 and in respect to that work the items are considered ready for release to service.

13b. Authorised Signature (Electronic Signature on File)	13c. Approval / Authorisation Number **CH.21G.0002**	14b. Authorised Signature	14c. Certificate / Approval Ref. No.:
13d. Name F. Micaletto	13e. Date (dd mmm yyyy) 13 Oct 2022	14d. Name	14e. Date (dd mmm yyyy)

USER / INSTALLER RESPONSIBILITIES

This certificate does not automatically constitute authority to install the item(s).

Where the user/installer performs work in accordance with regulations of an airworthiness authority different than the airworthiness authority specified in block 1, it is essential that the user/installer ensures that he/her airworthiness authority accepts items from the airworthiness authority specified in block 1.

Statements in block 13a and 14a do not constitute installation certification. In all cases the aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

BON DE SORTIE AUTORISÉE - AUTHORIZED RELEASE CERTIFICATE

FORM ONE

1. Autorité de l'aviation civile/Pays qui approuve le bon de sortie Approving Civil Aviation Authority/Country	2.	3. Numéro de suivi du formulaire - Form Tracking No.
Transport Canada		800732552-000002 4041795
4. Nom et adresse de l'organisme - Organization Name and Address	Pratt & Whitney Canada Cie. / Pratt & Whitney Canada Corp. 1000 boul. Marie-Victorin Longueuil, Québec Canada J4G 1A1 (450) 677-9411	5. Bon de travail/Contrat/Facture - Work Order/Contract/Invoice 3519811 4500361346

6. Article - Item	7. Description	8. Numéro de pièce - Part No.	9. Qté. - Qty	10. Numéro de série/Ne lot - Serial/Batch No.	11. Situation/Travail - Status/Work
000001	GASKET	ST3276-01	300 EA	N/A	NEW

12. Remarques - Remarks

NOTE: Les numéros de série figurant sur les étiquettes ou sur les pièces servent a la fabrication et ne doivent pas être inscrits sur le présent certificat.
NOTE: Any serial numbers that may be present on labels or parts are for manufacturing purposes and need not be referenced on this certificate.

Le cas échéant, la durée de stockage est précisée sur l'emballage de l'article -When applicable,shelf life is indicated on the item packaging

"Le numéro de pièce identifié sur la pièce peut inclure un suffixe alphabétique. Ce suffixe alphabétique denote le niveau de revision de la pièce. Tous les niveaux de revision pour une pièce donnée sont parfaitement interchangeables. Le niveau de revision d'une pièce n'est pas requis pour ce certificat."

"The part number marked on a part may include an alpha suffix.This alpha suffix denotes the parts's revision level.All revision levels for a given part number are fully interchangeable.The part's revision level is not required for this certification."

13a. Le présent bon de sortie certifie que les articles indiqués ci-dessus ont été construits conformément à :
Certifies that the items identified above were manufactured in conformity to :

☒ des données de conception approuvées et qu'ils peuvent être utilisés en toute sécurité
approved design data and are in condition for safe operation.

☐ des données de conception non approuvées indiquées à la case 12.
non approved design data specified in block 12.

14a. ☐ RAC 571.10 (certification après maintenance) - CAR 571.10 Maintenance Release

☐ Autre réglementation précisée à la case 12 - Other regulation specified in block 12.

Le présent bon de sortie certifie que, sauf indication contraire à la case 12, le travail indiqué à la case 11 et décrit à la case 12, a été effectué conformément au Règlement de l'aviation canadien.
Certifies that unless otherwise specified in block 12, the work identified in block 11 and described in block 12 has been performed in compliance with the Canadian Aviation Regulations.

13b. Signature	13c. Numéro de l'organisme agréé Approved Organization Number	14b. Signature	14c. Numéro de l'organisme agréé Approved Organization Number
T.C. 4-58 NO. 131	4 - 58	S/O - N/A	S/O - N/A
13d. Nom - Name	**13e. Date (dd/mmm/yyyy)**	**14d. Nom - Name**	**14e. Date (dd/mmm/yyyy)**
I. Drukker	17/AUG/2022	S/O - N/A	S/O - N/A

* Le monteur doit contre-vérifier l'admissibilité avec les données approuvées - *Installer must cross-check eligibility with approved data

RESPONSABILITÉS DU MONTEUR - INSTALLER RESPONSIBILITIES

1. Le présent bon de sortie ne constitue pas une autorisation de montage. - 1. This certificate does not constitute authority to install.

2. Le monteur qui travaille conformément à la réglementation d'un pays autre que celui spécifié à la case 1 doit s'assurer que la réglementation en question reconnaît la certification du pays ainsi spécifié. - 2. Installers working in accordance with the national regulations of a country other than that specified in block 1 must ensure that their regulations recognize certifications from country specified.

3. Les déclarations des cases 13a et 14a ne constituent pas une certification du montage. Dans tous les cas, le dossier technique de l'aéronef doit inclure une certification de montage délivrée conformément à la réglementation nationale qui s'applique, avant que l'aéronef puisse voler. - 3. Statements in block 13a and 14a do not constitute installation certification. In all cases, the technical record for the aircraft must contain an installation certification issued in accordance with the applicable national regulations before the aircraft may be flown.

DOCUMENT OFFICIEL – OFFICIAL RECORD

1. Approving Civil Aviation Authority/Country:	2.	3. Form Tracking Number:
FAA/United States	**AUTHORIZED RELEASE CERTIFICATE** FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG	1103309-1

4. Organization Name and Address:	5. Work Order/Contract/Invoice Number:
WOODWARD, INC. 5001 North Second Street Rockford, IL 61125 PQ2020CE	Inv.#4727926 P.O.#4500206473 S.O.#3219183

6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:
2	GOVERNOR - ASSY TURBO PROP	210615	1	20230749	NEW

12. Remarks: EXPORT AIRWORTHINESS APPROVAL. This article meets any special requirements of SWITZERLAND
This/these PMA part(s) is/are not a critical component

DRG.210615rev.D. For installation eligibility, login to "www.woodward.com/airsys/customer/pma_eligibility.cfm"

13a. Certifies the items identified above were manufactured in conformity to:

[X] **Approved design data and are in a condition for safe operation.**
[ ] **Non-approved design data specified in block 12.**

13b. Authorized Signature:	13c. Approval/Authorization No:
Linda M. Michaelis	676699092
13d. Name(Typed or Printed):	**13e. Date(dd/mmm/yyyy):**
Linda M. Michaelis	04/NOV/2015

14a. [ ] 14 CFR § 9 Subpart [illegible] [ ] Other regulation specified in Block 12

Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

14b. Authorized Signature:	14c. Approval/Certificate No:
14d. Name(Typed or Printed):	**14e. Date(dd/mmm/yyyy):**

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in block 1.

Statements in blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

1. Approving Competent Authority / Country	2.	3. Form Tracking Number
FEDERAL OFFICE OF CIVIL AVIATION SWITZERLAND	**AUTHORISED RELEASE CERTIFICATE** **EASA FORM 1**	80362001-0010 Page 1 of 1

4. Organisation Name and Address		5. Work order / Contract / Invoice
PILATUS	Address: PILATUS AIRCRAFT LTD CH-6371 STANS SWITZERLAND Telephone: +41 41 619 61 11	Your: PO#LJ-PO-050/SCA/X/2022 / Our : 243314

6. Item	7. Description	8. Part No.	9. Quantity	10. Serial No.	11. Status / Work
1	BOLT,BEARING	6201114901	4	N/A	New

12. Remarks

FOR ADDITIONAL INFORMATION TO THE ITEM/S LISTED ABOVE, SEE CERTIFICATE OF CONFORMANCE 80362001

This document has been issued according to an approved computer generated signature procedure

13a. Certifies that the items identified above were manufactured in conformity to:

[X] approved design data and are in condition for safe operation

[ ] non-approved design data specified in block 12

14a. [ ] Part-145A.50 Release to Service [ ] Other regulation specified in block 12

Certifies that unless otherwise specified in block 12 the work identified in block 11 and described in block 12 was accomplished in accordance with Part 145 and in respect to that work the items are considered ready for release to service.

13b. Authorised Signature (Electronic Signature on File)	13c. Approval / Authorisation Number CH.21G.0002	14b. Authorised Signature	14c. Certificate / Approval Ref. No.
13d. Name A. Keller	13e. Date (dd mmm yyyy) 25 Nov 2022	14d. Name	14e. Date (dd mmm yyyy)

USER / INSTALLER RESPONSIBILITIES

This certificate does not automatically constitute authority to install the item(s).

Where the user/installer performs work in accordance with regulations of an airworthiness authority different than the airworthiness authority specified in block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts items from the airworthiness authority specified in block 1.

Statements in block 13a and 14a do not constitute installation certification. In all cases the aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

1. Approving Civil Aviation Authority/Country:	2	3. Form Tracking Number:
FAA/UNITED STATES	**AUTHORIZED RELEASE CERTIFICATE** FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG	20211015F

4. Organization Name and Address:	5. Work Order/Contract/Invoice Number:
Hartzell Propeller Inc. One Propeller Place Piqua, Ohio 45356 (PC-10)	291336

6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:
1	BOLT,MOUNTING,9/16-18,12 POINT	B-3339	60	N/A	NEW
		END			

12. Remarks:

Additional Information: Eligibility information for Hartzell Propeller system components is contained in the appropriate Hartzell maintenance manual. This information is also available for Hartzell ice protection system components at www.hartzellprop.com

Packaging Methods used to protect against corrosion and damage: QC-05-003 **Duration of Methods: 6 months**

13a. Certifies the items identified above were manufactured in conformity to:

- [x] Approved design data and are in condition for safe operation
- [ ] Non-approved design data specified in Block 12

14a. [ ] 14 CFR 43.9 Return to Service [ ] Other regulations specified in Block 12

Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

13b. Authorized Signature:	13c. Approval/Authorization No.:	14b. Authorized Signature:	14c. Approval/Certificate No.:
Melissa Zickafoose	ODA-100082-CE		
13d. Name (Typed or Printed): Melissa . Zickafoose	13e. Date (dd/mmm/yyyy): 15/Oct/2021	14d. Name (Typed or Printed):	14e. Date (dd/mmm/yyyy):

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the engine/propeller/article.
Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness accepts engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.
Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.